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RESEARCH MEMORANDUM

TRANSONIC LOADS CHARACTERISTICS OF A
3-PERCENT-THICK 60° DELTA-WING-BODY
COMBINATION

By John M. Swihart and Willard E. Foss, Jr.

Langley Aeronautical Laboratory
Langley Field, Va.

CLASSIFIED DOCUMENT

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**NATIONAL ADVISORY COMMITTEE
FOR AERONAUTICS**

WASHINGTON

May 28, 1957

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SUMMARY

An investigation has been made in the Langley 16-foot transonic tunnel to determine the aerodynamic loading characteristics of a 3-percent-thick, aspect-ratio-2.06, 60° delta-wing--body combination. The Mach number range was from 0.80 to 1.05 and the average Reynolds number based on wing mean aerodynamic chord was 10×10^6 . The angle-of-attack range was from 0° to 26° but was limited at the highest Mach numbers by tunnel drive power.

Pressure distributions, spanwise loadings, integrated wing coefficients, and tabulated pressure coefficients are presented for the range of Mach numbers and angles of attack. The results indicate that a free leading-edge separation vortex is the dominant flow-field phenomenon at all Mach numbers and that, consequently, there are only slight changes in the spanwise loadings with Mach number. There is a slight outboard shift in center of pressure with an increase in Mach number. The chordwise position of the center of pressure varies from 46 to 55 percent of the mean aerodynamic chord when the Mach number is increased from 0.80 to 1.05.

INTRODUCTION

The delta-plan-form wing has been considered by many as particularly suited for supersonic flight because it has a low zero-lift drag coefficient and good structural integrity. Aerodynamic data have been obtained on many models with delta wings at subsonic, transonic, and supersonic speeds. (For example, see refs. 1 to 3.) Wing loads on the delta plan form have been reported at supersonic speeds in reference 4 and at low subsonic speeds in references 5 and 6. Flight tests of the Convair XF-92A

delta-wing research airplane have produced detailed wing pressure distributions and loads at low transonic speeds. (See refs. 7 and 8.) Reference 9 shows wind-tunnel data at transonic speeds for which the wing loads in the presence of a fuselage were obtained from force measurements.

The present paper shows wing pressure distributions and loadings at transonic speeds on a 3-percent-thick 60° delta-wing-body combination. This wing is a part of a series of investigations performed at the Langley 16-foot transonic tunnel of unswept, swept, and delta wings designed for supersonic flight and this investigation is a part of a program to obtain the aerodynamic characteristics, control characteristics, control loads, wing loads, and wing loads with deflected controls on a 60° delta-wing-body model.

The investigation covered a Mach number range from 0.80 to 1.05 with angles of attack to 26° . The Reynolds number range based on the wing mean aerodynamic chord was from 9.5×10^6 to 11.5×10^6 .

SYMBOLS

A	aspect ratio
b	wing span
c	local chord
c'	wing mean aerodynamic chord, $\frac{2}{3} \int_0^{b/2} c^2 dy$
\bar{c}	average wing chord, S/b
$c_m 0.35c$	wing-section pitching-moment coefficient about $0.35c$, $\int_0^{1.0} (c_{p,1} - c_{p,u}) (0.35 - \frac{x}{c}) d \frac{x}{c}$
c_m	wing-section pitching-moment coefficient about $0.35c'$, $c_m 0.35c' = c_n (0.35 - \frac{x'}{c})$
c_b	wing bending-moment coefficient about body center line, $\int_{0.19}^{1.0} c_n \frac{c}{\bar{c}} \frac{y}{b/2} d \frac{y}{b/2}$

c_m wing pitching-moment coefficient about $0.35c'$,

$$\int_{0.19}^{1.0} c_m \frac{c^2}{cc'} d \frac{y}{b/2}$$

c_n wing-section normal-force coefficient,

$$\int_0^{1.0} (c_{p,l} - c_{p,u}) d \frac{x}{c}$$

c_N wing normal-force coefficient perpendicular to body center
line, $\int_{0.19}^{1.0} c_n \frac{c}{c} d \frac{y}{b/2}$

l body length

M Mach number

p free-stream static pressure

c_p pressure coefficient, $\frac{p_{local} - p}{q}$

q free-stream dynamic pressure

r body radius

S wing area (includes area covered by body)

x distance from wing leading edge or body nose (positive
rearward)

x' distance from wing leading edge to a line perpendicular to
plane of symmetry and passing through $0.35c'$

$\frac{x_{cp}}{c}$ section chordwise center-of-pressure position

$\frac{x_{cp}}{c}$ wing chordwise center-of-pressure position, $0.35 - \frac{c_m}{c_N}$

y spanwise distance measured from body center line

$\frac{y_{cp}}{b/2}$ wing spanwise center-of-pressure position

α angle of attack of body center line

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- Λ angle of sweepback
 λ taper ratio
 ϕ meridian angle of body orifices

Subscripts:

- l wing lower surface
 u wing upper surface
 cp center of pressure

APPARATUS AND METHODS

Tunnel

This investigation was conducted in the Langley 16-foot transonic tunnel. It is a single-return atmospheric tunnel with a slotted test section and has been described in reference 10. The tunnel model support system is arranged so that the model is located near the center line of the tunnel at all angles of attack.

Model

Figure 1 shows the model mounted in the Langley 16-foot transonic tunnel test section. Figure 2 is a sketch showing model details and the locations of the wing and body pressure orifices. The steel wing was mounted in a midwing position on the body and had no geometric incidence, twist, or dihedral. The plane delta wing had 60° of leading-edge sweep, 0° of trailing-edge sweep, and NACA 65A003 airfoil sections parallel to the free stream. The controls shown in figure 2 were not deflected for this part of the investigation. The wing had slightly rounded wing tips which effectively reduced the span and area and increased the mean aerodynamic chord. (See fig. 2.) Actual span was used in determining the spanwise locations of the orifice rows. The body ordinates are given in figure 2 and the body consists of an ogival nose, a cylindrical midsection, and a slightly boattailed afterbody. The ratio of body base diameter to maximum diameter is 0.66 and of body maximum cross-sectional area to wing area is 0.061.

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Instrumentation

Pressures were measured at six spanwise stations on the wing and at seven meridian rows on the body as shown in figure 2. At the innermost wing station the orifices were installed in the fuselage shell about $1/16$ inch from the wing surface. The model angle of attack was measured by a calibrated pendulum strain-gage angle-of-attack indicator located in the model nose. The angles of attack thus determined were independent of sting and balance deflections under load.

After the pressure tests were complete, the pressure orifices located at 2.5- and 50-percent chord at each spanwise station were used to emit ink onto the wing upper surface for flow visualization. The ink-flow studies presented in this report were performed in a manner similar to that described in references 11 and 12.

Tests

Pressure data were obtained over a Mach number range from 0.80 to 1.05 at angles of attack to 26° . In general, the angle of attack was varied at a constant Mach number from 0° to 26° in increments of about 2° . Tunnel drive power limited the angle-of-attack range at Mach numbers of 1.00 and above. The Reynolds number range based on c' was from 9.5×10^6 to 11.5×10^6 over the Mach number range of the investigation.

Reduction of Data

The pressures were photographically recorded on five 100-tube mercury manometer boards. The film was read on an automatic film reader and recorded on punchcards. Automatic computing machines reduced the data from the punchcards to individual pressure coefficients and also performed a rectangular-step integration of the pressure coefficients to obtain c_n and $c_m 0.35c$. An automatic plotting device utilized the data cards to prepare the chordwise pressure-distribution plots presented subsequently.

Accuracy

The following is the estimated accuracy of the quantities presented in this paper:

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M	± 0.005
α , deg	± 0.10
C_p	± 0.005
C_N	± 0.01
C_m	± 0.001
C_b	± 0.05

Static loads were applied to the wing at two chordwise positions at each of five spanwise stations to determine the twisting characteristics of this steel wing in a manner somewhat similar to that described in reference 13. It was found that both the spanwise variation of twist and the maximum tip twist were very small during the static calibration. The wing twist calculated for the conditions of maximum loading obtained during the wind-tunnel investigation was found to vary approximately linearly to -0.2° at the $0.75b/2$ station and then increase to -0.5°

$$\text{at } \frac{y}{b/2} = 0.90.$$

RESULTS

The results of the investigation are presented as follows:

Figure

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Table

Wing pressure coefficients at six spanwise stations	I
Section c_n , section c_m , $\frac{x_{cp}}{c}$, and α	II

Pressure distributions on the wing were measured in 2° increments from 0° to the maximum obtainable at each test Mach number but all the pressure-distribution plots have not been presented herein; however, table I contains pressure coefficients for the complete Mach number and angle-of-attack range.

DISCUSSION

Flow Characteristics

Studies of the flow characteristics over swept wings at transonic speeds have been reported in references 11 and 13. These studies were for a 6-percent-thick wing and a 4-percent-thick wing with more leading-edge sweep, respectively. Low-speed studies of the flow over a 60° delta wing have been reported in reference 5 and some wake surveys behind a 60° delta wing have been shown in reference 14.

It would be expected that the flow over a 3-percent-thick 60° delta wing would be somewhat similar to the 4-percent-thick wing of $A = 3$, $\lambda = 0.2$, with $\Delta c/4 = 45^\circ$ shown in reference 13. It was found that they have some similar characteristics at low angles of attack. The 60° delta wing develops a strong leading-edge vortex similar to the delta wing described in reference 5 at low speeds and this conical vortex flow is the dominant flow disturbance over the Mach number range of the investigation.

Wing and body pressure-distribution observations. - Reference 5 suggested that at low speeds the separation vortex was well defined at an angle of attack of 4° , and that the tip sections were more highly loaded than the inboard sections because of the increased induced angle of attack indicated by triangular-wing potential flow. Figure 3(a) indicates that, at $M = 0.80$ and $\alpha = 4^\circ$, the vortex is well-defined, but that the vortex has probably turned streamwise near the 90-percent-semispan station and this station is separated. It was noted in reference 5 that the vortex core was marked by a low-pressure region and reattachment of the flow was marked by a pressure rise. This same flow phenomenon is noted herein at all Mach numbers and for all angles of attack below $\alpha = 14^\circ$ as is shown in figure 3. It should be noted that, as the conical vortex system moves spanwise, it increases in chordwise extent as is shown by the wider regions of negative pressures. Also, as the angle of attack is increased, the center of the vortex is swept back more at each spanwise station (fig. 3(a), $\alpha = 10.3^\circ$, for example). At the higher angles of attack ($\alpha = 14^\circ$ and above), the vortex core is swept back until the vortex has moved off the wing trailing edge near the $\frac{y}{b/2} = 0.65$ station.

Inspection of the pressure distributions (fig. 3) shows that the flow over the wing sections outboard of the station where the conical vortex leaves the wing is separated at all angles of attack shown. Tuft studies made on this wing indicated complete spanwise flow and even reversed flow in these regions. The tuft pictures also showed that the flow was predominantly streamwise at the stations inboard of the conical vortex. Similar flow on slender delta wings is discussed in reference 15. A Mach number of 0.94 has been shown for several previous transonic investigations to be representative of the mixed flows in the transonic regime and to be a particularly troublesome flight region; therefore, a more complete angle-of-attack coverage has been shown for this Mach number in figure 3(e). In addition to the predominant vortex system, the decelerating flow-field shock and possibly a weak leading-edge shock are indicated at this Mach number. The small pressure rise on the upper surface at the wing leading edge ($\alpha = 6^\circ$, 8° , and 10°) might indicate a weak oblique shock, and the pronounced pressure rise at the 90-percent-chord of all stations except the tip station locates the flow-field shock. This shock moves to the wing trailing edge at the higher angles of attack. The weak leading-edge shock remains; a trailing-edge shock is formed; and the decelerating flow-field shock moves rearward with increasing Mach number.

Figure 4 shows the pressure distribution along the 0° and 180° body meridians for several Mach numbers and angles of attack. The pressure coefficients from the 23° meridian are shown from $\frac{X}{l} = 0.82$ rearward because there are no pressure orifices at the 0° meridian in this region. It is estimated that the pressures shown for $\phi = 23^\circ$ are indicative of those at the $\phi = 0^\circ$ meridian. The pressure distributions are typical for a body in the presence of a wing, except that the body is influenced by the wing over a large portion of its length because of the large wing-root chord. The decelerating flow-field shock is noted also in the body pressure distributions at $M = 0.90$ and $M = 0.94$. In figure 4(b) a sharp expansion is noted at $\frac{X}{l} = 0.77$ at Mach numbers of 1.00 and 1.05 for the $\phi = 0^\circ$ meridian. This sharp expansion is probably caused by a local curvature in the body section joint just ahead of the wing trailing edge and by the rapid reduction in the wing-body area distribution at this point. The wing cross-sectional area is rapidly decreasing to zero and the body has just started to boattail from the cylindrical midsection. In general, the variations shown in figure 4 indicate that a relatively high load is being induced on the body by the wing, particularly at the higher angles of attack (fig. 4(c)).

Pressure contours.— Figure 5 shows pressure contours on the wing upper surface for Mach numbers of 0.80, 0.94, and 1.05 at angles of attack of 4° , 8° , 10° , and 14° . These contours are presented to show the sweeping back of the vortex core as the angle of attack is increased and the general similarity of the flow at all Mach numbers.

It was suggested in reference 13 that the flow on the 4-percent-thick sweptback wing was marked by a shock which originated at the wing-leading-edge-body juncture and became swept back more and more with increasing angle of attack. It is believed that the sharp pressure rise shown in figure 5 (near coincidence of the isobars) may be a shock. It cannot be conclusively proved that a shock exists at this location because the local Mach number perpendicular to the isobars is generally less than 1.0. The flow probably turns towards the body as it passes over the conical vortex and, since the local flow is supersonic, a shock might result upon reattachment of the flow to the wing. This shock would tend to turn the flow streamwise again.

Ink-flow pictures.— Ink-flow pictures for several Mach numbers and angles of attack are shown in figure 6. These pictures were obtained by photographing the wing as the angle of attack was increased at a constant Mach number. At a Mach number of 0.80, there is evidence of the conical vortex flow at an angle of attack of 4° , because some ink is being pulled into the leading-edge region and flowing down the leading edge. This trend continues and gets stronger as the angle of attack is increased to 6° and 8° . At 8° it is apparent that the vortex has increased in chordwise extent and has turned streamwise near the tip. An indication of the strength of this vortex core can be ascertained by the strong spanwise flow of ink at the higher angles of attack. The slight outboard turning of the ink (white arrow) in figure 6(b) at $\alpha = 12^\circ$ is typical of the flow turning caused by a shock and figure 3(c) ($\alpha = 12.3^\circ$) shows the flow-field shock at about 0.80c. Reference 16 shows the rolling up of the wing vortices behind low-aspect-ratio sweptback wings and indicates that the vortex centers move inboard from the tips as the lift is increased but stay in the same vertical position with respect to the airstream. These ink-flow pictures, pressure-contour plots, and wing pressure distributions all substantiate the theory regarding the inboard movement of the vortex center. The ink-flow pictures at the higher Mach numbers are presented to show similarity of the flow at all Mach numbers investigated. The persistence of the vortex type of flow is noted at a Mach number of 1.05, where the patterns are similar to those at the lower speeds except that the boundaries seem to be more sharply defined.

Loading Characteristics

Fuselage carryover load.— Figure 7 shows the variation of section normal-load parameter over the wing semispan for several angles of attack at a Mach number of 0.94. The load over the body was obtained by integrating the body pressures at the 0° , 180° , 204° , 233° , 307° , and 337° meridians and weighting them over the local chord of the blanketed wing. The data indicate considerable carryover at $\alpha = 6.3^\circ$, but a reduced amount is carried by the body at the higher angles of attack. The irregular body load shown at $\alpha = 17.8^\circ$ is probably the result of the body.

shedding a separation vortex near the wing leading edge. The disappearance of the peak at $\alpha = 26.2^\circ$ would indicate that the vortex has been shed much closer to the body nose.

Reference 17 presents a theory for the spanwise loading on a slender delta wing where the flow field consists of a concentrated conical separation vortex and a feeding vortex sheet. Since it appears that the flow over this wing is predominated by a conical separation vortex, it is of interest to compare the theoretical and experimental (fig. 7) spanwise loadings. The theoretical curves were obtained by using those of figure 8 of reference 17 and adjusting them to an apex angle of 30° . Also shown in figure 7 of this report is the slender-wing theory of reference 18 which produces elliptical spanwise loadings. At an angle of attack of 6° the vortex theory and the experimental data are in only fair agreement inboard of $\frac{y}{b/2} = 0.75$. Outboard of this station the theory overestimates the load considerably. The elliptical load of the slender-wing theory provides a better estimate of the load at this angle of attack but does not predict the influence of the vortex between $\frac{y}{b/2} = 0.50$ and 0.75 like the theory of reference 17. The vortex theory and the experimental data are in good agreement over the inboard 55 percent of the span at $\alpha = 12^\circ$. The vortex theory does not consider any sweepback, streamwise turning of the vortex from the leading edge, or tip separation outboard of the vortex; consequently, it considerably overestimates the load over the outboard portions. It is of interest to note, however, that the slender-wing theory seriously underestimates the load carried over the inboard 75 percent of the span. The effect of the body was not included in the two theories, but the presence of the body would increase the upflow at the wing root. This condition would tend to lessen the agreement with the slender-wing theory at the low angle and to improve it at $\alpha = 12.5^\circ$. It is apparent that the conical vortex is creating some high suction pressures over the wing upper surface, increasing the lift-curve slope, and causing the wing to carry a higher load inboard than would be predicted by the theory of reference 18. Reference 15 confirms this flow phenomenon.

Wing spanwise loads.—Figure 8 shows the variation of the section normal-load parameter over the wing semispan for constant values of wing normal-force coefficient at several Mach numbers. The loadings are typical of the distributions obtained on plane highly swept wings in the transonic speed range. (See ref. 13.) Inspection also reveals the influence of the leading-edge vortex described previously. Separation outboard causes the tip loads to increase only slightly with increasing C_N , but the low pressures associated with the vortex core cause the middle stations to carry additional load. (See fig. 8(c), $C_N = 0.60$.) This was also noted in reference 19 and was included in the calculations of the load presented therein. It should be noted, however, that the tip sections are not

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non-load-carrying sections and, at a Mach number of 1.00, they carry more load at a given C_N than at the lower Mach numbers. The sharply defined shift of load distribution over the outboard sections for $C_N = 0.60$ at $M = 0.80$ and 0.90 (figs. 8(a) and 8(b)) and $C_N = 0.80$ at $M = 0.94$ (fig. 8(c)) will be shown later to be the inflection point of the wing normal-force curve and pitching-moment curve.

Center-of-Pressure Characteristics

Section chordwise center of pressure. - Figure 9 shows the spanwise variation of wing-section center-of-pressure location for several Mach numbers at constant normal-force coefficient. The data indicate that the $\frac{y}{b/2} = 0.75$ station center-of-pressure shifts rapidly rearward between $C_N = 0.20$ and 0.40. As the Mach number is increased to $M = 1.00$ and 1.05, the local centers of pressure all tend to shift more rearward, as would be expected. In general, it will be noted that the least spanwise variation is shown for $C_N = 0.80$ at all Mach numbers. This is probably the result of separation at most of the spanwise sections.

Wing chordwise center of pressure. - The variation with Mach number of the wing center of pressure in terms of the mean aerodynamic chord is shown in figure 10 for several values of wing normal-force coefficient. The data indicate that the center of pressure varied from about $0.46c'$ to $0.55c'$ when the Mach number was increased from 0.80 to 1.05. The greatest chordwise variation was indicated at $C_N = 0.60$. The center-of-pressure locations shown here are in very good agreement with the location map presented in reference 9.

Wing lateral center of pressure. - Figure 11 shows the variation of spanwise center-of-pressure location for several Mach numbers and normal-force coefficients. The lateral center of pressure moves inboard as the wing normal-force coefficient increases at all Mach numbers and shifts outward slightly with increasing Mach number. The relatively large (3 percent) movement shown for $C_N = 0.60$ between $M = 0.90$ and 0.94 reflects the outboard shift in loading at this C_N . (See figs. 8(b) and 8(c).)

Wing Force and Moment Characteristics

Wing normal force. - The section loading parameter curves shown in figure 8 were integrated outboard of the body intersection to produce the wing normal-force coefficients shown plotted against angle of attack in figure 12. The inflection points in these normal-force curves correspond to the inboard shifts in the previously shown spanwise load distributions.

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Wing bending-moment characteristics. - The effect of Mach number on the wing bending-moment coefficient for constant values of wing normal-force coefficient is shown in figure 13. There are only very small variations over the Mach number range at a constant C_N and this result should be expected, since the spanwise loadings are very similar at all Mach numbers.

Wing pitching moment. - Figure 14 shows the variation of wing pitching-moment characteristics with wing normal-force coefficient for several Mach numbers. Some tendency towards pitchup is noted at $M = 0.80$ and definite pitchup is shown at $M = 0.90$ and $M = 0.94$. These tendencies were indicated by rapid local section center-of-pressure movement near the normal-force coefficients where pitchup occurs and are the result of wing-tip separation. Improvement in the pitchup tendencies could probably be obtained by the use of fences or camber.

SUMMARY OF RESULTS

An investigation has been made in the Langley 16-foot transonic tunnel to determine the aerodynamic loading characteristics of a 3-percent-thick, aspect-ratio-2.06, 60° delta-wing-body combination at Mach numbers from 0.80 to 1.05 and at angles of attack from 0° to 26° . The significant results may be summarized as follows:

1. The leading-edge separation vortex which was noted previously at low subsonic speeds was found to be the predominant flow phenomena over the entire Mach number range of this investigation.
2. Where the usual transonic flow-field shocks and wing shocks were noted, these shocks only slightly modified the flow field created by the leading-edge separation vortex.
3. The chordwise center of pressure shifted from about 46 percent to 55 percent of the mean aerodynamic chord when the Mach number was increased from 0.80 to 1.05.
4. The wing spanwise loading was very similar at all Mach numbers with only a slight outboard shift in center of load with an increase in Mach number and a small inboard shift as the normal-force coefficient increased.

5. Pitchup was indicated at Mach numbers of 0.90 and 0.94 and was caused by tip separation outboard of the free leading-edge separation vortex.

Langley Aeronautical Laboratory,
National Advisory Committee for Aeronautics,
Langley Field, Va., March 22, 1957.

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TABLE I
PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 60° DELTA WING

R = 0.80

$\alpha \approx -2.1$

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TABLE L - Continued
PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 60° DELTA WING

$$\mu = 0.80$$

a = 643

TABLE I.- Continued

PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 60° DELTA WING

 $M = 0.80$

$\alpha = 8.3$						
Upper surface					Lower surface	
Stations, fraction of semispan						
x/c	0.10	0.32	0.53	0.66	0.74	0.90
.00	.092	-1.091	-0.825	-0.856	-0.717	
.01	-.290	-.868	-.760	-.860	-.719	-.472
.03	-.338	-.913	-.775	-.830	-.726	-.461
.05	-.368	-1.114	-.791	-.767	-.726	-.460
.08	-.362	-1.140	-.837	-.750	-.726	-.456
.10	-.336	-.966	-.853	-.750	-.728	-.450
.15	-.320	-.257	-.941	-.777	-.734	-.442
.19			-.965	-.795	-.756	-.434
.20	-.290					
.25	-.298	-.255	-1.006	-.823	-.765	-.424
.30	-.268	-.268	-.983	-.849	-.798	-.412
.35	-.275	-.272	-.877	-.859	-.822	-.401
.40	-.242	-.255	-.708	-.868	-.849	-.393
.45	-.287	-.249	-.475	-.872	-.862	-.380
.50	-.253	-.245	-.270	-.821	-.874	-.365
.55	-.239	-.240	-.146	-.757	-.877	
.60	-.228	-.217	-.085	-.647	-.858	-.333
.65	-.208	-.203	-.059	-.473	-.796	
.68						
.70	-.180	-.171	-.081	-.450	-.749	-.302
.73						
.75	-.153	-.145	-.054		-.782	
.77						
.79						
.80	-.132	-.108				
.81						
.82						
.83						
.85	-.095	-.081	-.009			
.87						
.88						
.89						
.90	-.098					
.91						
.93	-.032	-.039				
.95	-.057	-.022				
.96						
.97						
.98						

$\alpha = 10.5$						
Upper surface					Lower surface	
Stations, fraction of semispan						
x/c	0.10	0.32	0.53	0.66	0.74	0.90
.00	.057	-1.278	-.939	-.918	-.799	
.01	-.437	-.146	-.928	-.923	-.809	-.503
.03	-.447	-.143	-.927	-.928	-.814	-.493
.05	-.510	-.145	-.927	-.922	-.814	-.493
.08	-.508	-.142	-.927	-.920	-.814	-.493
.10	-.447	-.140	-.1002	-.874	-.808	-.483
.13	-.415	-.082	-.1035	-.910	-.816	-.471
.19						
.20	-.370					
.25	-.370	-.296	-1.253	-.839	-.446	
.30	-.329	-.297	-.1296	-.968	-.835	-.493
.35	-.336	-.322	-.1271	-.1025	-.820	-.425
.40	-.297	-.301	-.1180	-.1094	-.822	-.421
.45	-.338	-.283	-.910	-.1139	-.831	-.417
.50	-.298	-.280	-.652	-.1136	-.874	-.406
.55	-.278	-.274	-.401	-.1129	-.935	
.60	-.266	-.244	-.182	-.1021	-.989	-.375
.65	-.236	-.228	-.113	-.058	-.1028	
.68						
.70	-.202	-.187	-.094	-.786	-.1068	-.334
.73						
.75	-.174	-.155	-.057		-.1026	
.77						
.80						
.81	-.143	-.116				
.82						
.83						
.85						
.87						
.88						
.89						
.90	-.095					
.91						
.93	-.031	-.065				
.95	-.059	-.017				
.96						
.97						
.98						

TABLE L- Continued
PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 60° DELTA WING

 $M = 0.80$ $\alpha = 12.3$

Upper surface						Lower surface						
Stations, fraction of semispan						Stations, fraction of semispan						
x/c	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90
.00	-0.004	-1.303	-1.050	-1.029	-0.778		.428	.313	.234	.167	.147	.042
.01	-0.580	-1.261	-1.060	-1.049	-0.781	-0.449	.428	.368	.282	.263	.227	.154
.03	-0.598	-1.282	-1.068	-1.046	-0.775	-0.443	.455	.372	.312	.305	.282	.210
.05	-0.723	-1.359	-1.068	-1.015	-0.761	-0.450	.455	.346	.307	.296	.286	.224
.08	-0.688	-1.489	-1.090	-0.994	-0.739	-0.456	.455	.346	.307	.296	.286	.224
.10	-0.570	-1.687	-1.116	-0.969	-0.714	-0.458	.382	.326	.298	.295	.282	.220
.15	-0.512	-1.402	-1.168	-0.975	-0.683	-0.461	.341	.293	.274	.262	.265	.201
.19					-0.982	-0.464		.264	.243	.235	.260	.196
.20	-0.463						.299					
.25	-0.451	-0.416	-1.123	-0.974	-0.649	-0.469	.277	.230	.216	.214	.159	
.30	-0.432	-0.358	-1.173	-0.925	-0.634	-0.469	.243	.213	.201	.189	.194	.147
.35	-0.407	-0.382	-1.369	-0.866	-0.605	-0.473	.224	.189	.175	.162	.167	.136
.40	-0.362	-0.366	-1.414	-0.813	-0.572	-0.479	.195	.176	.160	.159	.159	.093
.45	-0.393	-0.320	-1.233	-0.780	-0.555	-0.480	.158	.151	.140	.122	.075	.076
.50	-0.346	-0.298	-1.018	-0.772	-0.523	-0.480	.152	.142	.129	.094	.064	.056
.55	-0.312	-0.278	-0.764	-0.612	-0.488	-0.488	.129	.119	.105	.089	.064	
.60	-0.284	-0.256	-0.451	-0.496	-0.409	-0.475	.122	.109	.097	.077	.046	.020
.65	-0.246	-0.208	-0.250	-0.729	-0.588		.103	.088	.075	.054	.006	.032
.70	-0.209	-0.164	-0.173	-0.737	-0.675	-0.462	.080	.076	.053	.019	.001	-0.033
.75	-0.179	-0.146	-0.169		-0.673		.066	.051	.026		.029	
.77					-0.672					-0.001		
.79					-0.670					-0.008		
.80	-0.163	-0.129			-0.659	-0.449	.049	.041			-0.019	-0.093
.81					-0.667					-0.016		
.82										.011		
.83										.006	-0.024	
.85	-0.141	-0.125	-0.244		-0.642		.021	-0.001		-0.005	-0.043	
.87										.006	-0.042	
.88										.001		
.89										.006		
.90	-0.155		-0.227		-0.620	-0.432		-0.005		-0.020	-0.095	-0.190
.91	-0.120		-0.531							.011	-0.070	
.93	-0.108		-0.203							.021	-0.037	
.95	-0.117	-0.098			-0.569					-0.032		-0.183
.96					-0.484						-0.130	
.97					-0.188						-0.052	
.98					-0.080						-0.045	

 $\alpha = 14.4$

Upper surface												Lower surface												
Stations, fraction of semispan												Stations, fraction of semispan												
x/c	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	
.00	-0.065	-1.373	-1.258	-1.006	-0.648		.449	.312	.213	.148	.142	.029												
.01	-0.702	-1.372	-1.280	-1.018	-0.642	-0.482	.483	.401	.301	.270	.233	.122												
.03	-0.738	-1.374	-1.284	-1.016	-0.655	-0.478	.483	.401	.301	.270	.233	.122												
.05	-0.983	-1.380	-1.277	-1.003	-0.641	-0.483	.523	.420	.343	.329	.296	.193												
.08	-0.802	-1.471	-1.280	-0.977	-0.619	-0.489	.505	.397	.340	.329	.311	.219												
.10	-0.714	-1.669	-1.263	-0.955	-0.602	-0.492	.463	.385	.319	.335	.305	.222												
.15	-0.580	-1.660	-1.213	-0.919	-0.572	-0.497	.410	.350	.293	.309	.295	.207												
.19	-0.110	-1.195	-0.888	-0.561	-0.501			.320	.265	.283	.293	.204												
.20	-0.533						.360																	
.25	-0.499	-0.688	-1.223	-0.819	-0.541	-0.509	.336	.287	.245	.261	.246	.172												
.30	-0.436	-0.515	-1.172	-0.749	-0.541	-0.511	.304	.269	.225	.234	.228	.186												
.35	-0.436	-0.479	-1.200	-0.696	-0.551	-0.515	.281	.246	.205	.206	.200	.153												
.40	-0.396	-0.448	-1.252	-0.666	-0.568	-0.521	.255	.227	.182	.180	.172	.105												
.45	-0.412	-0.398	-1.220	-0.674	-0.588	-0.522	.212	.200	.159	.168	.157	.089												
.50	-0.358	-0.347	-1.132	-0.715	-0.620	-0.519	.204	.186	.159	.131	.129	.059												
.55	-0.319	-0.281	-0.978	-0.751	-0.660		.182	.164	.125	.124	.105													
.60	-0.277	-0.208	-0.756	-0.732	-0.694	-0.514	.172	.148	.099	.109	.068	.032												
.65	-0.228	-0.170	-0.648	-0.691	-0.693		.149	.123	.074	.075	.018													
.68					-0.701						.045													
.70	-0.181	-0.149	-0.564	-0.737	-0.704	-0.503		.124	.109	.041	.039	.018	-0.027											
.73					-0.694						.008													
.75	-0.163	-0.166	-0.523		-0.681			.104	.077	.010		.007												
.77					-0.708						.008													
.79					-0.715						.004													
.80	-0.176	-0.191			-0.666	-0.495		.077	.063			-0.006												
.81					-0.718																			
.82					-0.527						.002													
.83					-0.516	-0.684		-0.651			-0.006													
.85	-0.173	-0.204	-0.526		-0.673						.044	.001	.011	.015										
.87		-0.209	-0.519								.008	.031	.039											
.88		-0.211									.001													
.89		-0.206									-0.004													
.90	-0.218		-0.486			-0.629	-0.482		.006		-0.013		.061											
.91		-0.210			-0.624						.022		.084											
.93		-0.191	-0.435			-0.588					.041	-0.043		.099										
.95	-0.169	-0.175			-0.579																			
.96			-0.389																					
.97			-0.138																					
.98																								

~~NACA REPORT PLATE~~

TABLE I.- Continued
PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 60° DELTA WING
M_∞ = 0.80

TABLE I.- Continued													
PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 60° DELTA WING $X = 0.80$													
$\alpha = 16^\circ 4$													
Upper surface					Lower surface								
x/c	Stations, fraction of semispan	0.10	0.32	0.55	0.65	0.74	0.80	0.10	0.32	0.55	0.65	0.74	0.80
.00	-1.55	-1.537	-1.307	-0.946	-0.751	-0.601	-0.446	+0.294	+0.196	+0.126	+0.095	+0.046	
.01	-0.805	-1.509	-1.320	-0.951	-0.756	-0.602	+0.516	+0.415	+0.289	+0.263	+0.201	+0.071	
.03	-0.960	-1.491	-1.318	-0.944	-0.765	-0.604	+0.577	+0.451	+0.354	+0.336	+0.288	+0.172	
.05	-1.154	-1.817	-1.298	-0.921	-0.765	-0.590	+0.553	+0.434	+0.370	+0.349	+0.309	+0.219	
.08	-0.972	-1.801	-1.280	-0.892	-0.765	-0.586	+0.510	+0.425	+0.370	+0.354	+0.310	+0.233	
.10	-0.848	-1.703	-1.247	-0.873	-0.765	-0.579	+0.452	+0.392	+0.354	+0.331	+0.310	+0.228	
.13	-0.698	-1.817	-1.238	-0.866	-0.775	-0.566	+0.381	+0.328	+0.306	+0.312	+0.312	+0.233	
.19	-1.377	-1.245	-0.874	-0.795	-0.556	-0.404	+0.379	+0.328	+0.302	+0.287	+0.273	+0.207	
.20	-0.621	-1.045	-1.159	-0.850	-0.793	-0.543	+0.340	+0.306	+0.280	+0.259	+0.257	+0.191	
.30	-0.527	-0.801	-1.069	-0.821	-0.798	-0.534	+0.319	+0.283	+0.257	+0.232	+0.231	+0.188	
.35	-0.518	-0.643	-1.003	-0.826	-0.797	-0.529	+0.291	+0.265	+0.238	+0.210	+0.204	+0.145	
.40	-0.468	-0.529	-0.997	-0.862	-0.794	-0.523	+0.249	+0.235	+0.214	+0.190	+0.188	+0.128	
.45	-0.459	-0.439	-1.014	-0.901	-0.785	-0.520	+0.237	+0.220	+0.192	+0.163	+0.170	+0.101	
.50	-0.385	-0.356	-1.051	-0.907	-0.774	-0.513	+0.211	+0.193	+0.171	+0.155	+0.144	+0.088	
.55	-0.326	-0.297	-1.034	-0.906	-0.758	-0.502	+0.198	+0.177	+0.155	+0.135	+0.105	+0.068	
.60	-0.290	-0.292	-0.960	-0.863	-0.741	-0.492	+0.176	+0.150	+0.132	+0.109	+0.055	+0.032	
.65	-0.285	-0.374	-0.915	-0.821	-0.710	-0.470	+0.145	+0.139	+0.104	+0.069	+0.051	+0.007	
.68	-0.320	-0.428	-0.861	-0.838	-0.694	-0.492	+0.127	+0.099	+0.068	-	+0.041	+0.038	
.73	-0.360	-0.444	-0.769	-0.683	-0.780	-0.477	+0.104	+0.083	-	+0.037	+0.030	+0.017	
.77	-0.384	-0.386	-0.777	-0.644	-0.476	+0.058	+0.020	+0.017	+0.008	+0.024	+0.017	+0.007	
.81	-0.324	-0.571	-0.771	-0.602	-0.476	+0.023	+0.007	+0.007	+0.008	+0.018	+0.017	+0.007	
.82	-0.376	-0.650	-0.738	-0.626	+0.058	+0.026	+0.017	+0.021	+0.013	+0.038	+0.026	+0.010	
.83	-0.329	-0.643	-0.613	-0.718	+0.027	+0.017	+0.008	+0.021	+0.013	+0.038	+0.026	+0.017	
.87	-0.282	-0.265	-0.248	-0.550	-0.602	-0.461	+0.022	+0.024	+0.016	+0.018	+0.069	+0.074	+0.171
.88	-0.255	-0.234	-0.480	-0.674	-0.602	-0.461	+0.023	+0.024	+0.022	+0.016	+0.069	+0.074	+0.172
.91	-0.173	-0.163	-0.418	-0.621	-0.568	+0.041	+0.099	+0.099	+0.153	+0.153	+0.153	+0.234	
.98	-0.116	-0.116	-0.116	-0.102	-0.797	-0.530	+0.392	+0.390	+0.327	+0.327	+0.327	+0.237	
$\alpha = 17^\circ 7$													
.00	-1.229	-1.636	-1.236	-1.060	-0.863	-0.693	+0.409	+0.362	+0.325	+0.311	+0.289	+0.215	
.01	-0.861	-1.592	-1.234	-1.078	-0.871	-0.645	+0.335	+0.427	+0.302	+0.248	+0.177	+0.082	
.03	-1.127	-1.561	-1.216	-1.085	-0.867	-0.641	+0.319	+0.475	+0.370	+0.333	+0.274	+0.168	
.05	-1.308	-1.859	-1.183	-1.072	-0.856	-0.537	+0.291	+0.463	+0.345	+0.324	+0.309	+0.247	
.08	-1.112	-1.639	-1.158	-1.059	-0.840	-0.539	+0.248	+0.457	+0.380	+0.347	+0.319	+0.233	
.10	-0.955	-1.732	-1.155	-1.048	-0.826	-0.536	+0.289	+0.426	+0.375	+0.347	+0.325	+0.234	
.13	-0.756	-1.685	-1.203	-1.046	-0.808	-0.535	+0.289	+0.426	+0.375	+0.347	+0.325	+0.234	
.19	-1.481	-1.164	-1.042	-0.797	-0.530	-0.438	+0.409	+0.362	+0.325	+0.311	+0.289	+0.215	
.20	-0.683	-1.208	-1.153	-1.000	-0.771	-0.526	+0.374	+0.341	+0.302	+0.288	+0.275	+0.197	
.30	-0.588	-0.917	-1.115	-0.959	-0.755	-0.522	+0.349	+0.316	+0.282	+0.257	+0.249	+0.192	
.35	-0.593	-0.666	-1.085	-0.925	-0.735	-0.522	+0.319	+0.294	+0.263	+0.232	+0.220	+0.155	
.40	-0.470	-0.566	-1.092	-0.903	-0.727	-0.519	+0.279	+0.268	+0.238	+0.216	+0.209	+0.137	
.45	-0.441	-0.606	-1.082	-0.883	-0.712	-0.518	+0.263	+0.256	+0.213	+0.187	+0.192	+0.110	
.50	-0.419	-0.573	-1.069	-0.855	-0.704	-0.518	+0.236	+0.225	+0.195	+0.178	+0.160	+0.077	
.55	-0.446	-0.528	-1.025	-0.844	-0.701	-0.516	+0.223	+0.208	+0.180	+0.156	+0.115	+0.077	
.60	-0.435	-0.500	-0.940	-0.803	-0.701	-0.516	+0.200	+0.178	+0.150	+0.125	+0.064	+0.091	
.65	-0.415	-0.472	-0.879	-0.773	-0.687	-0.593	+0.187	+0.158	+0.120	+0.085	+0.061	+0.059	
.68	-0.378	-0.398	-0.820	-0.797	-0.696	-0.498	+0.169	+0.158	+0.120	+0.085	+0.061	+0.016	
.73	-0.325	-0.336	-0.743	-0.758	-0.680	-0.445	+0.145	+0.122	+0.085	+0.052	+0.050	+0.050	
.77	-0.363	-0.763	-0.673	-0.673	-0.484	-0.327	-	-	-	+0.042	+0.021	+0.057	
.80	-0.281	-0.281	-0.760	-0.673	-0.484	-0.316	+0.102	-	-	-	-	-	
.81	-0.583	-0.669	-0.736	-0.667	-0.484	-0.327	-	-	-	+0.047	+0.034	+0.015	
.82	-0.669	-0.669	-0.736	-0.667	-0.484	-0.327	-	-	-	+0.047	+0.034	+0.015	
.85	-0.232	-0.237	-0.671	-0.672	-0.484	-0.327	-	-	-	+0.074	+0.021	+0.013	
.87	-0.225	-0.225	-0.655	-0.727	-0.484	-0.327	-	-	-	+0.042	+0.012	+0.021	
.88	-0.222	-0.213	-0.655	-0.727	-0.484	-0.327	-	-	-	+0.024	-	-	
.90	-0.225	-0.617	-0.654	-0.470	-0.484	-0.327	-	-	-	+0.030	-	-	
.91	-0.214	-0.686	-0.654	-0.470	-0.484	-0.327	-	-	-	+0.018	-	-	
.93	-0.184	-0.563	-0.621	-0.655	-0.484	-0.327	-	-	-	+0.002	-	-	
.95	-0.176	-0.512	-0.655	-0.621	-0.484	-0.327	-	-	-	+0.019	-	-	
.96	-0.141	-0.512	-0.655	-0.621	-0.484	-0.327	-	-	-	-	-	-	
.98	-0.141	-0.512	-0.655	-0.621	-0.484	-0.327	-	-	-	+0.051	-	-	

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TABLE I - Continued
PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 60° DELTA WING

 $M = 0.80$ $\alpha = 19.8$

Upper surface						Lower surface						
Stations, fraction of semispan						Stations, fraction of semispan						
x/c	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90
.00	-0.376	-1.4786	-1.244	-0.985	-0.794							
.01	-0.928	-1.742	-1.260	-0.998	-0.804	-0.585						
.03	-1.402	-1.724	-1.260	-0.995	-0.811	-0.585						
.05	-1.515	-1.902	-1.237	-0.980	-0.809	-0.589						
.08	-1.303	-1.901	-1.235	-0.956	-0.802	-0.593						
.10	-1.131	-1.812	-1.226	-0.951	-0.802	-0.593						
.15	-0.888	-1.857	-1.175	-0.946	-0.808	-0.593						
.19	-0.688	-1.681	-1.130	-0.946	-0.816	-0.594						
.20	-0.743											
.25	-0.675	-1.411	-1.113	-0.940	-0.816	-0.597						
.30	-0.593	-1.216	-1.089	-0.938	-0.816	-0.597						
.35	-0.611	-1.054	-1.080	-0.938	-0.814	-0.599						
.40	-0.612	-0.915	-1.072	-0.940	-0.807	-0.601						
.45	-0.574	-0.746	-1.042	-0.933	-0.799	-0.597						
.50	-0.500	-0.649	-1.010	-0.903	-0.791	-0.597						
.55	-0.452	-0.629	-0.969	-0.886	-0.787							
.60	-0.447	-0.591	-0.905	-0.837	-0.780	-0.585						
.65	-0.447	-0.581	-0.868	-0.808	-0.758							
.68												
.70	-0.437	-0.542	-0.838	-0.823	-0.784	-0.570						
.73												
.75	-0.424	-0.505	-0.787									
.77												
.79												
.80	-0.407	-0.447										
.81												
.82												
.83												
.85	-0.357	-0.372	-0.744									
.87	-0.354	-0.727	-0.759	-0.704								
.88	-0.347											
.89	-0.331											
.90	-0.339	-0.692										
.91	-0.328	-0.723										
.93	-0.281	-0.641										
.95	-0.284	-0.267										
.96												
.97												
.98		-0.589										
	-0.224											
$\alpha = 21.8$												
.00	-0.507	-1.664	-1.124	-0.929	-0.796							
.01	-1.006	-1.646	-1.134	-0.956	-0.829	-0.640						
.03	-1.628	-1.631	-1.134	-0.963	-0.840	-0.640						
.05	-1.652	-1.637	-1.134	-0.963	-0.840	-0.643						
.08	-1.481	-1.644	-1.138	-0.965	-0.844	-0.647						
.10	-1.292	-1.629	-1.138	-0.971	-0.848	-0.647						
.15	-1.003	-1.622	-1.149	-0.987	-0.858	-0.651						
.19	-1.590	-1.156	-0.991	-0.866	-0.655							
.20	-0.842											
.25	-0.782	-1.457	-1.167	-0.986	-0.863	-0.653						
.30	-0.724	-1.290	-1.107	-0.978	-0.860	-0.654						
.35	-0.678	-1.040	-1.07	-0.968	-0.852	-0.644						
.40	-0.613	-1.013	-1.083	-0.961	-0.853	-0.634						
.45	-0.625	-0.918	-1.080	-0.947	-0.820	-0.650						
.50	-0.613	-0.846	-1.023	-0.914	-0.820	-0.647						
.55	-0.613	-0.792	-0.986	-0.901	-0.816							
.60	-0.621	-0.722	-0.930	-0.836	-0.808	-0.636						
.65	-0.597	-0.696	-0.899	-0.838	-0.789							
.68												
.70	-0.561	-0.633	-0.871	-0.848	-0.783	-0.625						
.73												
.75	-0.526	-0.583	-0.824									
.77												
.79												
.80	-0.503	-0.520										
.81												
.82												
.83												
.85	-0.455	-0.439	-0.792									
.87	-0.429	-0.772	-0.788	-0.741								
.88	-0.411											
.89	-0.393											
.90	-0.426	-0.744										
.91	-0.392											
.95	-0.355	-0.336	-0.688									
.96	-0.324											
.97	-0.279	-0.640										
.98												

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TABLE L - Continued

PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 60° DELTA WING

 $\alpha = 0.80$ $\alpha = 23.9$

Upper surface

Stations, fraction of semispan					
x/c	0.19	0.32	0.53	0.65	0.74
.00	-0.607	-1.523	-1.134	-0.941	-0.807
.01	-1.149	-1.521	-1.143	-0.970	-0.839
.03	-1.609	-1.525	-1.143	-0.980	-0.850
.05	-1.531	-1.510	-1.143	-0.976	-0.846
.08	-1.435	-1.519	-1.145	-0.976	-0.848
.10	-1.341	-1.517	-1.147	-0.981	-0.854
.15	-1.151	-1.510	-1.160	-0.996	-0.872
.19	-1.1475	-1.147	-1.005	-0.878	-0.874
.20	-0.948				
.25	-0.854	-1.421	-1.181	-1.006	-0.878
.30	-0.777	-1.500	-1.155	-1.004	-0.878
.35	-0.747	-1.173	-1.124	-0.958	-0.874
.40	-0.714	-1.057	-1.102	-0.990	-0.865
.45	-0.744	-0.956	-1.065	-0.978	-0.857
.50	-0.734	-0.879	-1.035	-0.946	-0.848
.55	-0.727	-0.818	-0.996	-0.932	-0.842
.60	-0.714	-0.781	-0.933	-0.884	-0.833
.65	-0.681	-0.732	-0.902	-0.863	-0.815
.70	-0.649	-0.680	-0.872	-0.874	-0.810
.73	-0.626	-0.635	-0.830	-0.790	-0.790
.75	-0.613	-0.563	-0.838	-0.838	-0.838
.80	-0.581	-0.801	-0.836	-0.779	-0.647
.82	-0.789	-0.814			
.85	-0.550	-0.468	-0.799	-0.766	
.87	-0.442	-0.787	-0.807		
.88	-0.433				
.89	-0.411				
.90	-0.477	-0.756	-0.761	-0.749	-0.638
.91	-0.407				
.92	-0.347	-0.697			
.95	-0.403	-0.342	-0.715		
.96	-0.648				
.97	-0.306				

Lower surface

Stations, fraction of semispan					
x/c	0.19	0.32	0.53	0.65	0.74
.00	.279	.268	.141	.051	-.029
.01	.575	.466	.299	.210	.120
.05	.747	.561	.393	.333	.255
.10	.716	.571	.436	.377	.308
.15	.666	.576	.454	.404	.334
.19	.609	.584	.457	.406	.359
.20	.523	.441	.393	.375	.229
.25	.532	.491	.416	.381	.341
.30	.493	.470	.397	.358	.328
.35	.466	.441	.378	.334	.310
.40	.437	.418	.355	.308	.277
.45	.394	.385	.352	.289	.263
.50	.378	.363	.304	.256	.246
.55	.345	.352	.277	.245	.211
.60	.323	.307	.258	.218	.169
.65	.291	.272	.226	.184	.136
.70	.255	.241	.189	.130	.108
.73	.220	.198	.136		.087
.75					.089
.77					.077
.80					.052
.81					.060
.82					.083
.83					.070
.85					.045
.87					.009
.88					.005
.89					.051
.90					.054
.91					-.060
.92					-.183
.93					.053
.95					.171
.96					.131
.97					.138
.98					.098

 $\alpha = 25.9$

Stations, fraction of semispan					
x/c	0.19	0.32	0.53	0.65	0.74
.00	-0.741	-1.545	-1.15	-0.958	-0.834
.01	-1.344	-1.532	-1.159	-0.950	-0.876
.03	-1.639	-1.566	-1.189	-1.001	-0.887
.05	-0.855	-1.543	-1.185	-0.996	-0.886
.08	-1.475	-1.554	-1.198	-0.999	-0.886
.10	-1.382	-1.559	-1.197	-1.009	-0.894
.15	-1.187	-1.554	-1.209	-1.034	-0.905
.19	-1.512	-1.216	-1.046	-0.925	-0.698
.20	-0.996				.594
.25	-0.919	-1.435	-1.236	-1.053	-0.925
.30	-0.759	-1.295	-1.206	-1.053	-0.925
.35	-0.613	-1.156	-1.177	-1.045	-0.921
.40	-0.779	-1.037	-1.149	-1.034	-0.912
.45	-0.803	-0.949	-1.108	-1.018	-0.905
.50	-0.794	-0.873	-1.072	-0.980	-0.895
.55	-0.770	-0.833	-1.054	-0.962	-0.888
.60	-0.759	-0.782	-0.959	-0.911	-0.879
.65	-0.748	-0.769	-0.926	-0.889	-0.857
.68	-0.735	-0.703	-0.897	-0.899	-0.852
.73	-0.710	-0.640	-0.852	-0.861	-0.827
.75	-0.668	-0.558	-0.861	-0.861	-0.827
.77	-0.618	-0.409	-0.780	-0.774	-0.686
.79	-0.581	-0.360	-0.715	-0.733	-0.637
.80	-0.535	-0.337	-0.652	-0.737	-0.590
.81					
.82					
.83					
.85	-0.586	-0.464	-0.824	-0.796	
.87	-0.437	-0.815	-0.821		
.88	-0.429				
.89	-0.407				
.90	-0.518				
.91	-0.409				
.93	-0.360				
.95	-0.431	-0.367			
.96	-0.477				
.98	-0.535				

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TABLE L- Continued

PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 60° DELTA WING

$\chi = 0.85$

a = 2,3

TABLE L- Continued

PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 80° DELTA WING

$M = 0.85$

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TABLE L- Continued
PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 60° DELTA WING

 $M = 0.85$ $\alpha = 8.3$

x/c	Upper surface						Lower surface					
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90
.00	-.109	-1.020	-.857	-.865	-.756	-.514	.340	.324	.276	.240	.211	.133
.01	-.264	-.843	-.751	-.875	-.756	-.514	.340	.314	.281	.274	.243	.215
.03	-.311	-.889	-.767	-.850	-.764	-.503	.336	.297	.274	.280	.262	.244
.05	-.346	-1.061	-.781	-.793	-.766	-.497	.307	.258	.250	.253	.247	.241
.08	-.357	-1.094	-.822	-.775	-.766	-.490	.272	.245	.235	.240	.227	.230
.10	-.323	-.979	-.839	-.772	-.768	-.486	.237	.208	.202	.202	.207	.201
.15	-.316	-.274	-.913	-.802	-.778	-.472	.178	.174	.175	.204	.194	
.19	-.286	-.935	-.821	-.794	-.663							
.20	-.290						.207					
.25	-.276	-.237	-.968	-.850	-.802	-.448	.192	.150	.156	.158	.162	
.30	-.272	-.277	-.962	-.866	-.828	-.436	.165	.137	.132	.130	.140	.143
.35	-.285	-.293	-.888	-.865	-.846	-.420	.151	.115	.114	.110	.121	.132
.40	-.285	-.277	-.849	-.846	-.853	-.414	.125	.103	.101	.090	.099	.093
.45	-.310	-.270	-.613	-.849	-.861	-.398	.091	.083	.076	.074	.071	.076
.50	-.272	-.278	-.416	-.832	-.873	-.385	.090	.077	.068	.057	.080	.061
.55	-.244	-.278	-.234	-.794	-.876		.072	.062	.057	.059	.057	
.60	-.264	-.251	-.114	-.696	-.848	-.347	.072	.056	.057	.053	.029	.034
.65	-.239	-.235	-.110	-.549	-.806		.059	.040	.045	.039	-.001	
.68											.025	
.70	-.208	-.195	-.101	-.523	-.799	-.316	.042	.037	.032	.017	.005	-.006
.73											.055	
.75	-.177	-.161	-.063				.039	.019	.017		.016	
.77											.012	
.79											.014	
.80	-.147	-.120					.025	.023			.016	-.052
.81											.014	
.82											.017	
.83											.017	
.85	-.104	-.086	-.015	-.157			.014	-.005	-.017		.006	
.87											.005	
.88											.020	
.89											.015	
.90	-.099											
.91												
.93												
.95	-.060	-.032	-.039									
.96												
.97												
.98												

 $\alpha = 10.3$

x/c	0.06	-1.180	-.926	-.935	-.852	-.555	.396	.332	.278	.211	.177	.091
.01	-.366	-.995	-.901	-.943	-.863	-.540	.403	.349	.296	.275	.235	.183
.03	-.388	-1.055	-.907	-.943	-.869	-.540	.409	.339	.312	.298	.274	.235
.05	-.468	-1.116	-.910	-.918	-.866	-.537	.379	.307	.294	.278	.271	.247
.08	-.485	-1.271	-.947	-.905	-.860	-.534	.338	.290	.282	.274	.256	.244
.10	-.422	-1.343	-.974	-.894	-.859	-.527	.302	.253	.252	.239	.242	.223
.15	-.401	-.897	-1.017	-.921	-.866	-.514	.222	.223	.212	.238	.214	
.19	-.610	-1.066	-.943	-.895	-.497							
.20	-.352						.265					
.25	-.369	-.310	-.182	-.955	-.887	-.480	.245	.193	.196	.191	.193	.183
.30	-.324	-.317	-.208	-.977	-.888	-.466	.215	.175	.181	.167	.179	.168
.35	-.344	-.353	-.227	-.1019	-.880	-.457	.200	.155	.160	.142	.155	.160
.40	-.311	-.331	-.180	-.1072	-.881	-.454	.178	.140	.146	.124	.133	.116
.45	-.363	-.324	-.1014	-.1010	-.895	-.447	.139	.120	.129	.109	.122	.102
.50	-.330	-.320	-.795	-.1086	-.932	-.439	.134	.113	.111	.084	.110	.084
.55	-.308	-.326	-.545	-.1095	-.976		.114	.096	.100	.082	.087	
.60	-.308	-.303	-.283	-.1027	-.1027	-.411	.111	.087	.096	.078	.056	.054
.65	-.283	-.281	-.201	-.934	-.1038		.096	.068	.080	.061	.021	
.68											.046	
.70	-.244	-.228	-.124	-.658	-.1084	-.367	.079	.064	.068	.034	.028	.012
.75	-.198	-.185	-.068		-.1097		.071	.044	.048		.028	
.77											.029	
.79											.029	
.80	-.161	-.132					.056	.044			.027	-.047
.81											.024	
.82											.045	
.83											.042	
.85	-.112	-.095	-.001	-.442			.040	.011	.041		.013	
.87											.020	
.88											.016	
.89											.014	
.90	-.098											
.91												
.93												
.95	-.057	-.023										
.96												
.97												
.98												

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TABLE I.- Continued
PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 80° DELTA WING

 $\alpha = 0.85$ $\alpha = 12.4$

x/c	Upper surface					Lower surface				
	0.19	0.32	0.53	0.85	0.90	0.19	0.32	0.53	0.85	0.90
.00	+.050	-1.228	-0.988	-0.999	-1.01	+.437	.340	.267	.193	.178
.01	+.489	-1.176	-1.006	-1.020	-0.99	+.437	.340	.267	.193	.178
.03	+.519	-1.192	-1.010	-1.024	-0.989	+.456	.385	.317	.251	.247
.05	+.590	-1.252	-1.009	-1.002	-0.968	+.456	.385	.336	.220	.305
.08	+.616	-1.364	-1.039	-0.982	-0.939	+.474	.362	.320	.305	.212
.10	+.598	-1.479	-1.079	-0.963	-0.912	+.488	.353	.330	.311	.303
.15	+.470	-1.360	-1.129	-0.972	-0.976	+.404	.344	.321	.309	.294
.19	+.477	-1.129	-0.975	-0.970	-0.941	+.366	.305	.294	.279	.275
.20	+.429	-1.433	-1.140	-0.922	-0.557	+.499	.324	.265	.252	.201
.30	+.391	-1.574	-1.176	-0.878	-0.548	+.503	.297	.243	.232	.224
.35	+.404	-1.595	-1.192	-0.797	-0.532	+.507	.264	.224	.206	.205
.40	+.565	-1.590	-1.366	-0.720	-0.509	+.514	.243	.202	.198	.180
.45	+.409	-1.357	-1.211	-0.704	-0.514	+.514	.213	.186	.156	.152
.50	+.366	-1.316	-0.989	-0.709	-0.533	+.514	.180	.163	.164	.139
.55	+.328	-1.283	-0.724	-0.718	-0.575	+.514	.167	.153	.143	.117
.60	+.286	-1.220	-0.415	-0.693	-0.626	+.506	.146	.130	.126	.105
.65	+.232	-1.179	-0.311	-0.650	-0.642	+.506	.123	.096	.094	.089
.68	+.232	-1.179	-0.311	-0.650	-0.663	+.506	.114	.091	.084	.035
.70	+.189	-1.198	-1.192	-0.692	-0.670	+.486	.100	.085	.072	.050
.73	+.159	-1.196	-1.240	-0.664	-0.638	+.486	.086	.059	.039	.011
.75	+.159	-1.196	-1.240	-0.664	-0.638	+.486	.086	.059	.039	.011
.80	+.159	-1.148	-0.666	-0.658	-0.476	+.464	.050	+.008	+.003	+.003
.81	+.329	-1.337	-0.638	+.666	+.666	+.006	+.006	+.006	+.006	+.006
.83	+.337	-1.337	-0.638	+.666	+.666	+.019	+.013	+.012	+.012	+.012
.85	+.157	+.166	-1.372	-0.642	+.605	+.039	+.003	+.006	+.003	+.029
.87	+.173	+.173	-1.361	-0.605	+.605	+.004	+.003	+.035	+.003	+.003
.88	+.175	+.175	-1.361	-0.605	+.605	+.006	+.006	+.006	+.006	+.006
.89	+.171	+.171	-1.361	-0.605	+.605	+.006	+.006	+.006	+.006	+.006
.90	+.186	+.186	-1.398	-0.617	-0.667	+.007	+.007	+.012	+.012	+.163
.91	+.179	+.179	-1.310	-0.561	+.561	+.030	+.013	+.064	+.064	+.064
.93	+.161	+.161	-1.310	-0.561	+.561	+.042	+.042	+.039	+.039	+.151
.95	+.150	+.150	-1.280	-0.514	+.514	+.060	+.060	+.118	+.118	+.118
.97	+.115	+.115	-1.280	-0.514	+.514	+.059	+.059	+.059	+.059	+.059

 $\alpha = 14.3$

.00	+.022	-1.340	-1.190	-1.040	-0.649	+.385	.335	.237	.163	.148
.01	+.608	-1.330	-1.204	-1.054	-0.646	+.572	.301	.214	.198	.179
.03	+.609	-1.265	-1.054	-0.644	-0.566	+.566	.282	.247	.228	.107
.05	+.860	-1.394	-1.115	-1.024	-0.629	+.566	.235	.228	.347	.295
.08	+.747	-1.347	-1.224	-0.991	-0.615	+.570	.495	.405	.349	.312
.10	+.694	-1.362	-1.207	-0.961	-0.601	+.549	.469	.394	.344	.339
.15	+.581	-1.381	-1.210	-0.908	-0.588	+.549	.410	.258	.219	.210
.19	+.221	-1.224	-0.873	-0.593	-0.565	+.324	.295	.288	.294	.210
.20	+.513	-1.750	-1.229	-0.822	-0.595	+.367	+.019	+.019	+.019	+.019
.25	+.517	-1.750	-1.229	-0.822	-0.595	+.367	+.019	+.019	+.019	+.019
.30	+.478	-1.565	-1.256	-0.704	-0.624	+.556	.308	.249	.239	.163
.35	+.481	-1.509	-1.098	-0.704	-0.624	+.556	.282	.247	.228	.022
.40	+.432	-1.500	-1.086	-0.679	-0.647	+.555	.257	.230	.183	.115
.45	+.462	-1.432	-1.121	-0.690	-0.666	+.550	.212	.204	.183	.167
.50	+.411	-1.353	-1.114	-0.723	-0.686	+.545	.204	.190	.162	.139
.55	+.356	-1.282	-0.967	-0.769	-0.705	+.545	.181	.166	.143	.132
.60	+.297	-1.193	-0.791	-0.767	-0.725	+.532	.169	.153	.129	.110
.65	+.247	-1.164	-0.694	-0.744	-0.712	+.495	.128	.105	.087	.020
.68	+.247	-1.164	-0.694	-0.744	-0.712	+.495	.128	.105	.087	.020
.70	+.200	-1.165	-0.669	-0.785	-0.711	-0.520	+.121	.114	.080	.045
.73	+.191	-1.221	-0.642	-0.748	+.689	+.100	.080	.040	+.016	+.023
.77	+.235	-1.277	-0.748	-0.748	+.673	+.077	.065	+.013	+.004	+.067
.80	+.235	-1.277	-0.748	-0.748	+.673	+.077	.065	+.004	+.004	+.067
.81	+.606	-1.598	-0.715	-0.748	+.673	+.077	.065	+.003	+.003	+.067
.85	+.255	-1.296	-0.616	-0.654	+.654	+.040	+.002	+.002	+.002	+.026
.87	+.255	-1.296	-0.593	-0.694	+.654	+.040	+.002	+.006	+.006	+.026
.88	+.292	-1.292	-0.654	-0.694	+.654	+.002	+.002	+.007	+.007	+.026
.89	+.284	-1.284	-0.654	-0.694	+.654	+.001	+.028	+.028	+.028	+.169
.90	+.295	-1.533	-0.647	-0.634	-0.492	+.001	+.016	+.016	+.016	+.169
.91	+.289	-1.291	-0.647	-0.634	-0.492	+.001	+.016	+.016	+.016	+.169
.93	+.251	-1.466	-1.216	-0.595	+.192	+.054	+.052	+.055	+.055	+.159
.95	+.222	-1.416	-1.416	-1.416	-1.416	+.077	+.100	+.133	+.133	+.133
.96	+.171	-1.171	-1.171	-1.171	-1.171	+.077	+.077	+.077	+.077	+.077

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TABLE I.- Continued
PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 60° DELTA WING

$$\star = 0.85$$

a = 16.4

Upper surface

Stations, fraction of semipan						
x/c	0.19	0.32	0.55	0.65	0.74	0.90
.00	-1.103	-1.466	-1.314	-0.897	-0.692	
.01	-0.656	-1.148	-1.321	-0.966	-0.697	-0.626
.03	-0.846	-1.401	-1.326	-0.898	-0.701	-0.619
.05	-1.025	-1.562	-1.323	-0.875	-0.700	-0.618
.08	-0.914	-1.668	-1.328	-0.845	-0.692	-0.611
.10	-0.790	-1.602	-1.303	-0.830	-0.692	-0.616
.15	-0.662	-1.685	-1.244	-0.809	-0.697	-0.638
.19	-0.646	-1.464	-1.206	-0.797	-0.707	-0.605
.20	-0.592					
.25	-0.575	-1.087	-1.131	-0.781	-0.714	-0.596
.30	-0.547	-0.848	-1.057	-0.759	-0.725	-0.589
.35	-0.560	-0.723	-0.975	-0.744	-0.737	-0.581
.40	-0.515	-0.619	-0.949	-0.768	-0.749	-0.581
.45	-0.525	-0.479	-0.928	-0.817	-0.746	-0.575
.50	-0.440	-0.753	-0.944	-0.867	-0.749	-0.568
.55	-0.433	-0.291	-0.925	-0.861	-0.744	-0.565
.60	-0.274	-0.236	-0.884	-0.833	-0.767	-0.557
.65	-0.239	-0.282	-0.886	-0.809	-0.751	
.68					-0.751	
.70	-0.259	-0.382	-0.848	-0.832	-0.744	-0.544
.73					-0.732	
.75	-0.340	-0.477	-0.790		-0.721	
.77				-0.797		
.79				-0.791		
.80	-0.440	-0.501			-0.703	-0.536
.81				-0.788		
.82			-0.735			
.83			-0.720	-0.762		
.85	-0.424	-0.432	-0.724		-0.686	
.87		-0.397	-0.701	-0.746		
.88		-0.376				
.89		-0.358				
.90	-0.378		-0.648		-0.645	-0.528
.91		-0.345		-0.709		
.93		-0.297	-0.582			
.95	-0.259	-0.262			-0.630	
.96				1.188		
.97				-0.529		
.98		-0.204				

a = 17.7

.00	=.171	=.1553	=.1302	=.919	=.885									
.01	=.735	=.1514	=.1299	=.946	=.895	=.583	=.385	=.326	=.213	=.126	=.059	=.064		
.03	=.995	=.1482	=.1292	=.959	=.897	=.580	=.560	=.448	=.300	=.262	=.180	=.051		
.05	=.1181	=.1425	=.1258	=.953	=.891	=.580	=.639	=.490	=.380	=.342	=.278	=.153		
.08	=.1031	=.1741	=.1222	=.948	=.876	=.576	=.608	=.475	=.394	=.359	=.310	=.212		
.10	=.899	=.1641	=.1173	=.946	=.864	=.576	=.565	=.471	=.398	=.371	=.318	=.224		
.15	=.734	=.1720	=.1179	=.959	=.850	=.572	=.501	=.435	=.384	=.354	=.324	=.227		
.19	=.562	=.1562	=.1121	=.965	=.841	=.567		=.405	=.360	=.334	=.330	=.233		
.20	=.549						=.451							
.25	=.526	=.1247	=.1070	=.959	=.824	=.563	=.423	=.371	=.334	=.316	=.291	=.210		
.30	=.397	=.005	=.1071	=.942	=.808	=.560	=.383	=.349	=.309	=.290	=.274	=.196		
.35	=.613	=.793	=.1059	=.925	=.793	=.557	=.360	=.324	=.288	=.265	=.251	=.164		
.40	=.556	=.624	=.1062	=.923	=.778	=.556	=.330	=.306	=.269	=.237	=.223	=.152		
.45	=.519	=.590	=.1047	=.917	=.764	=.554	=.284	=.274	=.246	=.221	=.208	=.138		
.50	=.423	=.211	=.031	=.888	=.751	=.550	=.211	=.252	=.224	=.192	=.194	=.119		
.55	=.405	=.486	=.1010	=.872	=.747		=.252	=.230	=.203	=.184	=.184	=.156		
.60	=.389	=.941	=.950	=.822	=.741	=.538	=.237	=.215	=.188	=.161	=.155	=.082		
.65	=.432	=.806	=.897	=.794	=.727		=.209	=.185	=.159	=.136	=.062			
.68					=.727									
.70	=.480	=.555	=.843	=.804	=.725	=.530	=.179	=.166	=.127	=.087	=.069	=.026		
.73					=.718									
.75	=.470	=.452	=.771		=.761	=.707	=.152	=.131	=.090		=.059		=.057	
.77					=.761									
.79					=.761									
.80	=.385	=.371			=.699	=.518	=.122	=.107			=.055		=.045	
.81					=.759									
.82					=.717									
.83					=.699	=.737								
.85	=.297	=.306	=.702		=.690		=.081	=.035	=.031					
.87		=.291	=.679	=.727				=.043	=.024		=.005			
.88		=.281						=.031						
.89		=.268						=.027						
.90	=.260		=.645		=.675	=.511	=.037		=.005		=.051		=.152	
.91		=.266		=.694				=.014			=.050			
.93		=.233		=.595				=.005		=.050				
.95	=.212	=.215			=.641		=.023	=.032			=.127		=.143	
.96					=.668									
.97					=.546						=.108			
.98					=.178						=.063			

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TABLE L - Continued
PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 60° DELTA WING

 $M = 0.85$ $\alpha = 19.9$

Upper surface							Lower surface						
Stations, fraction of semispan							Stations, fraction of semispan						
x/c	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	
.00	-1.291	-1.616	-1.143	-0.987	-0.824		.362	.325	.201	.098	.035	-0.117	
.01	-0.785	-1.614	-1.154	-1.009	-0.834	-0.622	.385	.467	.291	.251	.166	.007	
.03	-1.239	-1.598	-1.160	-1.007	-0.839	-0.621	.495	.525	.391	.346	.270	.118	
.05	-1.359	-1.653	-1.146	-0.992	-0.828	-0.621	.558	.515	.411	.370	.312	.187	
.08	-1.197	-1.748	-1.152	-0.971	-0.832	-0.623	.611	.515	.418	.387	.324	.208	
.10	-1.056	-1.684	-1.158	-0.968	-0.828	-0.623	.645	.580	.409	.378	.334	.218	
.15	-0.894	-1.680	-1.122	-0.963	-0.828	-0.621	.654	.588	.409	.378	.334	.218	
.19	-0.894	-1.597	-1.099	-0.957	-0.833	-0.621	.654	.588	.359	.344	.228		
.20	-0.716						.492						
.25	-0.666	-1.349	-1.101	-0.946	-0.824	-0.621	.464	.420	.363	.341	.306	.205	
.30	-0.591	-1.178	-1.079	-0.931	-0.822	-0.620	.426	.398	.340	.318	.289	.190	
.35	-0.588	-1.053	-1.061	-0.923	-0.819	-0.621	.402	.372	.319	.290	.265	.093	
.40	-0.600	-0.937	-1.049	-0.922	-0.810	-0.621	.375	.347	.299	.263	.236	.153	
.45	-0.634	-0.801	-1.021	-0.912	-0.802	-0.618	.329	.318	.277	.245	.226	.140	
.50	-0.575	-0.717	-0.995	-0.881	-0.794	-0.614	.313	.300	.252	.217	.209	.119	
.55	-0.498	-0.719	-0.988	-0.868	-0.788		.284	.271	.227	.206	.174		
.60	-0.487	-0.677	-0.898	-0.821	-0.784	-0.604	.266	.249	.209	.181	.129	.084	
.65	-0.520	-0.636	-0.873	-0.804	-0.764		.240	.216	.178	.150	.070		
.68												.104	
.70	-0.497	-0.574	-0.841	-0.817	-0.760	-0.593	.208	.197	.147	.101	.079	.027	
.73												.095	
.75	-0.464	-0.542	-0.791		-0.742		.179	.154	.099			.067	
.77					-0.790							.065	
.79					-0.787							.058	
.80	-0.446	-0.487			-0.731	-0.584	.146	.131				.045	
.81					-0.787								
.82					-0.765							.058	
.83					-0.756	-0.769						.039	
.85	-0.391	-0.416	-0.766		-0.719		.092	.059	.054			.004	
.87	-0.399	-0.754	-0.762					.048	.025	.001			
.88	-0.396							.036					
.89	-0.382							.030					
.90	-0.364		-0.720		-0.703	-0.579	.032		.005		-0.054	-0.156	
.91		-0.986		-0.730				.019			-0.046		
.93	-0.349	-0.671						.008	-0.048				
.95	-0.323	-0.338			-0.671			.006	.042			-0.142	
.96					-0.704								
.97					-0.632								
.98					-0.292								
								.090					

 $\alpha = 21.6$

Upper surface							Lower surface						
Stations, fraction of semispan							Stations, fraction of semispan						
x/c	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	
.00	-0.401	-1.521	-1.093	-0.937	-0.819		.347	.218	.196	.087	.014	-0.163	
.01	-0.830	-1.506	-1.103	-0.960	-0.840	-0.663	.605	.484	.305	.246	.152	-0.027	
.03	-1.420	-1.493	-1.098	-0.967	-0.848	-0.660	.746	.552	.415	.350	.266	.093	
.05	-1.456	-1.511	-1.098	-0.963	-0.846	-0.664	.703	.552	.436	.384	.314	.169	
.08	-1.331	-1.511	-1.096	-0.962	-0.846	-0.665	.652	.552	.447	.402	.336	.199	
.10	-1.172	-1.479	-1.096	-0.963	-0.848	-0.665	.590	.525	.440	.399	.351	.214	
.15	-0.931	-1.471	-1.096	-0.969	-0.854	-0.665	.495	.421	.380	.363	.229		
.19	-1.444	-1.094	-0.971	-0.859	-0.665		.536						
.20	-0.787						.505	.463	.399	.349	.325	.211	
.25	-0.709	-1.346	-1.107	-0.965	-0.853	-0.665	.465	.437	.379	.348	.311	.195	
.30	-0.671	-1.219	-1.079	-0.958	-0.852	-0.666	.443	.413	.355	.317	.288		
.35	-0.682	-1.102	-1.059	-0.947	-0.849	-0.668	.414	.388	.332	.290	.262	.164	
.40	-0.638	-1.004	-1.048	-0.939	-0.837	-0.671	.367	.357	.309	.274	.247	.152	
.45	-0.647	-0.912	-1.021	-0.930	-0.827	-0.666	.354	.349	.284	.258	.220	.130	
.50	-0.629	-0.849	-0.999	-0.899	-0.822	-0.665	.323	.307	.258	.229	.195		
.85	-0.621	-0.816	-0.966	-0.889	-0.814		.304	.283	.238	.205	.170	.095	
.60	-0.644	-0.755	-0.914	-0.845	-0.808	-0.655	.271	.247	.209	.172	.088		
.65	-0.649	-0.730	-0.892	-0.836	-0.789							.127	
.68													
.70	-0.615	-0.672	-0.866	-0.845	-0.758	-0.644	.237	.222	.174	.123	.098	.039	
.73													
.75	-0.576	-0.627	-0.821		-0.770		.205	.180	.121			.085	
.77					-0.821								
.79					-0.820								
.80	-0.552	-0.570			-0.763	-0.638	.168	.149			.057	-0.032	
.81					-0.820							.061	
.82					-0.793								
.83					-0.790	-0.804							
.85	-0.497	-0.489	-0.803		-0.753		.108	.054	.049			.021	
.87	-0.470	-0.790	-0.798					.062	.0743	.016			
.88	-0.464							.049					
.89	-0.448							.043					
.90	-0.457		-0.759		-0.737	-0.627	.047		.011		-0.039	-0.148	
.91	-0.451		-0.765					.028		-0.033			
.93	-0.410	-0.705						.001	-0.038				
.95	-0.409	-0.404			-0.706			.026	-0.042			-0.131	
.96					-0.738								
.97					-0.660								
.98					-0.357								
								.102	-0.113				

TABLE I.- Continued
PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 80° DELTA WING

 $M = 0.85$ $\alpha = 23.9$

Upper surface						Lower surface						
Stations, fraction of semispan						Stations, fraction of semispan						
x/c	0.19	0.32	0.53	0.65	0.74	0.80	0.19	0.32	0.53	0.65	0.74	0.80
.00	-1.506	-1.352	-1.118	-0.993	-0.880		.316	.319	.175	.054	-0.020	-0.215
.01	-1.578	-1.354	-1.128	-1.014	-0.899	-0.723	.611	.498	.320	.228	.128	-0.073
.03	-1.413	-1.354	-1.128	-1.020	-0.904	-0.718	.778	.581	.413	.342	.258	-0.046
.05	-1.350	-1.349	-1.128	-1.020	-0.904	-0.720	.741	.590	.447	.383	.315	-0.153
.08	-1.271	-1.362	-1.127	-1.006	-0.900	-0.720	.686	.593	.463	.411	.341	-0.187
.10	-1.206	-1.350	-1.128	-1.014	-0.901	-0.720	.629	.567	.464	.411	.363	-0.212
.15	-1.046	-1.348	-1.131	-1.016	-0.904	-0.720	.540	.445	.398	.376	.228	
.19	-1.046	-1.324	-1.132	-1.022	-0.909	-0.721						
.20	-0.898						.576					
.25	-0.819	-1.296	-1.144	-1.021	-0.904	-0.721	.549	.509	.424	.389	.345	-0.215
.30	-0.766	-1.221	-1.120	-1.014	-0.907	-0.722	.508	.485	.406	.385	.334	-0.202
.35	-0.746	-1.141	-1.104	-1.009	-0.909	-0.723	.483	.455	.382	.341	.313	-0.011
.40	-0.722	-1.057	-1.093	-1.005	-0.898	-0.724	.454	.431	.362	.314	.282	-0.176
.45	-0.756	-0.980	-1.073	-0.993	-0.893	-0.721	.404	.399	.339	.297	.272	-0.166
.50	-0.759	-0.911	-1.052	-0.962	-0.886	-0.717	.386	.378	.312	.266	.253	-0.145
.55	-0.759	-0.888	-1.020	-0.956	-0.880		.358	.344	.286	.253	.216	
.60	-0.759	-0.807	-0.869	-0.806	-0.873	-0.711	.337	.320	.266	.228	.171	-0.106
.65	-0.737	-0.786	-0.941	-0.896	-0.855		.308	.286	.228	.194	.111	
.68												-0.147
.70	-0.700	-0.734	-0.910	-0.909	-0.854	-0.703	.267	.258	.196	.139	.119	-0.053
.73												-0.155
.75	-0.670	-0.691	-0.864		-0.853		.237	.214	.151			-0.102
.77												-0.099
.79												-0.092
.80	-0.556	-0.632			-0.826	-0.693	.195	.180				-0.073 -0.023
.81												
.82												
.83												
.85	-0.609	-0.535	-0.811	-0.856		-0.813	.130	.075	.072	.062		
.87								.086	.058	.024		
.88												
.89												
.90	-0.524	-0.810			-0.796	-0.654	.066					
.91	-0.462				-0.808		.051					
.93	-0.409	-0.749					.018					
.95	-0.432	-0.406					-0.026	-0.023				
.96												-0.122
.97												-0.109
.98	-0.370	-0.689										-0.091

 $\alpha = 26.0$

Upper surface						Lower surface						
Stations, fraction of semispan						Stations, fraction of semispan						
x/c	0.19	0.32	0.53	0.65	0.74	0.80	0.19	0.32	0.53	0.65	0.74	0.80
.00	-1.618	-1.372	-1.182	-1.026	-0.907		.286	.310	.158	.030	-0.052	-0.258
.01	-1.173	-1.366	-1.176	-1.052	-0.935	-0.762	.603	.510	.322	.212	.108	-0.105
.03	-1.457	-1.389	-1.176	-1.056	-0.941	-0.753	.798	.654	.421	.337	.211	-0.041
.05	-1.371	-1.371	-1.161	-1.041	-0.941	-0.755	.765	.623	.462	.391	.213	-0.142
.08	-1.307	-1.390	-1.177	-1.048	-0.937	-0.757	.721	.626	.485	.424	.246	-0.181
.10	-1.239	-1.385	-1.182	-1.050	-0.939	-0.757	.662	.608	.492	.451	.277	-0.214
.15	-1.092	-1.383	-1.184	-1.058	-0.945	-0.756	.579	.477	.421	.393	.235	
.19	-1.358	-1.184	-1.066	-0.953	-0.756							
.20							.615					
.25	-0.887	-1.223	-1.197	-1.067	-0.933	-0.758	.583	.551	.459	.414	.371	-0.225
.30	-0.828	-1.231	-1.176	-1.045	-0.956	-0.758	.544	.526	.438	.393	.357	-0.214
.35	-0.806	-1.146	-1.155	-1.058	-0.957	-0.758	.518	.498	.419	.368	.338	-0.033
.40	-0.778	-1.057	-1.143	-1.054	-0.951	-0.760	.491	.474	.398	.347	.313	-0.196
.45	-0.823	-0.971	-1.113	-1.043	-0.942	-0.756	.445	.441	.375	.327	.296	-1.83
.50	-0.827	-0.905	-1.091	-1.007	-0.937	-0.756	.424	.419	.349	.296	.277	-0.163
.55	-0.807	-0.872	-1.110	-0.996	-0.930		.395	.385	.321	.283	.245	
.60	-0.790	-0.828	-0.941	-0.924	-0.747		.372	.360	.300	.258	.200	-0.128
.65	-0.776	-0.818	-0.969	-0.930	-0.897		.339	.322	.264	.224	.137	
.68												-0.173
.70	-0.760	-0.772	-0.932	-0.940	-0.899	-0.736	.303	.292	.227	.166	.144	-0.075
.73												-0.160
.75	-0.749	-0.721	-0.882		-0.871		.264	.244	.172			-0.126
.77												-0.126
.79												
.80	-0.739	-0.640			-0.859	-0.729	.222	.210				-0.094 -0.003
.81												
.82												
.83												
.85	-0.660	-0.521	-0.857		-0.843		.155	.102	.092			-0.055
.87												
.88												
.89												
.90	-0.547	-0.641			-0.823	-0.718	.090					-0.005 -0.124
.91												
.92												
.93												
.95	-0.466	-0.399	-0.783		-0.776		.011	.008				-0.106
.96												
.97												
.98	-0.373	-0.705										-0.092

TABLE I.- Continued

$$\pi = 0.90$$

TABLE L- Continued
PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 60° DELTA WING

 $M = 0.90$ $\alpha = 4.2$

Upper surface							Lower surface						
Stations, fraction of semispan							Stations, fraction of semispan						
x/c	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	
.00	+148	-+225	-+622	-+636	-+720								
.01	-+059	-+414	-+611	-+722	-+672	-+625	+211	+212	+233	+235	+218	+136	
.03	-+094	-+182	-+608	-+547	-+595	-+605	+211	+177	+205	+204	+203	+213	
.05	-+121	-+227	-+610	-+537	-+710	-+605	+185	+151	+162	+182	+179	+202	
.08	-+141	-+228	-+615	-+451	-+719	-+605	+185	+130	+135	+145	+153	+166	
.10	-+132	-+215	-+581	-+422	-+731	-+602	+194	+127	+116	+131	+129	+144	
.15	-+146	-+198	-+422	-+414	-+743	-+593	+106	+091	+080	+091	+099	+115	
.19	-+138	-+275	-+394	-+788	-+585		+067	+055	+066	+097	+108		
.25	-+158	-+204	-+229	-+364	-+648	-+577	+081	+044	+036	+047	+055	+077	
.30	-+150	-+196	-+206	-+354	-+519	-+571	+061	+033	+025	+029	+038	+070	
.35	-+163	-+208	-+209	-+341	-+536	-+566	+050	+016	+006	+007	+022	+054	
.40	-+143	-+204	-+226	-+329	-+254	-+584	+028	+007	+004	+006	+005	+015	
.45	-+207	-+204	-+235	-+321	-+191	-+533	+004	+010	+018	+015	+003	+007	
.50	-+183	-+207	-+237	-+294	-+206	-+542	+002	+012	+029	+032	+003	+002	
.55	-+183	-+217	-+231	-+267	-+184		+012	+024	+032	+026	+019		
.60	-+194	-+205	-+186	-+229	-+168	-+501	+012	+025	+029	+026	+037	+012	
.65	-+187	-+207	-+177	-+175	-+121		+022	+036	+036	+032	+045		
.68	-+125	-+116			-+133					-+035			
.70	-+168	-+177	-+158	-+186	-+125	-+456	+029	+033	+039	+051	+033	+030	
.73	-+152	-+152	-+116		-+100		+025	+043	+052		-+007		
.75	-+152	-+152	-+116		-+076					-+039			
.77				-+138						-+052			
.80	-+134	-+112		-+120			+035	+038			-+014	-+042	
.81				-+113						-+029			
.82				-+078						-+040			
.83				-+047	-+092					-+054	-+026		
.85	-+092	-+080	-+063	-+033	-+070		+035	-+053	-+031		-+006		
.87				-+070				-+043	-+029	-+014			
.88				-+052				-+045					
.89	-+052							-+044					
.90	-+096	-+029			-+001	-+320	+050	-+044	-+021		-+004	-+060	
.91	-+052			-+035				-+044	-+009	-+002			
.93	-+033			-+005				-+044					
.95	-+055	-+026			-+026		+052	-+041			-+016	-+027	
.96				-+017						-+012			
.98				-+006						-+025			

 $\alpha = 6.3$

.00	+147	-+639	-+815	-+874	-+745								
.01	-+127	-+734	-+623	-+809	-+734	-+593	+283	+284	+270	+248	+223	+120	
.03	-+173	-+714	-+635	-+818	-+743	-+579	+283	+257	+259	+245	+229	+209	
.05	-+210	-+738	-+635	-+711	-+748	-+577	+264	+252	+229	+236	+224	+222	
.08	-+227	-+659	-+648	-+673	-+752	-+568	+264	+204	+203	+201	+203	+203	
.10	-+206	-+410	-+651	-+660	-+756	-+561	+209	+187	+187	+188	+180	+188	
.15	-+216	-+171	-+651	-+661	-+779	-+546	+178	+152	+149	+148	+151	+158	
.19			-+647	-+663	-+811	-+533		+128	+120	+117	+148	+147	
.20	-+201							-+155					
.25	-+216	-+226	-+621	-+659	-+812	-+515	+142	+099	+098	+099	+104	+114	
.30	-+204	-+421	-+547	-+650	-+812	-+499	+113	+086	+085	+077	+085	+100	
.35	-+218	-+258	-+455	-+614	-+781	-+482	+104	+089	+088	+085	+085	+089	
.40	-+191	-+253	-+503	-+579	-+743	-+466	+082	+055	+051	+058	+048	+049	
.45	-+264	-+264	-+398	-+519	-+718	-+452	+044	+037	+027	+039	+039	+038	
.50	-+234	-+254	-+346	-+479	-+675	-+436	+049	+034	+024	+008	+039	+027	
.55	-+231	-+270	-+293	-+429	-+609		+033	+020	+016	+012	+015		
.60	-+249	-+262	-+229	-+354	-+543	-+405	+033	+017	+016	+006	-+010	-+007	
.65	-+242	-+254	-+206	-+277	-+448		+022	+002	+006	-+006	-+033		
.68				-+402						-+012			
.70	-+221	-+216	-+170	-+261	-+379	-+376	+007	+000	+000	-+028	-+029	-+023	
.73	-+186	-+180	-+123		-+337		+007	-+011	-+019		-+023	-+011	
.75				-+188						-+019			
.79				-+163						-+019			
.80	-+155	-+129		-+148			+002	-+009			-+007	-+058	
.81				-+074						-+018			
.83				-+060	-+119					-+013			
.85	-+103	-+090	-+059	-+087	-+144		-+007	-+032	-+010		-+016		
.87	-+073	-+045	-+087					-+021	-+008	-+012		-+010	
.88	-+066							-+025					
.89	-+058							-+026					
.90	-+096	-+018		-+092	-+345		-+019		-+001		-+014	-+115	
.91	-+055	-+043						-+028		-+007			
.93	-+036	-+008						-+028	-+006				
.95	-+054	-+026			-+032		-+037	-+027			-+014		
.96				-+012						-+007			
.98	-+016			-+020						-+024			
								-+019					

TABLE L - Continued

PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 60° DELTA WING

 $M = 0.90$

$\alpha = 8.3$							$\alpha = 10.3$						
Upper surface							Lower surface						
Stations, fraction of semispan							Stations, fraction of semispan						
x/c	0.19	0.32	0.53	0.65	0.74	0.90		0.19	0.32	0.53	0.65	0.74	0.90
.00	.137	-.895	-.838	-.865	-.809			.352	.335	.281	.246	.207	.084
.01	-.209	-.811	-.724	-.873	-.812	-.607		.352	.318	.285	.272	.234	.188
.03	-.254	-.852	-.737	-.860	-.821	-.593		.357	.296	.271	.276	.249	.226
.05	-.301	-.978	-.747	-.805	-.824	-.587		.357	.264	.248	.249	.236	.224
.08	-.327	-1.000	-.784	-.782	-.826	-.575		.278	.245	.232	.238	.217	.212
.10	-.287	-.924	-.804	-.779	-.830	-.566		.244	.209	.199	.198	.189	.163
.15	-.298	-.250	-.865	-.799	-.845	-.547		.183	.169	.172	.188	.177	
.19			-.886	-.813	-.869	-.528							
.20	-.266							.214					
.22	-.293	-.262	-.958	-.830	-.886	-.507		.196	.153	.149	.144	.142	
.30	-.259	-.276	-.902	-.840	-.905	-.487		.149	.137	.130	.126	.124	.124
.35	-.277	-.298	-.862	-.845	-.906	-.449		.157	.118	.104	.092	.093	.079
.40	-.238	-.312	-.788	-.819	-.847	-.455		.128	.104	.094	.085	.083	.075
.45	-.320	-.301	-.747	-.845	-.894	-.441		.093	.092	.077	.072	.072	.065
.50	-.301	-.307	-.938	-.810	-.891	-.422		.093	.077	.062	.052	.072	.050
.55	-.289	-.316	-.281	-.794	-.881			.076	.060	.051	.054	.041	
.60	-.303	-.314	-.227	-.709	-.866	-.389		.071	.054	.052	.047	.014	.027
.65	-.293	-.314	-.201	-.600	-.815			.057	.038	.037	.034	-.019	
.68												.008	
.70	-.284	-.279	-.161	-.570	-.798	-.357		.044	.036	.028	.007	-.008	-.009
.73												.024	
.75	-.246	-.224	-.094		-.759			.040	.020	.010			.004
.77													
.79													
.80	-.192	-.149			-.716	-.336		.029	.021			.005	-.053
.81												.010	
.82												.012	
.83												.012	
.85	-.121	-.097	-.024		-.677			.017	-.007	.012			-.002
.87	-.078	-.006	-.160						.004	.013	.011		
.88												.001	
.89												.002	
.90	-.096		.021		-.623	-.326		.003					
.91			.054		-.037							-.019	-.124
.93			.037		.048							.016	
.95	-.053		.019			-.522			-.025	-.008	.024		-.061
.96													
.97													
.98													
$\alpha = 10.3$													
.00	.111	-.1087	-.898	-.923	-.903			.410	.352	.280	.223	.185	.017
.01	-.306	-.926	-.868	-.928	-.913	-.588		.250	.200	.189	.185	.181	.165
.03	-.332	-.980	-.872	-.933	-.922	-.722		.215	.182	.170	.163	.162	.152
.05	-.430	-.1053	-.869	-.907	-.927	-.711		.407	.347	.308	.297	.264	.211
.08	-.437	-.1176	-.899	-.894	-.924	-.693		.407	.315	.288	.277	.259	.227
.10	-.403	-.1235	-.925	-.891	-.918	-.678		.342	.295	.279	.270	.249	.224
.15	-.370	-.860	-.966	-.923	-.934	-.647		.305	.260	.242	.236	.229	.202
.19		-.1023	-.940	-.966	-.621			.229	.217	.206	.226	.201	
.20	-.346							.268					
.25	-.356	-.307	-.131	-.957	-.969	-.588		.250	.200	.189	.185	.181	.165
.30	-.332	-.310	-.131	-.976	-.983	-.564		.215	.182	.170	.163	.162	.152
.35	-.340	-.144	-.1008	-.993	-.538			.199	.160	.147	.136	.141	.150
.40	-.294	-.354	-.1131	-.1041	-.1006	-.520		.174	.145	.134	.119	.120	.108
.45	-.370	-.360	-.1058	-.1066	-.1005	-.504		.135	.122	.116	.105	.109	.097
.50	-.353	-.365	-.921	-.1052	-.1005	-.483		.152	.113	.099	.083	.107	.081
.55	-.356	-.379	-.748	-.1064	-.1002			.111	.096	.087	.081	.074	
.60	-.372	-.388	-.523	-.993	-.1010	-.449		.104	.088	.085	.074	.045	.055
.65	-.369	-.399	-.404	-.958	-.993			.088	.069	.070	.056	.006	
.70	-.364	-.377	-.530	-.907	-.1005	-.416		.072	.065	.056	.026	.016	.015
.73	-.343	-.439	-.186			-.1005		.066	.044	.033			.027
.77													
.79													
.80	-.278	-.219			-.1011	-.387		.047	.041			.023	-.037
.81												.021	
.82												.031	
.83												.031	
.85	-.162	-.111	-.001			-.1009		.036	.008	.029		.018	
.87			.084	.018		-.490		.020		.029		.018	
.88			.074						.016				
.89			.059						.014				
.90	-.105		.052			-.885	-.365		.014		.032		-.015
.91			.055							.010		.018	
.93			.032			-.255				.005	.039		
.95	-.067		.018			-.734				.015	.001		-.075
.96						-.030						.027	
.97						.083					.004	.061	
.98													

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TABLE I.- Continued

PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 60° DELTA WING

 $\alpha = 0.00$ $\alpha = 12.3$

Upper surface							Lower surface						
Stations, fraction of semispan							Stations, fraction of semispan						
x/c	0.19	0.32	0.53	0.66	0.74	0.90	0.19	0.32	0.53	0.66	0.74	0.90	
.00	.092	-1.132	-9.45	-9.68	-9.75		.470	.367	.279	.211	.148	-.046	
.01	-.398	-1.084	-9.57	-9.83	-9.95	-.898	.470	.402	.313	.269	.228	+.100	
.03	-.445	-1.097	-9.58	-9.89	-1.009	-.864	.470	.402	.313	.269	.228	+.100	
.05	-.630	-1.143	-9.57	-9.73	-1.009	-.849	.463	.399	.341	.322	.277	+.191	
.08	-.540	-1.244	-9.91	-9.62	-1.013	-.826	.463	.370	.329	.310	.282	+.221	
.10	-.490	-1.300	-1.018	-9.63	-1.009	-.802	.448	.355	.318	.310	.276	+.227	
.15	-.431	-1.243	-1.050	-1.010	-1.017	-.735	.371	.317	.292	.275	.265	+.220	
.19						-.710	.287	.265	.249	.249	.265	+.220	
.20							.530						
.25	-.399	-.427	-1.113	-1.037	-1.037	-.664	.505	.253	.234	.228	.217	.193	
.30	-.379	-.374	-1.154	-1.059	-1.073	-.628	.478	.236	.219	.205	.200	.174	
.35	-.284	-.383	-1.263	-1.073	-1.088	-.591	.450	.213	.196	.179	.185	.174	
.40	-.349	-.399	-1.156	-1.089	-1.096	-.576	.424	.197	.181	.160	.159	.134	
.45	-.414	-.398	-1.162	-1.113	-1.090	-.555	.410	.172	.163	.143	.144	.124	
.50	-.393	-.399	-1.186	-1.135	-1.063	-.541	.477	.142	.144	.125	.139	.108	
.55	-.392	-.419	-1.095	-1.193	-1.017	-.555	.455	.141	.129	.117	.109		
.60	-.426	-.426	-1.171	-1.088	-1.095	-.495	.448	.133	.125	.106	.077	.079	
.65	-.413	-.438	-1.782	-1.249	-1.056	-.566	.429	.111	.104	.085	.031	.065	
.68						-.589							
.70	-.413	-.427	-.652	-1.201	-1.046	-.451	.410	.103	.087	.055	.043	.032	
.73						-.704						.075	
.75	-.400	-.416	-.487		-1.093		.400	.081	.063			.049	
.77						-.060						.044	
.79						-.024						.044	
.80	-.343	-.332				-.089	-.432	.080	.076			.042	-.018
.81						-.956						.041	
.82						-.187						.056	
.83						-.119	-.915					.031	.036
.85	-.232	-.173	-.091			-.080		.061	.034	.048		.024	
.87	-.120	-.054				-.809		.046	.046	.048		.030	
.88						-.093						.041	
.89						-.076						.037	
.90	-.158		.013			-.089	-.444	.040				.045	-.007
.91	-.064					-.669		.040				.016	-.100
.93	-.036		.072					.003	.018	.026		.046	-.078
.95	-.083		.017			-.549						-.006	
.96						-.990							
.97													
.98													
$\alpha = 14.4$													
.00	.036	-1.234	-1.077	-1.082	-1.006		.496	.366	.261	.184	.123	.021	
.01	-.502	-1.107	-1.085	-1.105	-1.013	-.546	.522	.428	.320	.288	.219	.117	
.03	-.571	-1.202	-1.094	-1.111	-1.006	-.535	.553	.440	.358	.338	.291	.192	
.05	-.742	-1.241	-1.095	-1.109	-1.079	-.539	.521	.414	.355	.334	.305	.227	
.08	-.674	-1.366	-1.105	-1.098	-1.098	-.542	.482	.403	.347	.339	.304	.230	
.10	-.620	-1.419	-1.091	-1.093	-1.086	-.541	.426	.344	.326	.311	.299	.219	
.15	-.324	-1.434	-1.161	-1.091	-1.084	-.539	.426	.333	.297	.287	.301	.219	
.19		-1.059	-1.234	-1.113	-1.054	-.535							
.20							.376						
.25	-.444	-.717	-1.238	-1.108	-1.074	-.535	.450	.300	.269	.267	.258	.186	
.30	-.435	-.545	-1.213	-1.074	-1.097	-.533	.415	.279	.251	.242	.238	.174	
.35	-.462	-.499	-1.218	-1.074	-1.080	-.536	.429	.256	.231	.217	.215	.177	
.40	-.429	-.513	-1.234	-1.074	-1.067	-.543	.465	.237	.219	.196	.187	.132	
.45	-.485	-.492	-1.207	-1.013	-1.065	-.545	.422	.212	.192	.176	.175	.117	
.50	-.477	-.474	-1.182	-1.079	-1.081	-.545	.410	.200	.172	.149	.152	.100	
.55	-.473	-.483	-1.142	-1.079	-1.078	-.545	.410	.175	.154	.144	.123		
.60	-.483	-.479	-1.048	-1.078	-1.073	-.545	.418	.163	.141	.126	.083	.070	
.65	-.464	-.398	-1.647	-1.079	-1.028	-.547	.418	.138	.117	.099	.035		
.70	-.387	-.205	-.722	-.828	-.739	-.535	.431	.124	.093	.057	.040	.018	
.73						-.731						.080	
.75	-.268	-.153	-.663			-.719		.116	.093	.054		.041	
.77						-.796						.035	
.79						-.786						.032	
.80	-.210	-.189				-.716	-.525	.091	.079			.028	-.031
.81						-.782						.025	
.82						-.637						.027	
.83						-.628	-.753					.020	.015
.85	-.211	-.242	-.642			-.704		.054	.007	.014		.006	
.87	-.264	-.626	-.733					.020	.009	-.005			
.88		-.277										.015	
.89		-.281										.010	
.90	-.272	-.568				-.687	-.517	.010				.010	-.035
.91	-.293					-.672						.001	-.035
.93	-.274	-.482										-.018	-.033
.95	-.229	-.228				-.647		.043	-.018	-.033			-.105
.96						-.630						-.092	
.97						-.424						-.076	
.98						-.162						-.070	

TABLE I.- Continued

• 0.90

g = 16.4

$a = 17.8$

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TABLE I.- Continued
PRESSURE COEFFICIENTS ON A 5-PERCENT-THICK, 60° DELTA WING

$\alpha = 0.90$

$\alpha = 19.9$

x/c	Upper surface						Lower surface					
	0.10	0.32	0.53	0.65	0.74	0.90	0.10	0.32	0.53	0.65	0.74	0.90
.00	-1.480	-1.059	-0.960	-0.853								
.01	-1.637	-1.445	-1.070	-0.982	-0.867	-0.673						
.03	-1.463	-1.436	-1.080	-0.986	-0.872	-0.667						
.05	-1.183	-1.451	-1.068	-0.973	-0.866	-0.667						
.08	-1.051	-1.580	-1.066	-0.956	-0.858	-0.667						
.10	-1.926	-1.532	-1.086	-0.954	-0.854	-0.667						
.15	-1.762	-1.540	-1.072	-0.957	-0.849	-0.665						
.19	-1.475	-1.037	-0.957	-0.852	-0.662							
.20	-1.657											
.25	-1.624	-1.262	-1.043	-0.949	-0.842	-0.661						
.30	-1.580	-1.097	-1.029	-0.939	-0.838	-0.658						
.35	-1.549	-1.994	-1.014	-0.928	-0.833	-0.658						
.40	-1.523	-0.921	-1.013	-0.924	-0.821	-0.657						
.45	-1.594	-0.817	-0.993	-0.916	-0.813	-0.653						
.50	-1.611	-1.704	-0.977	-0.883	-0.807	-0.649						
.55	-1.568	-0.687	-0.952	-0.872	-0.801							
.60	-1.499	-0.705	-0.908	-0.821	-0.797	-0.638						
.65	-1.512	-0.721	-0.882	-0.816	-0.778							
.68												
.70	-1.549	-0.656	-0.852	-0.830	-0.778	-0.629						
.73												
.75	-1.553	-0.598	-0.798		-0.768	-0.629						
.77												
.80	-1.532	-0.552		-0.802		-0.748	-0.621					
.81												
.82							-0.802					
.83							-0.771					
.85	-1.467	-0.494	-0.782	-0.776	-0.738							
.87	-1.469	-0.776	-0.782	-0.776								
.88												
.89	-1.439											
.90	-1.425											
.91	-1.437											
.93	-1.407	-0.694										
.95	-1.366	-0.400										
.96												
.97												
.98												

$\alpha = 21.9$

.00	-1.323	-1.444	-1.091	-0.952	-0.859							
.01	-1.700	-1.442	-1.087	-0.973	-0.875	-0.725						
.03	-1.242	-1.429	-1.087	-0.979	-0.888	-0.720						
.05	-1.323	-1.460	-1.079	-0.974	-0.885	-0.720						
.08	-1.233	-1.470	-1.076	-0.964	-0.882	-0.720						
.10	-1.054	-1.432	-1.078	-0.969	-0.880	-0.720						
.15	-1.862	-1.414	-1.071	-0.975	-0.885	-0.721						
.19	-1.396	-1.066	-0.979	-0.889	-0.721							
.20	-1.738											
.23	-1.628	-1.290	-1.078	-0.973	-0.879	-0.738						
.30	-1.615	-1.171	-1.061	-0.965	-0.879	-0.716						
.35	-1.643	-1.069	-1.042	-0.957	-0.878	-0.716						
.40	-1.640	-0.975	-1.034	-0.953	-0.869	-0.716						
.45	-1.670	-0.884	-1.013	-0.943	-0.862	-0.712						
.50	-1.643	-0.824	-0.997	-0.912	-0.856	-0.708						
.55	-1.613	-0.812	-0.973	-0.908	-0.852	-0.703						
.60	-1.616	-0.793	-0.932	-0.856	-0.848	-0.701						
.65	-1.647	-0.780	-0.912	-0.858	-0.831							
.68												
.70	-1.667	-0.729	-0.885	-0.872	-0.827	-0.693						
.73							-0.819					
.75	-1.647	-0.685	-0.843		-0.811							
.77												
.79												
.80	-1.635	-0.633		-0.850	-0.679							
.81												
.82												
.83												
.85	-1.568	-0.583	-0.623	-0.825	-0.790							
.87												
.88												
.89												
.90	-1.511	-0.604		-0.776	-0.674							
.91	-1.513		-0.796									
.93	-1.485	-0.757										
.95	-1.457	-0.477										
.96												
.97												
.98												

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TABLE L - Continued
PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 60° DELTA WING

 $\chi = 0.90$ $\alpha = 24.0$

x/c	Upper surface					Lower surface						
	0.10	0.32	0.53	0.65	0.74	0.80	0.10	0.32	0.53	0.65	0.74	0.80
.00	-0.425	-1.346	-1.106	-1.009	-0.925	-0.806	.353	.363	.214	.093	.008	-0.216
.01	-0.798	-1.343	-1.114	-1.026	-0.948	-0.806	.636	.530	.348	.256	.151	-0.067
.03	-1.346	-1.339	-1.114	-1.033	-0.956	-0.795	.803	.607	.439	.367	.276	-0.071
.05	-1.331	-1.349	-1.112	-1.039	-0.956	-0.794	.764	.611	.470	.403	.330	-0.162
.08	-1.243	-1.361	-1.115	-1.022	-0.948	-0.795	.710	.613	.486	.432	.357	-0.158
.10	-1.126	-1.346	-1.124	-1.031	-0.948	-0.795	.646	.587	.483	.435	.379	-0.224
.15	-0.954	-1.350	-1.122	-1.029	-0.951	-0.795	.646	.587	.483	.435	.379	-0.224
.19	-1.326	-1.124	-1.044	-0.955	-0.791	-0.486	.446	.429	.375	.345	.245	
.25	-0.824	-1.268	-1.138	-1.042	-0.947	-0.789	.459	.527	.444	.405	.364	-0.231
.30	-0.706	-1.183	-1.124	-1.036	-0.950	-0.787	.528	.504	.421	.385	.351	-0.221
.35	-0.709	-1.104	-1.112	-1.030	-0.953	-0.785	.503	.476	.406	.362	.329	-0.020
.40	-0.688	-1.030	-1.107	-1.030	-0.945	-0.788	.472	.450	.383	.337	.304	-0.201
.45	-0.735	-0.948	-1.092	-1.020	-0.937	-0.785	.428	.421	.359	.320	.291	-0.193
.50	-0.740	-0.891	-1.079	-0.988	-0.934	-0.785	.409	.400	.334	.288	.271	-0.174
.55	-0.736	-0.847	-1.056	-0.987	-0.929	-0.782	.382	.348	.321	.279	.241	
.60	-0.740	-0.837	-1.019	-0.932	-0.929	-0.772	.357	.345	.290	.253	.294	-0.144
.65	-0.747	-0.827	-0.991	-0.938	-0.908	-0.770	.327	.308	.256	.220	.135	
.70	-0.737	-0.789	-0.966	-0.953	-0.913	-0.761	.289	.283	.224	.149	.148	-0.094
.73	-0.712	-0.754	-0.918	-0.902	-0.889	-0.760	.280	.239	.172		.135	
.77	-0.79		-0.927	-0.930	-0.880	-0.749	.224	.204			.134	
.80	-0.710	-0.711	-0.932	-0.880	-0.749	-0.749					.126	
.81			-0.878	-0.932	-0.880	-0.749					.113	
.82			-0.870	-0.917	-0.864	-0.749					.120	
.83			-0.663	-0.887	-0.864	-0.749					.115	
.85			-0.615	-0.883	-0.805	-0.749					.103	
.87			-0.604	-0.883	-0.805	-0.749					.076	
.88			-0.580	-0.883	-0.805	-0.749					.112	
.90			-0.622	-0.868	-0.843	-0.757					.095	
.91			-0.573	-0.854	-0.843	-0.757					.078	
.93			-0.529	-0.824	-0.797	-0.757					.049	
.95			-0.558	-0.502	-0.803	-0.797					.017	
.96			-0.574	-0.803	-0.797	-0.797					.005	
.97			-0.453	-0.754	-0.602	-0.602					.055	
.98			-0.453	-0.754	-0.602	-0.602						

 $\alpha = .26 \cdot 1$

x/c	0.10	0.32	0.53	0.65	0.74	0.80	0.10	0.32	0.53	0.65	0.74	0.80
.00	-0.504	-1.344	-1.151	-1.050	-0.967	-0.870	.333	.365	.204	.076	.017	-0.259
.01	-0.94	-1.345	-1.163	-1.072	-0.994	-0.870	.642	.546	.338	.249	.136	-0.058
.03	-1.383	-1.359	-1.163	-1.078	-1.008	-0.855	.832	.640	.447	.369	.279	-0.048
.05	-1.353	-1.346	-1.165	-1.078	-1.005	-0.852	.795	.649	.468	.417	.337	-0.181
.08	-1.268	-1.365	-1.177	-1.068	-1.008	-0.855	.748	.656	.508	.448	.364	-0.193
.10	-1.157	-1.360	-1.183	-1.077	-0.999	-0.855	.689	.637	.514	.456	.397	-0.226
.12	-1.003	-1.366	-1.185	-1.088	-0.999	-0.853	.607	.497	.448	.414	.249	
.15	-0.942	-1.185	-1.097	-1.009	-0.849	-0.757						
.20	-0.883	-1.281	-1.201	-1.095	-1.001	-0.848						
.30	-0.768	-1.198	-1.189	-1.092	-1.007	-0.843	.548	.551	.460	.419	.379	-0.232
.35	-0.769	-1.119	-1.177	-1.087	-1.006	-0.839	.544	.522	.442	.398	.358	-0.054
.40	-0.738	-1.045	-1.174	-1.084	-1.003	-0.846	.518	.501	.421	.373	.332	-0.218
.45	-0.782	-0.945	-1.159	-1.081	-0.998	-0.841	.475	.468	.400	.354	.320	-0.220
.50	-0.802	-0.905	-1.145	-1.043	-0.995	-0.839	.457	.445	.372	.327	.301	-0.195
.55	-0.798	-0.874	-1.143	-1.046	-0.991	-0.839	.425	.414	.348	.312	.270	
.60	-0.787	-0.848	-1.080	-0.982	-0.991	-0.824	.402	.389	.329	.285	.225	-0.164
.65	-0.770	-0.848	-1.053	-0.993	-0.985	-0.815	.375	.355	.295	.256	.164	
.68	-0.772	-0.820	-1.028	-1.001	-0.974	-0.812	.330	.324	.233	.201	.180	-0.113
.73	-0.744	-0.801	-0.976	-0.963	-0.955	-0.788	.300	.278	.202		.193	
.75	-0.773	-0.770	-0.971	-0.938	-0.793	-0.758	.241				.164	
.81	-0.82	-0.914	-0.903	-0.960	-0.974	-0.874					.137	
.85	-0.765	-0.717	-0.911	-0.909	-0.909	-0.818	.139	.145	.130			
.87	-0.671	-0.901	-0.940	-0.940	-0.940	-0.818	.148	.121	.096		.102	
.88	-0.654	-0.625	-0.902	-0.902	-0.902	-0.818	.136	.128				
.90	-0.712	-0.872	-0.878	-0.776	-0.776	-0.814	.112	.089	.049		.048	-0.060
.91	-0.612	-0.874	-0.874	-0.776	-0.776	-0.814	.112	.085	.044		.049	
.93	-0.559	-0.813	-0.813	-0.727	-0.727	-0.814	.034	.036	.044		.043	
.95	-0.579	-0.512	-0.817	-0.817	-0.817	-0.814					.026	
.96	-0.447	-0.731	-0.731	-0.633	-0.633	-0.814						
.98	-0.447	-0.447	-0.447	-0.447	-0.447	-0.814						

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TABLE L- Continued
PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 60° DELTA WING

 $\alpha = 0.92$ $\alpha = 0.3$

Upper surface

Lower surface

x/c	0.19	0.32	0.53	0.65	0.74	0.80	0.19	0.32	0.53	0.65	0.74	0.80
.00	+.167	.+284	.+257	.+265	.+270		.+100	.+021	-.001	.+025	-.020	-.310
.01	-.090	-.025	-.074	-.113	-.177	-.307	-.089	-.002	-.014	-.015	-.049	-.006
.03	-.063	-.043	-.093	-.113	-.160	-.193	-.073	-.022	-.032	-.035	-.060	-.028
.05	-.023	-.024	-.076	-.100	-.132	-.147	-.054	-.008	-.042	-.046	-.063	-.075
.08	-.004	-.026	-.084	-.108	-.112	-.157	-.023	-.004	-.048	-.049	-.076	-.082
.10	-.000	-.032	-.075	-.084	-.120	-.149	-.011	-.016	-.064	-.056	-.088	-.091
.15	-.049	-.049	-.079	-.114	-.120	-.154	-.032	-.072	-.075	-.091	-.082	
.19	-.059	-.082	-.132	-.179	-.152							
.20	-.022						-.005					
.25	-.041	-.072	-.108	-.115	-.123	-.141	-.018	-.049	-.075	-.074	-.105	-.090
.30	-.066	-.077	-.112	-.118	-.142	-.135	-.030	-.059	-.079	-.091	-.107	-.071
.35	-.066	-.091	-.116	-.123	-.140	-.123	-.033	-.075	-.096	-.105	-.115	-.114
.40	-.045	-.096	-.128	-.131	-.133	-.145	-.057	-.077	-.099	-.114	-.121	-.138
.45	-.105	-.101	-.135	-.131	-.128	-.154	-.088	-.091	-.108	-.120	-.127	-.134
.50	-.091	-.115	-.141	-.132	-.148	-.162	-.079	-.090	-.119	-.129	-.124	-.131
.55	-.094	-.124	-.145	-.128	-.141		-.090	-.100	-.118	-.115	-.130	
.60	-.105	-.117	-.113	-.110	-.137	-.144	-.082	-.098	-.105	-.105	-.134	-.108
.65	-.105	-.124	-.109	-.083	-.110		-.090	-.103	-.103	-.105	-.120	
.68												
.70	-.093	-.106	-.103	-.097	-.110	-.080	-.093	-.094	-.102	-.113	-.134	-.089
.73												
.75	-.088	-.094	-.074				-.079	-.097	-.106			
.77												
.79												
.80	-.081	-.066					-.083	-.079				
.81												
.82												
.83												
.85	-.051	-.051	-.049	-.041			-.070	-.090	-.059	-.051		
.87												
.88												
.89												
.90	-.062		-.017	-.001	-.001	-.009	-.109		-.035		-.016	-.004
.91	-.031						-.066		-.005			
.93	-.017	-.004					-.058	-.013				
.95	-.041	-.013					-.064	-.044				
.96												
.97												
.98												

 $\alpha = 2.3$

.00	.+189	.+128	.+142	.+189	.+555		.+169	.+146	.+160	.+184	.+169	.+070
.01	-.040	-.219	-.353	-.622	-.445	-.597	-.069	-.026	-.004	-.024	-.011	-.027
.03	-.003	-.112	-.283	-.486	-.446	-.555	-.140	-.109	-.138	-.135	-.130	-.175
.05	-.035	-.114	-.220	-.280	-.410	-.545	-.144	-.101	-.021	-.100	-.097	-.160
.08	-.054	-.101	-.201	-.255	-.348	-.535	-.124	-.076	-.070	-.080	-.074	-.060
.10	-.054	-.178	-.217	-.234	-.535	-.522	-.089	-.070	-.067	-.068	-.050	-.070
.15	-.068	-.106	-.157	-.234	-.299	-.064	-.070	-.051	-.041	-.032	-.026	-.042
.19	-.068	-.106	-.157	-.234	-.311	-.1495		-.051	-.017	-.006	-.043	-.034
.20												
.25	-.088	-.119	-.164	-.208	-.256	-.461	-.046	-.009	-.003	-.005	-.001	-.020
.30	-.088	-.119	-.167	-.196	-.253	-.480	-.026	-.004	-.004	-.024	-.011	-.027
.35	-.106	-.134	-.172	-.203	-.246	-.381	-.024	-.020	-.023	-.041	-.029	-.006
.40	-.082	-.137	-.180	-.210	-.224	-.210	-.002	-.026	-.033	-.053	-.042	-.040
.45	-.143	-.141	-.178	-.206	-.208	-.299	-.034	-.042	-.044	-.061	-.046	-.043
.50	-.126	-.153	-.182	-.202	-.215	-.261	-.029	-.042	-.035	-.076	-.046	-.051
.55	-.131	-.162	-.183	-.192	-.206		-.041	-.054	-.057	-.063	-.062	
.60	-.143	-.154	-.151	-.164	-.191	-.175	-.037	-.054	-.050	-.060	-.072	-.048
.65	-.140	-.156	-.144	-.137	-.147	-.150	-.047	-.063	-.053	-.065	-.071	-.066
.68												
.70	-.128	-.140	-.126	-.130	-.138	-.083	-.053	-.055	-.053	-.077	-.075	-.055
.73												
.75	-.119	-.121	-.089				-.043	-.062	-.063			
.77												
.79												
.80	-.103	-.084					-.049	-.050			-.030	-.042
.81												
.82												
.83												
.85	-.066	-.059	-.041				-.045	-.066	-.033			
.87												
.88												
.89												
.90	-.067	-.007					-.069	-.015				
.91	-.033							-.051				
.93	-.017	.012						-.046	-.004			
.95	-.042	-.011						-.051	-.037			
.96												
.97												
.98												

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TABLE I- Continued

M 8 0.92

$$\alpha \approx 4.3$$

Upper surface						Lower surface						
Stations, fraction of semispan						Stations, fraction of semispan						
x/c	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90
.00	+.171	-+.213	-+.617	-+.611	-+.703		+.231	+.229	+.251	+.246	+.225	+.135
.01	-+.027	-+.407	-+.601	-+.682	-+.673	-+.646	+.231	+.194	+.225	+.217	+.208	+.217
.03	-+.072	-+.349	-+.596	-+.719	-+.682	-+.622	+.214	+.168	+.186	+.193	+.189	+.206
.05	-+.102	-+.244	-+.594	-+.548	-+.688	-+.622	+.193	+.157	+.153	+.156	+.162	+.173
.08	-+.122	-+.215	-+.609	-+.465	-+.699	-+.620	+.155	+.147	+.137	+.142	+.134	+.153
.10	-+.113	-+.204	-+.566	-+.424	-+.707	-+.617	+.132	+.110	+.105	+.103	+.109	+.118
.15	-+.124	-+.185	-+.418	-+.417	-+.741	-+.609	+.085	+.080	+.072	+.075	+.075	+.112
.19	-+.185	-+.276	-+.400	-+.771	-+.601							
.20	-+.120						+.108					
.25	-+.141	-+.189	-+.221	-+.381	-+.677	-+.593	+.100	+.061	+.059	+.059	+.063	+.080
.30	-+.134	-+.188	-+.196	-+.374	-+.586	-+.584	+.077	+.049	+.049	+.037	+.050	+.077
.35	-+.158	-+.200	-+.199	-+.361	-+.649	-+.582	+.071	+.052	+.029	+.016	+.029	+.062
.40	-+.127	-+.201	-+.223	-+.352	-+.316	-+.579	+.047	+.024	+.017	+.002	+.014	+.019
.45	-+.194	-+.196	-+.237	-+.337	-+.231	-+.569	+.013	+.005	+.002	-.007	+.006	+.014
.50	-+.175	-+.203	-+.247	-+.310	-+.220	-+.555	+.013	+.005	-.011	-.024	+.002	+.005
.55	-+.175	-+.216	-+.243	-+.284	-+.193		+.000	-.008	-.017	-.017	-.014	
.60	-+.191	-+.212	-+.201	-+.241	-+.172	-+.511	+.003	-.011	-.013	-.016	-.033	-.006
.65	-+.191	-+.216	-+.180	-+.176	-+.115		-.007	-.023	-.019	-.024	-.046	
.68					-+.124						-.031	
.70	-+.178	-+.189	-+.168	-+.180	-+.111	-+.461	-.018	-.020	-.024	-.043	-.043	-.025
.73					-+.085						-.008	
.75	-+.163	-+.163	-+.113		-+.062		-.011	-.032	-.036			-.021
.77					-+.126						-.033	
.79					-+.106						-.025	
.80	-+.140	-+.109			-+.035	-+.393	-.022	-.021				-.009
.81					-+.094							-.039
.82					-+.073							-.022
.83					-+.056	-+.071						-.022
.85	-+.081	-+.072	-+.054		-+.005		-.023	-.044	-.019	-.018	-.000	
.87					-+.058	-+.039	-+.048		-.032	-.015	-.007	
.88					-+.051				-.033			
.89					-+.041				-.038			
.90					-+.079							-.024
.91					-+.039	-+.012	-+.020	-+.014	-+.326			
.93					-+.017							-.021
.95					-+.040							-.018
.96					-+.014							-.000
.97					-+.022							-.059
.98					-+.001							

9 = 603

CONFIDENTIAL

TABLE L - Continued

PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 60° DELTA WING

 $\alpha = 0.92$ $\alpha = 8.3$

x/c	Upper surface					Lower surface						
	0.10	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90
.00	-142	-870	-835	-842	-809		350	343	282	240	200	.086
.01	-207	-791	-719	-864	-816	-747	353	322	281	262	228	.168
.03	-245	-835	-729	-850	-823	-726	349	302	271	267	242	.205
.05	-293	-964	-740	-805	-828	-718	318	268	247	238	227	.205
.08	-329	-980	-775	-785	-830	-705	279	249	231	230	206	.196
.10	-293	-923	-793	-781	-834	-688	244	213	197	189	185	.169
.15	-289	-263	-854	-800	-850	-656	183	167	160	181	163	
.19	-257	-878	-819	-876	-631		212					
.20	-273						198	155	142	141	136	.129
.25	-287	-260	-901	-836	-892	-593	168	140	126	114	119	.114
.30	-265	-272	-897	-846	-922	-564	153	118	102	091	098	.108
.35	-285	-297	-862	-853	-928	-531	126	106	091	069	074	.068
.40	-244	-314	-797	-859	-942	-509	092	084	073	057	065	.058
.45	-322	-306	-707	-864	-936	-488	088	077	056	037	057	.042
.50	-305	-314	-579	-837	-937	-469	069	059	045	039	031	
.55	-301	-331	-435	-839	-925	-435	066	054	042	033	020	
.60	-324	-337	-281	-780	-896	-429	052	035	029	016	034	
.65	-324	-349	-278	-716	-841	-428						
.68												
.70	-324	-333	-297	-671	-807	-395	035	032	019	010	022	.020
.75	-321	-328	-254			-759	028	016	004		006	
.77												
.79												
.80	-303	-273					015	017				
.81												
.82												
.83												
.85	-207	-153	-020			-678	004	016	-001		008	
.87	-107	-001	-0227				004	004	002	-004	009	
.88	-082											
.89	-065											
.90	-128	-032	-090	-627	-366		-010					
.91	-056											
.93	-030	-053										
.95	-063	-017										
.96												
.97												
.98												

 $\alpha = .10 \cdot 3$

x/c	-1.053	-1.080	-0.888	-0.879	-0.891	-0.403	-0.359	-0.291	-0.232	-0.187	-0.042	
.00	-132	-1.053	-0.880	-0.888	-0.879	-0.403	-0.359	-0.291	-0.232	-0.187	-0.042	
.01	-278	-0.890	-0.845	-0.887	-0.889	-0.891	-0.419	-0.369	-0.308	-0.283	-0.235	.132
.03	-339	-0.855	-0.895	-0.900	-0.867	-0.867	-0.424	-0.354	-0.312	-0.303	-0.270	.195
.05	-403	-1.025	-0.847	-0.876	-0.901	-0.864	-0.396	-0.323	-0.297	-0.282	-0.263	.213
.08	-412	-1.118	-0.876	-0.866	-0.899	-0.863	-0.352	-0.305	-0.284	-0.272	-0.245	.211
.10	-378	-1.184	-0.900	-0.861	-0.894	-0.856	-0.315	-0.266	-0.254	-0.237	-0.227	.189
.15	-354	-0.827	-0.940	-0.896	-0.911	-0.829	-0.236	-0.226	-0.211	-0.225		
.19	-540	-0.996	-0.913	-0.945	-0.800		279					
.20	-323						257	-0.204	-0.194	-0.192	-0.182	.154
.25	-335	-294	-1.100	-0.931	-0.941	-0.745	227	-0.189	-0.177	-0.165	-0.163	.138
.30	-312	-299	-1.098	-0.954	-0.961	-0.705	208	-0.167	-0.154	-0.141	-0.139	.138
.35	-331	-326	-1.109	-0.979	-0.973	-0.655	178	-0.153	-0.138	-0.122	-0.117	.099
.40	-291	-338	-1.039	-1.013	-0.992	-0.629	140	-0.127	-0.122	-0.107	-0.104	.090
.45	-355	-343	-1.042	-1.042	-1.008	-0.597	116	-0.120	-0.101	-0.084	-0.073	.076
.50	-344	-335	-0.912	-1.028	-1.034	-0.570	114	-0.111	-0.099	-0.065	-0.077	
.55	-345	-372	-0.937	-1.046	-1.007	-0.577	111	-0.092	-0.085	-0.074	-0.053	.050
.60	-363	-377	-1.017	-0.990	-1.077	-0.519	094	-0.073	-0.069	-0.057	-0.007	
.65	-363	-392	-1.026	-1.007	-1.046							
.68												
.70	-363	-383	-0.373	-1.012	-1.009	-0.484	076	067	053	027	009	.013
.73	-367	-387	-0.352			-0.983	066	047	032		023	
.77						-0.882						
.79						-0.829						
.80	-367	-361				-0.954	-0.441	049	045		021	-0.037
.81						-0.740						
.82												
.83												
.85	-293	-276	-1.126			-0.458	035	007	026		011	
.87	-209	-0.41	-0.561				021	029	021			
.88	-166						014					
.89	-128						014					
.90	-186	-0.057	-0.323	-0.882	-0.411		014		030	-0.013	-0.107	
.91	-0.097						010		023			
.93	-0.047	-0.098					006	038				
.95	-0.079	-0.020				-0.655	014	002		-0.064		
.96						-0.085						
.97						-0.101						
.98						-0.005						

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TABLE L- Continued

$\mu = 0.98$

a = 12.4

TABLE L- Continued

PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 60° DELTA WING

 $M = 0.92$ $c = 17.8$

Upper surface						Lower surface							
Stations, fraction of semispan						Stations, fraction of semispan							
x/c	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	
.00	-0.071	-1.340	-1.189	-0.946	-0.805		+0.28	+3.85	+2.57	+1.68	+1.04	+0.61	
.01	-0.142	-1.320	-1.189	-0.949	-0.829	-0.722	+0.02	+4.87	+3.26	+2.91	+2.08	+0.53	
.03	-0.298	-1.190	-0.946	-0.842	-0.709		+0.79	+5.21	+4.02	+3.47	+2.92	+1.42	
.05	-0.933	-1.322	-1.189	-0.925	-0.848	-0.708		+6.38	+5.03	+4.14	+3.79	+3.23	+2.04
.08	-0.859	-1.503	-1.189	-0.907	-0.849	-0.704		+5.93	+4.98	+4.13	+3.89	+3.30	+2.22
.10	-0.765	-1.439	-1.185	-0.891	-0.849	-0.699		+5.26	+4.42	+3.95	+3.72	+3.33	+2.26
.15	-0.642	-1.488	-1.192	-0.914	-0.855	-0.689		+4.30	+3.72	+3.50	+3.39	+3.39	+2.37
.19	-1.392	-1.242	-0.956	-0.881	-0.680								
.20	-0.547	-1.127	-0.934	-0.862	-0.666		+4.76	+2.95	+3.48	+3.30	+2.99	+2.13	
.30	-0.532	-0.937	-1.053	-0.902	-0.856	-0.59	+4.10	+3.72	+3.27	+3.04	+2.84	+1.98	
.35	-0.544	-0.806	-1.027	-0.883	-0.842	-0.653	+3.82	+3.47	+3.05	+2.80	+2.61	+0.14	
.40	-0.517	-0.722	-1.021	-0.886	-0.839	-0.648	+3.55	+3.25	+2.83	+2.55	+2.32	+1.68	
.45	-0.561	-0.668	-1.027	-0.894	-0.834	-0.643	+3.13	+2.99	+2.61	+2.37	+2.20	+1.58	
.50	-0.552	-0.559	-0.993	-0.891	-0.837	-0.538	+2.97	+2.83	+2.37	+2.10	+2.09	+1.40	
.55	-0.531	-0.426	-0.968	-0.910	-0.835		+2.72	+2.56	+2.19	+2.01	+1.73		
.60	-0.441	-0.426	-0.948	-0.867	-0.832	-0.627	+2.57	+2.37	+2.02	+1.79	+1.50	+1.16	
.65	-0.372	-0.527	-0.941	-0.867	-0.806		+2.35	+2.09	+1.79	+1.52	+0.73		
.68					-0.806							+1.14	
.70	-0.400	-0.586	-0.917	-0.882	-0.803	-0.614	+2.03	+1.91	+1.47	+1.03	+0.91	+0.67	
.73	-0.482	-0.605	-0.863			-0.779	+1.80	+1.53	+1.02			+1.08	
.77					-0.847							+0.77	
.79					-0.846							+0.74	
.80	-0.549	-0.566			-0.761	-0.599	+1.49	+1.29				+0.68	
.81					-0.835							+0.63	
.82					-0.818							+0.63	
.83					-0.809	-0.811						+0.60	
.85	-0.506	-0.488	-0.819			-0.747	+0.98	+0.45	+0.54	+0.58		+0.42	
.87					-0.453	-0.809	+0.791		+0.59	+0.46	+0.31		
.88					-0.442				+0.51				
.89					-0.427				+0.45				
.90	-0.455	-0.427	-0.772	-0.756	-0.728	-0.587	+0.49		+0.24		+0.03	-0.01	
.91								+0.33					
.93	-0.412	-0.712					+0.17	+0.12	-0.09				
.95	-0.386	-0.398						-0.017	-0.024			-0.07	
.96										-0.06			
.97					-0.727								
.98	-0.347				-0.637				-0.064				
								-0.082					

 $c = 19.9$

.00	-0.194	-1.448	-1.095	-0.956	-0.916		+5.18					
.01	-0.605	-1.412	-1.101	-0.985	-0.932	-0.730	+4.07	+3.75	+2.45	+1.39	+0.64	+1.19
.03	-1.022	-1.403	-1.107	-0.996	-0.938	-0.717	+6.28	+5.03	+4.29	+2.71	+1.82	+0.02
.05	-1.144	-1.384	-1.099	-0.995	-0.934	-0.717	+7.36	+5.45	+4.18	+3.31	+2.26	+1.1
.08	-1.020	-1.548	-1.089	-0.976	-0.927	-0.717	+6.92	+5.40	+4.54	+3.85	+2.25	+1.95
.10	-0.897	-1.507	-1.098	-0.976	-0.917	-0.714	+6.43	+5.34	+4.59	+3.99	+2.40	+2.17
.15	-1.533	-1.128	-0.994	-0.911	-0.709	-0.709	+5.74	+5.03	+4.27	+3.88	+3.51	+2.20
.19	-1.463	-1.081	-1.003	-0.911	-0.705		+4.73	+4.04	+3.69	+3.59	+2.41	
.20	-0.651											
.25	-1.628	-1.242	-1.056	-0.988	-0.899	-0.700	+4.90	+4.36	+3.80	+3.52	+2.22	
.30	-1.402	-1.077	-1.054	-0.980	-0.882	-0.696	+4.48	+4.14	+3.28	+3.09	+2.08	
.35	-0.887	-0.950	-1.056	-0.972	-0.888	-0.696	+4.23	+3.89	+3.36	+3.04	+2.84	+0.04
.40	-0.532	-0.883	-1.057	-0.965	-0.869	-0.695	+3.97	+3.65	+3.17	+2.78	+2.58	+1.80
.45	-0.565	-0.842	-1.058	-0.963	-0.857	-0.688	+3.50	+3.34	+2.95	+2.62	+2.44	+1.72
.50	-0.590	-0.758	-1.021	-0.929	-0.849	-0.685	+3.35	+3.19	+2.88	+2.31	+2.32	+1.54
.55	-0.612	-0.684	-0.999	-0.925	-0.842	-0.682	+3.09	+2.90	+2.46	+2.22	+1.96	
.60	-0.566	-0.688	-0.983	-0.866	-0.840	-0.673	+2.91	+2.68	+2.32	+1.99	+1.52	+1.28
.65	-0.525	-0.728	-0.941	-0.865	-0.817	-0.617	+2.66	+2.37	+2.00	+1.68	+0.94	
.68					-0.823							+1.34
.70	-0.556	-0.705	-0.914	-0.875	-0.822	-0.662	+2.29	+2.17	+1.69	+1.18	+1.10	-0.079
.73	-0.590	-0.660	-0.861		-0.801		+2.03	+1.78	+1.23			+1.28
.77					-0.853							+0.89
.80	-0.603	-0.609			-0.850	-0.795	+1.69	+1.49				+0.85
.81												+0.74
.82					-0.821				+0.85			
.83					-0.815	-0.831		+0.72	+0.68			
.85	-0.538	-0.542	-0.831		-0.780		+1.07	+0.52	+0.64			+0.55
.87					-0.510	-0.824	+0.87	+0.59	+0.41			
.88					-0.500			+0.58				
.89					-0.482			+0.51				
.90	-0.483	-0.798		-0.765	-0.642		+0.63	+0.41	+0.34		+0.14	-0.073
.91	-0.476		-0.791					+0.17	-0.005	+0.04		
.93	-0.447	-0.744			-0.723		+0.010	-0.020				-0.059
.95	-0.410	-0.436			-0.766					-0.058		
.96					-0.687				-0.066			
.98	-0.411							-0.085				

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TABLE I.- Continued

M = 0.93

$\alpha = 22.0$

Upper surface							Lower surface						
Stations, fraction of semispan							Stations, fraction of semispan						
x/c	0.19	0.32	0.53	0.66	0.74	0.90	0.18	0.32	0.53	0.65	0.74	0.74	0.90
.00	-0.283	-1.466	-1.090	-0.772	-0.895	-0.763	.397	.372	.245	.130	.049	.158	
.01	-0.654	-1.441	-1.097	-0.995	-0.913	-0.763	.648	.519	.339	.281	.179	.026	
.03	-1.202	-1.431	-1.102	-1.001	-0.920	-0.755	.788	.580	.440	.376	.294	.101	
.05	-1.278	-1.482	-1.093	-0.994	-0.917	-0.753	.745	.579	.463	.407	.358	.185	
.08	-1.187	-1.512	-1.093	-0.980	-0.913	-0.754	.691	.577	.473	.428	.357	.212	
.10	-1.011	-1.477	-1.103	-0.986	-0.913	-0.754	.619	.548	.449	.421	.374	.234	
.15	-0.820	-1.462	-1.103	-0.993	-0.913	-0.751	.519	.448	.404	.386	.326	.247	
.19	-1.433	-1.079	-0.997	-0.913	-0.749								
.20	-0.713						.570						
.25	-0.641	-1.297	-1.084	-0.901	-0.808	-0.745	.527	.482	.425	.390	.350	.232	
.30	-0.551	-1.171	-1.070	-0.983	-0.806	-0.742	.501	.442	.403	.349	.324	.218	
.35	-0.604	-1.057	-1.056	-0.977	-0.804	-0.742	.476	.436	.382	.344	.316	.205	
.40	-0.613	-0.982	-1.058	-0.973	-0.802	-0.742	.450	.413	.361	.319	.287	.195	
.45	-0.601	-0.885	-1.039	-0.967	-0.803	-0.739	.402	.380	.336	.301	.276	.189	
.50	-0.660	-0.805	-1.023	-0.954	-0.802	-0.735	.383	.362	.313	.272	.258	.169	
.55	-0.620	-0.784	-0.999	-0.933	-0.802	-0.734	.355	.330	.288	.245	.224		
.60	-0.596	-0.781	-0.960	-0.877	-0.807	-0.724	.326	.308	.270	.240	.181	.142	
.65	-0.609	-0.793	-0.946	-0.884	-0.851		.309	.277	.241	.208	.121		
.68												.162	
.70	-0.641	-0.753	-0.917	-0.896	-0.858	-0.712	.272	.250	.207	.155	.136	.095	
.73												.155	
.75	-0.654	-0.712	-0.871				.244	.207	.160			.128	
.77												.120	
.79												.116	
.80	-0.659	-0.659			-0.876		.208	.178				.106	.034
.81					-0.876							.105	
.82					-0.842							.117	
.83					-0.837	-0.864						.111	
.85	-0.595	-0.604	-0.855		-0.817		.146	.073	.097			.076	
.87	-0.559	-0.853	-0.856					.092	.092	.089	.070		
.88	-0.554								.079				
.89	-0.534								.074				
.90	-0.529		-0.838		-0.802	-0.691	.092		.062		.030	-0.063	
.91	-0.532				-0.816			.061			.032		
.93	-0.508	-0.798						.034	.023				
.95	-0.468	-0.498			-0.780	-0.756	.011	-0.008			-0.033		
.97					-0.733					-0.041			
.98			-0.476							-0.074			

$\alpha = 24.1$

CONFEDERATION

TABLE I.- Continued

M = 0.92

g = 2642

TABLE L- Continued

PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 60° DELTA WING

X = 0.94

$\alpha = 0.3$

Upper surface

Stations, fraction of semispans							Stations, fraction of semispans						
x/c	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	
.00	.+082	.+000	.+266	.+270	.+267		.+112	.+038	.+016	.+034	.+003	.+294	
.01	.+014	.-009	.+014	.+017	.+011		.+111	.+022	.+005	.+004	.-028	.+022	
.03	.-027	.+080	.-166	.-152	.-202		.+092	.+040	.+016	.+028	.+037	.+008	
.05	.+042	.-006	.+041	.+090	.+122		.+076	.+027	.+028	.+040	.+042	.+056	
.08	.+021	.-007	.+071	.+097	.+101		.+042	.+022	.+034	.+044	.+056	.+045	
.10	.+017	.-015	.+043	.+077	.+108		.+028	.-001	.+049	.+052	.+071	.+076	
.15	.-001	.-031	.+045	.+108	.+114		.+016	.-058	.+072	.+072	.+067		
.19		.-041	.-060	.+129	.+177		.+009						
.20							.+002	.-031	.+062	.+073	.+086	.+080	
.25	.-028	.+055	.-093	.+111	.+112		.+012	.+043	.+046	.+089	.+091	.+050	
.30	.-033	.+061	.+098	.+136	.+133		.+016	.+048	.+084	.+104	.+101	.+103	
.35	.-050	.+076	.+108	.+123	.+130		.+038	.+063	.+090	.+115	.+109	.+120	
.40	.-030	.+081	.+120	.+130	.+133		.+071	.+081	.+100	.+122	.+112	.+130	
.45	.-095	.+087	.+127	.+135	.+126		.+064	.+079	.+112	.+136	.+111	.+130	
.50	.-079	.+101	.+138	.+140	.+150		.+077	.+090	.+112	.+119	.+119		
.55	.-085	.+115	.+140	.+139	.+145		.+072	.+090	.+105	.+109	.+122	.+099	
.60	.-100	.+112	.+115	.+119	.+133		.+087	.+101	.+101	.+110	.+107		
.65	.-102	.+121	.+111	.+099	.+104		.+089	.+094	.+099	.+118	.+145	.+070	
.68					.+114		.+076	.+094	.+101	.+112	.+086		
.70	.-094	.+107	.+105	.+103	.+106		.+080	.+078					
.73							.+076	.+094	.+101	.+112	.+086		
.75	.+092	.+097	.+069		.+065		.+087	.+091	.+095	.+091	.+042	.+024	
.77							.+097						
.79							.+087						
.80	.+085	.+063					.+069	.+095	.+092	.+051	.+018		
.81							.+069	.+077	.+049	.+023			
.82							.+070	.+070					
.83							.+063						
.85	.+049	.+040	.+055		.+003		.+069	.+095	.+092				
.87			.+028	.+024	.+012		.+069	.+077	.+049	.+023			
.88			.+024				.+070						
.89			.+017				.+063						
.90	.+055		.+001		.+022		.+117	.+024					
.91		.+017		.+024			.+062						
.93		.+003	.+020				.+050	.+001					
.95	.+030	.+003			.+056		.+057	.+036				.+050	
.96					.+037							.+040	
.97								.+052					
.98								.+013					

g = 2.3

TABLE L - Continued

PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 60° DELTA WING

 $\alpha = 0.94$ $\alpha = 4.3$

Upper surface						Lower surface						
x/c	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90
.00	+.173	-+.184	-+.578	-+.560	-+.702							
.01	=.011	-+.393	-+.593	-+.658	-+.669							
.03	=.060	-+.335	-+.588	-+.702	-+.680	-+.673						
.05	=.091	-+.233	-+.583	-+.556	-+.685	-+.673						
.08	=.116	-+.205	-+.598	-+.468	-+.693	-+.679						
.10	=.104	-+.192	-+.549	-+.426	-+.699	-+.679						
.15	=.116	-+.178	-+.399	-+.416	-+.729	-+.673						
.19	-+.178	-+.270	-+.397	-+.759	-+.667							
.20	=.116											
.25	=.154	-+.182	-+.230	-+.376	-+.679	-+.660						
.30	=.129	-+.185	-+.206	-+.370	-+.594	-+.653						
.35	=.150	-+.198	-+.203	-+.361	-+.508	-+.645						
.40	=.126	-+.211	-+.225	-+.359	-+.342	-+.635						
.45	=.125	-+.199	-+.238	-+.357	-+.281	-+.613						
.50	=.168	-+.204	-+.244	-+.404	-+.345	-+.585						
.55	=.178	-+.217	-+.259	-+.324	-+.246	-+.502						
.60	=.196	-+.218	-+.227	-+.302	-+.246	-+.520						
.65	=.197	-+.233	-+.244	-+.256	-+.205							
.68					-+.198							
.70	=.197	-+.219	-+.244	-+.270	-+.181	-+.457						
.73						-+.186						
.75	=.202	-+.220	-+.195			-+.088						
.77						-+.177						
.79						-+.138						
.80	=.196	-+.178				-+.030	-+.374					
.81						-+.113						
.82											-+.037	
.83						-+.069	-+.080				-+.034	
.85	=.127	-+.104	-+.055			-+.005					-+.031	
.87						-+.077	-+.038	-+.048			-+.024	
.88						-+.060					-+.014	
.89						-+.046					-+.042	
.90	=.093					-+.004	-+.030	-+.299			-+.015	
.91						-+.041	-+.014				-+.043	
.93						-+.015	-+.018				-+.040	
.95	=.042					-+.009		-+.051			-+.049	
.96							-+.011				-+.034	
.97							-+.032				-+.028	
.98						-+.005					-+.017	

 $\alpha = 6.3$

.00	+.145	-+.583	-+.802	-+.836	-+.740							
.01	=.093	-+.683	-+.611	-+.787	-+.729	-+.754						
.03	=.143	-+.647	-+.620	-+.784	-+.737	-+.735						
.05	=.176	-+.660	-+.620	-+.709	-+.742	-+.734						
.08	=.205	-+.601	-+.631	-+.670	-+.744	-+.734						
.10	=.182	-+.402	-+.632	-+.653	-+.753	-+.727						
.15	=.199	-+.173	-+.637	-+.657	-+.773	-+.710						
.19		-+.203	-+.633	-+.657	-+.803	-+.694						
.20	=.179										-+.166	
.25	=.206	-+.214	-+.608	-+.651	-+.818	-+.668					-+.155	-+.110
.30	-+.193	-+.216	-+.535	-+.638	-+.818	-+.645					-+.130	-+.094
.35		-+.211	-+.233	-+.454	-+.616	-+.805	-+.613				-+.118	-+.075
.40	-+.174	-+.251	-+.404	-+.590	-+.804	-+.592					-+.090	-+.065
.45	-+.255	-+.268	-+.368	-+.565	-+.749	-+.565					-+.054	-+.042
.50	-+.236	-+.258	-+.352	-+.520	-+.752	-+.540					-+.056	-+.039
.55	-+.236	-+.278	-+.335	-+.497	-+.710						-+.037	-+.023
.60	-+.256	-+.278	-+.290	-+.450	-+.663	-+.496					-+.037	-+.018
.65	-+.261	-+.292	-+.301	-+.398	-+.567						-+.019	-+.005
.68					-+.538						-+.031	
.70	-+.261	-+.281	-+.308	-+.407	-+.499	-+.458					-+.009	-+.007
.73					-+.448						-+.002	
.75	-+.267	-+.285	-+.289		-+.379						-+.031	
.77					-+.305						-+.025	
.79					-+.261						-+.015	
.80	-+.275	-+.266			-+.283	-+.434					-+.006	-+.015
.81					-+.224						-+.023	
.82					-+.204						-+.022	
.83					-+.158	-+.172					-+.019	-+.021
.85	-+.222	-+.211	-+.119		-+.107	-+.188					-+.045	-+.019
.87					-+.163	-+.076					-+.029	-+.017
.88					-+.130						-+.031	
.89					-+.099						-+.033	
.90	-+.162				-+.019	-+.116	-+.424				-+.009	-+.019
.91					-+.080	-+.061					-+.036	-+.006
.93					-+.057	-+.013					-+.036	-+.003
.95	-+.063	-+.016			-+.011	-+.046					-+.046	-+.018
.96					-+.030						-+.010	
.97					-+.001						-+.027	
.98								-+.019				

TABLE I.- Continued

M = 0.91

TABLE L - Continued

PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 60° DELTA WING

M = 0.94

$\alpha = 12.4$

Upper surface							Lower surface						
Stations, fraction of semispan						Stations, fraction of semispan							
x/c	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	
.00	-1.123	-1.076	-0.897	-0.904	-0.926	-0.961	.464	.384	.299	.222	.168	.002	
.01	-1.340	-1.027	-0.903	-0.919	-0.943	-0.961	.488	.414	.332	.294	.236	.095	
.03	-1.386	-1.038	-0.903	-0.933	-0.956	-0.939	.511	.411	.349	.329	.284	.172	
.05	-1.573	-1.083	-0.908	-0.923	-0.959	-0.936	.483	.382	.340	.315	.284	.205	
.08	-1.495	-1.176	-0.941	-0.937	-0.961	-0.943	.436	.366	.326	.312	.275	.206	
.10	-1.453	-1.195	-0.963	-0.922	-0.961	-0.943	.388	.327	.302	.279	.261	.190	
.15	-1.407	-1.200	-0.997	-0.968	-0.964	-0.948	.297	.272	.249	.261	.211	.193	
.19	-1.754	-1.033	-1.007	-0.995	-0.947								
.20	-1.364						.345						
.25	-1.367	-0.398	-1.063	-0.981	-0.990	-0.935	.321	.263	.241	.232	.214	.159	
.30	-1.343	-1.351	-1.097	-1.008	-1.016	-0.925	.284	.244	.221	.206	.200	.142	
.35	-1.367	-1.362	-1.159	-1.020	-1.028	-0.908	.264	.219	.201	.180	.172	.147	
.40	-1.329	-1.385	-1.247	-1.035	-1.034	-0.926	.234	.204	.185	.157	.149	.107	
.45	-1.390	-1.414	-1.222	-1.074	-1.047	-0.918	.191	.183	.153	.137	.137	.099	
.50	-1.374	-1.385	-1.059	-1.064	-1.093	-0.869	.183	.166	.140	.116	.114	.088	
.55	-1.381	-1.408	-1.083	-1.124	-1.041	-0.861	.161	.145	.126	.108	.105	.085	
.60	-1.405	-1.418	-0.946	-1.083	-1.065	-0.763	.155	.134	.117	.097	.050	.065	
.65	-1.406	-1.430	-0.808	-1.186	-1.087		.134	.111	.097	.073	-.002		
.68						-1.121					.039		
.70	-1.406	-1.424	-1.673	-1.208	-1.154	-0.701	.112	.099	.076	.031	.018	.024	
.73						-1.176					.066		
.75	-1.414	-1.435	-1.541		-1.193		.097	.072	.041			.028	
.77				-1.170							.017		
.79				-1.169							.021		
.80	-1.432	-1.427			-1.171	-0.638	.074	.059				.023	-0.023
.81				-1.138							.018		
.82				-1.464							.022		
.83				-1.442	-1.129						.020		
.85	-1.415	-1.414	-1.446		-1.090		.045	.001	.017			.010	
.87				-1.394	-1.439	-1.078					.014		
.88				-1.388							.009		
.89				-1.378							.008		
.90	-1.406		-1.399		-0.981	-0.590	.008		.009			-0.018	-0.092
.91		-1.376			-0.955				.001			-0.008	
.93		-1.346	-1.373						-0.012				
.95	-1.288	-1.283				-0.931		-0.037	-0.031			-0.073	
.96					-0.814						-0.037		
.97				-1.228							.012		
.98		-1.182									-0.054		

5 = 14.4

TABLE L - Continued
PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 60° DELTA WING

 $M = 0.94$ $\alpha = 17.8$

Upper surface							Lower surface							
Stations, fraction of semispan							Stations, fraction of semispan							
x/c	0.10	0.32	0.53	0.65	0.74	0.90	0.10	0.32	0.53	0.65	0.74	0.90		
.00	-0.049	-1.306	-1.246	-1.094	-0.926	-0.765	.443	.396	.277	.178	.106	-0.077		
.01	-0.513	-1.273	-1.140	-1.099	-0.928	-0.765	.617	.503	.357	.300	.212	.033		
.03	-0.759	-1.252	-1.146	-1.102	-0.928	-0.748	.692	.532	.415	.373	.302	.141		
.05	-0.934	-1.263	-1.146	-1.094	-0.916	-0.752	.649	.514	.426	.386	.331	.202		
.08	-0.817	-1.482	-1.147	-1.080	-0.887	-0.759	.610	.504	.426	.398	.337	.221		
.10	-0.727	-1.389	-1.127	-1.071	-0.860	-0.759	.542	.468	.408	.381	.342	.224		
.13	-0.604	-1.440	-1.171	-1.049	-0.822	-0.765	.437	.383	.358	.350	.320			
.19	-0.346	-1.268	-1.066	-0.816	-0.762									
.20	-0.537	-1.110	-1.050	-1.072	-0.802	-0.753	.488							
.25	-0.522	-1.110	-1.050	-1.072	-0.802	-0.753	.459	.401	.356	.340	.307	.210		
.30	-0.506	-0.911	-1.200	-1.038	-0.803	-0.745	.418	.383	.327	.319	.291	.162		
.35	-0.521	-0.786	-1.169	-0.946	-0.776	-0.745	.395	.354	.314	.288	.268	.014		
.40	-0.492	-0.708	-1.123	-0.891	-0.766	-0.748	.366	.336	.299	.267	.240	.147		
.45	-0.559	-0.677	-1.095	-0.891	-0.780	-0.742	.322	.305	.253	.250	.202	.158		
.50	-0.536	-0.643	-1.056	-0.873	-0.718	-0.738	.309	.281	.247	.222	.193	.144		
.55	-0.544	-0.598	-0.959	-0.891	-0.854	-0.742	.264	.243	.228	.214	.175			
.60	-0.546	-0.518	-0.927	-0.864	-0.844	-0.722	.249	.246	.216	.210	.132	.116		
.65	-0.504	-0.480	-0.921	-0.880	-0.860	-0.867	.241	.219	.189	.167		.119		
.68														
.70	-0.447	-0.401	-0.912	-0.915	-0.867	-0.699	.215	.201	.157	.116	.096	.076		
.73	-0.391	-0.366	-0.874	-0.847			.187	.167	.112					
.77														
.79														
.80	-0.355	-0.418			-0.833	-0.677	.158	.140						
.81					-0.903									
.82														
.83														
.85	-0.374	-0.465	-0.849	-0.818			.110	.052	.065					
.87	-0.461	-0.847	-0.861					.068	.069	.049				
.88	-0.471								.060					
.89	-0.483								.057					
.90	-0.428	-0.838		-0.801	-0.662		.060		.058		.012	-0.043		
.91	-0.471		-0.802					.046		.017				
.93	-0.465	-0.805						.022	.002					
.94	-0.422	-0.462			-0.755		.000	-0.014				-0.050		
.96					-0.760									
.97					-0.748									
.98	-0.436								-0.050					
								-0.072						
Stations, fraction of semispan							Stations, fraction of semispan							
x/c	0.10	0.32	0.53	0.65	0.74	0.90	0.10	0.32	0.53	0.65	0.74	0.90		
.00	-0.148	-1.365	-1.195	-0.940	-0.909	-0.834	.432	.404	.267	.164	.089	-0.122		
.01	-0.548	-1.351	-1.189	-0.962	-0.928	-0.834	.648	.526	.354	.302	.205	.006		
.03	-0.933	-1.324	-1.189	-0.972	-0.941	-0.814	.747	.570	.441	.383	.306	.127		
.05	-1.071	-1.301	-1.181	-0.976	-0.954	-0.809	.711	.561	.453	.404	.343	.200		
.08	-0.955	-1.460	-1.170	-0.957	-0.954	-0.809	.664	.557	.458	.420	.356	.227		
.10	-0.833	-1.420	-1.136	-0.957	-0.952	-0.804	.585	.525	.443	.407	.365	.240		
.15	-0.688	-1.450	-1.192	-0.992	-0.958	-0.796	.494	.420	.385	.376	.323			
.19	-1.382	-1.245	-1.044	-0.982	-0.784									
.20	-0.608	-1.215	-1.146	-1.025	-0.975	-0.769	.334							
.30	-0.590	-1.215	-1.146	-1.025	-0.975	-0.769	.208	.459	.396	.346	.339	.233		
.35	-0.563	-1.053	-1.095	-1.002	-0.974	-0.739	.467	.453	.379	.347	.323	.221		
.40	-0.574	-0.926	-1.091	-1.002	-0.969	-0.765	.443	.408	.356	.322	.301	.215		
.45	-0.542	-0.882	-1.116	-0.998	-0.939	-0.765	.411	.387	.340	.307	.276	.198		
.50	-0.594	-0.743	-1.121	-0.998	-0.928	-0.754	.369	.377	.313	.279	.264	.190		
.55	-0.584	-0.700	-1.104	-0.981	-0.927	-0.746	.354	.341	.290	.251	.246	.175		
.56	-0.561	-0.637	-1.065	-0.996	-0.913	-0.719	.332	.312	.268	.241	.215			
.60	-0.540	-0.618	-1.018	-0.938	-0.917	-0.731	.311	.293	.254	.220	.170	.147		
.65	-0.497	-0.650	-1.003	-0.898	-0.898	-0.788	.285	.262	.222	.190		.112		
.70	-0.505	-0.662	-0.987	-0.966	-0.906	-0.711	.248	.241	.168	.138	.131	.105		
.75	-0.544	-0.665	-0.943	-0.885	-0.883		.230	.202	.147					
.77														
.79														
.80	-0.604	-0.652		-0.881	-0.693		.196	.173			.108	.048		
.81				-0.957							.100			
.82											.107			
.83					-0.910						.097	.093		
.85	-0.585	-0.613	-0.915	-0.926	-0.858		.140	.074	.089			.083		
.87	-0.579	-0.911						.093	.081	.067				
.88	-0.572								.085					
.89	-0.555								.080					
.90	-0.565		-0.892	-0.881	-0.834	-0.670	.066		.062		.043	-0.041		
.91	-0.551			-0.881				.069		.034				
.93	-0.528	-0.850			-0.778		.014	.047	.026					
.95	-0.518	-0.517			-0.822				.008		-0.028			
.96					-0.785					-0.024				
.97									-0.030					
.98	-0.485								-0.054					

TABLE L- Continued
PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 60° DELTA WING

$M = 0.94$																						
Upper surface				$\alpha = 22.1$			Lower surface															
x/c	Stations, fraction of semispan						Stations, fraction of semispan															
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90										
.00	-.244	-1.395	-1.105	-1.012	-0.974	-0.869	.419	.406	.262	.147	.063	-.160										
.01	-.595	-1.363	-1.118	-1.037	-0.984	-0.869	.645	.548	.360	.293	.193	-.021										
.03	-.123	-1.358	-1.128	-1.048	-0.996	-0.852	.802	.608	.455	.390	.304	.108										
.05	-.197	-1.406	-1.128	-1.047	-0.998	-0.846	.758	.602	.476	.421	.351	.192										
.08	-.115	-1.480	-1.121	-1.026	-0.993	-0.848	.703	.602	.487	.440	.349	.225										
.10	-.950	-1.435	-1.136	-1.031	-0.988	-0.845	.637	.572	.479	.435	.384	.248										
.15	-.782	-1.442	-1.211	-1.055	-0.990	-0.840	.544	.462	.417	.398	.281											
.19	-.398	-1.475	-1.075	-0.998	-0.834	-0.834	.585															
.20	-.693							.556	.509	.439	.403	.365	.247									
.25	-.651	-1.259	-1.143	-1.057	-0.986	-0.823	.514	.486	.418	.381	.352	.235										
.30	-.598	-1.120	-1.122	-1.045	-0.984	-0.816	.491	.459	.396	.357	.329	.201										
.35	-.598	-1.014	-1.122	-1.042	-0.989	-0.822	.462	.436	.379	.333	.303	.216										
.40	-.560	-.944	-1.135	-1.044	-0.968	-0.819	.414	.408	.355	.317	.292	.209										
.45	-.610	-.896	-1.125	-1.044	-0.956	-0.813	.401	.389	.327	.287	.276	.193										
.50	-.639	-.842	-1.118	-1.010	-0.950	-0.806	.375	.360	.307	.278	.243											
.55	-.660	-.792	-1.095	-1.021	-0.945	-0.769	.294	.285	.223	.171	.160	.125										
.60	-.670	-.757	-1.062	-.954	-0.945	-0.787	.327	.306	.257	.226	.183											
.65	-.634	-.755	-1.055	-.975	-.918																	
.68							.933															
.70	-.617	-1.730	-1.026	-.981	-.938	-0.769	.265	.241	.175													
.73	-.619	-1.718	-.973					.965														
.75							.965															
.79	-.662	-1.701					.919	-.745	.226	.211												
.80							.977															
.81							.931															
.82							.921	-.970														
.83							.967	-.935	-.891	.173	.105	.115										
.85	-.635	-.667	-.921	-.970																		
.87	-.630	-.630	-.930	-.979																		
.88	-.621																					
.89	-.604																					
.90	-.599	-.820					-.860	-.717	.116	.084	.101	.058	.059	-.026								
.91	-.599	-.859	-.907					-.860	-.717	.028	.047	.073										
.93	-.578	-.895					-.811															
.94	-.560	-.572					-.841															
.96							-.845															
.98	-.545																					
$\alpha = 24.1$																						
.00	-.347	-1.411	-1.131	-1.054	-1.010	-0.846	.391	.404	.250	.125	.032	-.214										
.01	-.451	-1.354	-1.141	-1.070	-1.026	-0.846	.673	.561	.375	.282	.171	-.043										
.03	-.274	-1.363	-1.150	-1.084	-1.038	-0.931	.833	.639	.466	.389	.297	.079										
.05	-.132	-1.431	-1.149	-1.078	-1.041	-0.924	.790	.642	.496	.428	.353	.177										
.08	-.123	-.147	-1.158	-1.065	-1.035	-0.927	.735	.644	.512	.454	.376	.214										
.10	-.059	-.1434	-1.194	-1.077	-1.033	-0.923	.672	.618	.512	.455	.397	.239										
.15	-.895	-.1439	-1.240	-1.121	-1.043	-0.922	.590	.493	.443	.414												
.19	-.390	-1.199	-1.136	-1.063	-0.914	-0.619	.594	.557	.473	.431	.382	.249										
.20	-.660							.553	.533	.451	.409	.371	.238									
.25	-.652	-1.267	-1.185	-1.114	-1.050	-0.904	.530	.505	.435	.388	.352											
.30	-.639	-1.146	-1.173	-1.101	-1.048	-0.895	.502	.484	.412	.363	.327	.225										
.35	-.653	-.055	-1.143	-1.088	-1.042	-0.894	.456	.452	.390	.344	.314	.216										
.40	-.633	-.988	-1.167	-1.085	-1.028	-0.901	.438	.433	.366	.317	.295	.201										
.45	-.693	-.942	-1.158	-1.086	-1.017	-0.894	.410	.403	.339	.301	.264											
.50	-.715	-.885	-1.148	-1.054	-1.014	-0.886	.390	.378	.321	.280	.219	.178										
.55	-.724	-.845	-1.135	-1.072	-1.008	-0.864	.363	.345	.291	.248												
.60	-.733	-.817	-1.111	-1.010	-0.908	-0.864	.329	.319	.254	.195	.179	.132										
.65	-.712	-.827	-1.101	-1.043	-.982																	
.68																						
.70	-.706	-1.800	-1.082	-1.046	-1.002	-0.841	.329	.319	.254	.195												
.73																						
.75	-.710	-1.788	-1.034					.995	.294	.274	.206											
.77																						
.79																						
.80	-.738	-.771					-.990	-.810	.258	.240												
.81																						
.82																						
.83																						
.85	-.700	-.742	-.975					-.960	.193	.130	.138											
.87	-.705	-.765	-.965	-1.029					.193	.182	.130	.111										
.88	-.702																					
.89	-.687																					
.90	-.656	-.952					-.909	-.779	.135	.102	.029	.069	.066	-.024								
.91	-.685	-.964																				

TABLE I-- Continued

PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 60° DELTA WING

 $\alpha = 0.96$ $\alpha = 0.2$

Upper surface							Lower surface							
Stations, fraction of semispan							Stations, fraction of semispan							
x/c	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90		
.00	.185	.179	.093	-.731	-.511		.174	.143	.148	.7-274	.154	.055		
.01	.070	-.175	-.232	-.4256	-.441	-.628	.174	.107	1.251	-7.655	.115	.137		
.03	.029	-.147	-.147	-.249	-.419	-.580	.174	.093	.081	7.925	.092	.101		
.05	-.065	-.07	-.171	-.1401	-.362	-.474	.147	.095	.081	7.925	.092	.101		
.08	-.028	-.073	-.183	-.1704	-.359	-.535	.147	.081	.053	8-157	.067	.054		
.10	-.028	-.077	-.160	-.1448	-.309	-.536	.101	.066	.038	8-224	.040	.031		
.15	-.044	-.086	-.151	-.1534	-.271	-.525	.079	.055	.032	8-426	.017	.002		
.19					-.146	-.4601	-.289	-.525	.033	.003	8-682	.027	-.007	
.20	-.048						.039							
.25	-.068	-.103	-.166	-.1475	-.252	-.497	.051	.010	-.012	8-740	-.022	-.018		
.30	-.074	-.105	-.154	-.1471	-.259	-.458	.028	-.003	-.018	8-901	-.035	-.019		
.35	-.086	-.124	-.163	-.1498	-.259	-.409	.023	-.023	-.040	9-049	-.052	-.043		
.40	-.066	-.137	-.182	-.1561	-.253	-.378	-.002	-.028	-.048	9-152	-.064	-.075		
.45	-.136	-.136	-.191	-.1614	-.238	-.355	-.039	-.047	-.066	9-247	-.071	-.097		
.50	-.125	-.144	-.201	-.1659	-.269	-.348	-.033	-.048	-.077	9-377	-.071	-.108		
.55	-.125	-.158	-.211	-.1726	-.276	-.342	-.049	-.061	-.082	9-287	-.096			
.60	-.141	-.168	-.188	-.1578	-.279	-.342	-.047	-.064	-.076	9-256	-.109	-.099		
.65	-.154	-.181	-.202	-.1601	-.258	-.342	-.059	-.079	-.081	9-314	-.108			
.68														
.70	-.154	-.176	-.207	-.1798	-.266	-.227	-.070	-.076	-.081	9-439	-.145	-.105		
.73														
.75	-.161	-.182	-.190	-.231			-.063	-.084	-.092					
.77														
.80	-.175	-.170	-.1578	-.1439	-.165	-.050	-.071	-.072		-.9-552				
.81														
.82														
.83														
.85	-.150	-.151	-.136	-.1040			-.067	-.095	-.070					
.87														
.88														
.89														
.90	-.149		-.049	-.022	-.001	-.033	-.110		-.041		-.010	-.022		
.91														
.93														
.95	-.066	-.024					-.067	-.069	-.012					
.96														
.97														
.98														

a = 2.2														
Stations, fraction of semispan														
x/c	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90		
.00	.185	.179	.093	-.731	-.511		.174	.143	.148	.7-274	.154	.055		
.01	.070	-.175	-.232	-.4256	-.441	-.628	.174	.107	1.251	-7.655	.115	.137		
.03	.029	-.147	-.147	-.249	-.419	-.580	.174	.093	.081	7.925	.092	.101		
.05	-.065	-.07	-.171	-.1401	-.362	-.474	.147	.095	.081	7.925	.092	.101		
.08	-.028	-.073	-.183	-.1704	-.359	-.535	.147	.081	.053	8-157	.067	.054		
.10	-.028	-.077	-.160	-.1448	-.309	-.536	.101	.066	.038	8-224	.040	.031		
.15	-.044	-.086	-.151	-.1534	-.271	-.525	.079	.055	.032	8-426	.017	.002		
.19					-.146	-.4601	-.289	-.525	.033	.003	8-682	.027	-.007	
.20	-.048						.039							
.25	-.068	-.103	-.166	-.1475	-.252	-.497	.051	.010	-.012	8-740	-.022	-.018		
.30	-.074	-.105	-.154	-.1471	-.259	-.458	.028	-.003	-.018	8-901	-.035	-.019		
.35	-.086	-.124	-.163	-.1498	-.259	-.409	.023	-.023	-.040	9-049	-.052	-.043		
.40	-.066	-.137	-.182	-.1561	-.253	-.378	-.002	-.028	-.048	9-152	-.064	-.075		
.45	-.136	-.136	-.191	-.1614	-.238	-.355	-.039	-.047	-.066	9-247	-.071	-.097		
.50	-.125	-.144	-.201	-.1659	-.269	-.348	-.033	-.048	-.077	9-377	-.071	-.108		
.55	-.125	-.158	-.211	-.1726	-.276	-.342	-.049	-.061	-.082	9-287	-.096			
.60	-.141	-.168	-.188	-.1578	-.279	-.342	-.047	-.064	-.076	9-256	-.109	-.099		
.65	-.154	-.181	-.202	-.1601	-.258	-.342	-.059	-.079	-.081	9-314	-.108			
.68														
.70	-.154	-.176	-.207	-.1798	-.266	-.227	-.070	-.076	-.081	9-439	-.145	-.105		
.73														
.75	-.161	-.182	-.190	-.231			-.063	-.084	-.092					
.77														
.80	-.175	-.170	-.1578	-.1439	-.165	-.050	-.071	-.072		-.9-552				
.81														
.82														
.83														
.85	-.150	-.151	-.136	-.1040			-.067	-.095	-.070					
.87														
.88														
.89														
.90	-.149		-.049	-.022	-.001	-.033	-.110		-.041		-.010	-.022		
.91														
.93														
.95	-.066	-.024					-.067	-.069	-.012					
.96														
.97														
.98	-.002													

CONFIDENTIAL

TABLE I.- Continued

PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 60° DELTA WING

 $M = 0.96$ $a = 4:3$

x/c	Upper surface					Lower surface						
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90
.00	.180	-.159	-.547	-.526	-.710	-.707	.253	.234	.241	.244	.231	.102
.01	-.004	-.392	-.580	-.633	-.676	-.676	.239	.196	.220	.206	.192	.182
.03	-.048	-.321	-.575	-.677	-.686	-.678	.219	.173	.174	.184	.173	.170
.05	-.081	-.214	-.569	-.541	-.690	-.678	.199	.160	.147	.145	.144	.135
.08	-.105	-.198	-.589	-.457	-.698	-.682	.160	.149	.126	.131	.114	.112
.10	-.097	-.191	-.548	-.419	-.700	-.679	.138	.114	.092	.090	.088	.073
.15	-.113	-.173	-.405	-.413	-.728	-.679	.088	.064	.053	.056	.067	
.19	-.171	-.268	-.398	-.398	-.752	-.678						
.20	-.112						.113					
.25	-.129	-.176	-.229	-.574	-.673	-.678	.100	.065	.042	.047	.041	.027
.30	-.129	-.176	-.206	-.367	-.590	-.674	.080	.049	.032	.035	.024	.017
.35	-.144	-.131	-.200	-.361	-.494	-.578	.060	.030	.009	.001	.004	-.015
.40	-.118	-.203	-.234	-.450	-.334	-.685	.041	.020	.003	-.017	-.012	-.031
.45	-.197	-.201	-.250	-.359	-.263	-.685	.001	-.001	-.017	-.024	-.023	-.055
.50	-.181	-.207	-.245	-.345	-.472	-.682	.007	-.003	-.033	-.047	-.030	-.037
.55	-.188	-.222	-.270	-.335	-.257	-.682	.011	-.020	-.042	-.041	-.051	
.60	-.200	-.226	-.238	-.313	-.307	-.641	.009	-.024	-.038	-.043	-.080	-.056
.65	-.207	-.244	-.252	-.280	-.278	-.622	.022	-.040	-.047	-.057	-.116	
.68					-.271							
.70	-.209	-.233	-.267	-.333	-.290	-.550	-.032	-.038	-.055	-.084	-.141	-.080
.73					-.282							
.75	-.219	-.237	-.254		-.274		-.034	-.052	-.075			
.77					-.302					-.125		
.79					-.290					-.097		
.80	-.233	-.225			-.285	-.435	-.050	-.048			-.055	-.104
.81										-.077		
.82					-.236					-.110		
.83					-.221	-.263				-.086	-.064	
.85	-.210	-.214	-.230		-.212	-.204	-.057	-.097	-.073		-.043	
.87					-.198	-.220			-.084	-.065	-.043	
.88					-.168				-.081			
.89					-.177				-.079			
.90	-.229				-.161				-.097	-.047		
.91					-.176	-.120			-.079	-.023		
.93					-.138	-.070			-.074	-.027		
.95	-.136				-.089		.024		-.081	-.063		
.96											.005	
.97						-.034					-.009	
.98						-.001					-.006	
										-.042		

 $a = 6:3$

x/c	0.172	-.549	-.787	-.821	-.726	-.766	.287	.307	.276	.248	.222	.101
.01	-.002	-.662	-.605	-.729	-.714	-.766	.303	.272	.267	.244	.222	.174
.03	-.128	-.619	-.613	-.758	-.723	-.747	.294	.247	.233	.232	.218	.187
.05	-.159	-.631	-.614	-.703	-.727	-.747	.269	.223	.206	.195	.193	.167
.08	-.200	-.578	-.621	-.672	-.731	-.753	.269	.223	.206	.195	.193	.151
.10	-.172	-.385	-.621	-.656	-.735	-.751	.227	.206	.187	.181	.168	.113
.15	-.188	-.168	-.624	-.660	-.753	-.751	.198	.168	.153	.140	.143	.105
.19	-.192	-.623	-.661	-.784	-.747		.142	.123	.110	.139	.105	
.20	-.174						.170					
.25	-.199	-.213	-.603	-.650	-.792	-.742	.157	.113	.099	.090	.092	.064
.30	-.192	-.215	-.524	-.639	-.801	-.736	.132	.097	.085	.065	.071	.049
.35	-.208	-.225	-.440	-.614	-.798	-.729	.113	.075	.062	.042	.051	.041
.40	-.168	-.247	-.391	-.590	-.792	-.729	.089	.067	.049	.023	.030	.003
.45	-.239	-.245	-.357	-.563	-.766	-.713	.052	.044	.033	.009	.014	.001
.50	-.231	-.252	-.344	-.521	-.756	-.695	.054	.039	.014	.012	.007	-.009
.55	-.233	-.269	-.331	-.492	-.726		.035	.022	.008	-.009	-.006	
.60	-.254	-.272	-.287	-.446	-.695	-.644	.052	.016	.004	-.018	-.035	-.022
.65	-.257	-.291	-.301	-.400	-.614		.019	.001	-.009	-.033	-.105	
.68					-.605					-.079		
.70	-.261	-.282	-.313	-.427	-.591	-.598	.002	-.005	-.021	-.070	-.081	-.052
.73					-.561					-.018		
.75	-.271	-.289	-.299		-.520		-.001	-.023	-.050		-.051	
.77					-.389					-.080		
.79					-.370					-.062		
.80	-.286	-.279			-.447	-.556	-.020	-.025			-.042	-.088
.81					-.363					-.057		
.82					-.284					-.061		
.83					-.274	-.342				-.051	-.056	
.85	-.271	-.270	-.287		-.347		-.031	-.077	-.049		-.044	
.87					-.280	-.298			-.059	-.048	-.053	
.88					-.249				-.039			
.89					-.240				-.060			
.90	-.290	-.248			-.240	-.526	-.053		-.045		-.049	-.139
.91		-.244			-.194				-.064		-.046	
.93		-.228			-.167				-.068	-.036		
.95	-.212	-.183			-.094				-.081	-.071		-.054
.96					-.058					-.011		
.98					-.103					-.066		

TABLE L- Continued

X = 0.95

Upper surface							Lower surface						
Stations, fraction of semispan						Stations, fraction of semispan							
x/c	0.19	0.32	0.53	0.65	0.74	0.90	x/c	0.19	0.32	0.53	0.65	0.74	0.90
.00	-.180	-.710	-.779	-.804	-.752		.00						
.01	-.139	-.688	-.670	-.795	-.753	-.479	.01	.392	.314	.308	.257	.225	.092
.03	-.179	-.710	-.681	-.800	-.760	-.479	.03	.378	.298	.308	.276	.247	.146
.05	-.239	-.806	-.685	-.759	-.747	-.477	.05	.374	.278	.292	.278	.237	.203
.08	-.266	-.833	-.714	-.734	-.766	-.477	.08	.353	.247	.268	.248	.242	.198
.10	-.246	-.792	-.730	-.732	-.770	-.477	.10	.309	.231	.253	.236	.220	.188
.15	-.234	-.207	-.779	-.748	-.787	-.477	.15	.273	.195	.217	.197	.196	.154
.19	-.209	-.803	-.762	-.809	-.477		.19	.169	.186	.167	.193	.149	
.20	-.224						.20	.237					
.25	-.243	-.224	-.826	-.775	-.821	-.477	.25	.220	.142	.159	.146	.147	.110
.30	-.230	-.234	-.818	-.784	-.849	-.476	.30	.192	.127	.145	.123,	.127	.091
.35	-.245	-.245	-.792	-.789	-.860	-.476	.35	.174	.108	.122	.096	.104	.091
.40	-.208	-.267	-.736	-.798	-.871	-.476	.40	.151	.096	.107	.075	.077	.051
.45	-.284	-.267	-.659	-.802	-.869	-.478	.45	.109	.075	.090	.061	.068	.045
.50	-.265	-.274	-.546	-.778	-.876	-.477	.50	.089	.068	.071	.058	.050	.033
.55	-.269	-.289	-.413	-.783	-.882		.55	.083	.051	.051	.037	.034	
.60	-.289	-.294	-.262	-.734	-.889	-.636	.60	.066	.042	.054	.027	.014	.019
.65	-.294	-.306	-.265	-.709	-.865		.65	.037	.008			.056	
.68							.68					.039	
.70	-.294	-.303	-.286	-.720	-.810	-.579	.70	.050	.022	.023	-.032	-.041	-.014
.75	-.301	-.306	-.285				.75	.039	.002	-.011		-.016	
.77							.77						
.79							.79						
.80	-.317	-.301					.80	.022	-.005				
.81							.81						
.82							.82						
.83							.83						
.85	-.302	-.293	-.280				.85	.002	-.057				
.87							.87						
.88							.88						
.89							.89						
.90	-.323		-.269				.90						
.91							.91						
.93							.93						
.95	-.265	-.246					.95						
.96							.96						
.97							.97						
.98							.98						
	-.192												
a = 10x3							a = 10x3						
.00	.158	-.960	-.618	-.829	-.833		.00	.436	.384	.309	.253	.203	.056
.01	-.225	-.833	-.790	-.829	-.840	-.864	.01	.266	.220	.201	.197	.178	.128
.03	-.284	-.876	-.794	-.839	-.849	-.838	.03	.295	.201	.184	.171	.140	.112
.05	-.356	-.972	-.793	-.827	-.853	-.837	.05	.214	.177	.159	.144	.135	.114
.08	-.372	-.1041	-.821	-.818	-.853	-.845	.08	.119	.337	.306	.291	.266	.201
.10	-.334	-.1095	-.843	-.816	-.850	-.845	.10	.365	.320	.290	.284	.251	.197
.15	-.322	-.774	-.876	-.843	-.864	-.847	.15	.326	.282	.258	.245	.231	.170
.19	-.510	-.931	-.859	-.895	-.847		.19	.252	.229	.217	.228	.168	
.20	-.289						.20	.288					
.25	-.308	-.267	-.1026	-.877	-.890	-.836	.25	.266	.220	.201	.197	.178	.128
.30	-.290	-.277	-.1020	-.898	-.906	-.830	.30	.295	.201	.184	.171	.140	.112
.35	-.304	-.301	-.1019	-.924	-.918	-.830	.35	.214	.177	.159	.144	.135	.114
.40	-.267	-.320	-.1024	-.949	-.937	-.846	.40	.191	.163	.142	.121	.112	.074
.45	-.335	-.323	-.982	-.969	-.952	-.849	.45	.147	.139	.124	.107	.098	.068
.50	-.318	-.335	-.878	-.954	-.974	-.844	.50	.144	.129	.107	.083	.077	.056
.55	-.325	-.350	-.700	-.976	-.970		.55	.122	.110	.095	.079	.059	
.60	-.348	-.354	-.503	-.921	-.1018	-.747	.60	.114	.097	.083	.084	.041	.035
.65	-.347	-.367	-.426	-.949	-.1009		.65	.094	.077	.066	.041	.045	
.68							.68						
.70	-.347	-.365	-.353	-.964	-.1023	-.654	.70	.074	.067	.046	.002	.021	.001
.75	-.360	-.373	-.334				.75	.062	.043	.012		.004	
.77							.77						
.79							.79						
.80	-.378	-.365					.80	.042	.033				
.81							.81						
.82							.82						
.83							.83						
.85	-.361	-.356	-.318				.85	.020	.029				
.87	-.337	-.318	-.802				.87	.013	.012				
.88							.88						
.89							.89						
.90	-.372	-.295					.90	.013					
.91	-.354		-.717				.91						
.93	-.330	-.293					.93						
.95	-.323	-.311					.95						
.96							.96						
.97							.97						
.98							.98						

TABLE L- Continued

PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 60° DELTA WING

 $M = 0.96$ $\alpha = 12.4$

Upper surface							Lower surface						
Stations, fraction of semispan							Stations, fraction of semispan						
x/c	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	
.00	.144	-1.025	-1.875	-1.872	-1.890		.483	.405	.311	.240	.188	.017	
.01	-.300	-1.970	-1.868	-1.879	-1.902	-1.922		.503	.431	.347	.268	.249	.109
.03	-.350	-1.983	-1.874	-1.894	-1.916	-1.905		.523	.425	.347	.339	.294	.184
.05	-.530	-1.029	-1.875	-1.888	-1.921	-1.899		.488	.394	.337	.323	.293	.209
.08	-.465	-1.110	-1.905	-1.884	-1.924	-1.908		.445	.379	.306	.322	.285	.211
.10	-.418	-1.121	-1.929	-1.890	-1.922	-1.908		.399	.341	.278	.288	.270	.194
.15	-.372	-1.132	-1.958	-1.930	-1.925	-1.912		.310	.249	.261	.269	.194	
.19													
.20	-.339												
.25	-.342	-1.368	-1.032	-1.944	-1.950	-1.902		.331	.276	.230	.238	.223	.161
.30	-.326	-1.322	-1.062	-1.970	-1.977	-1.888		.293	.257	.206	.212	.207	.140
.35	-.347	-1.334	-1.110	-1.983	-1.983	-1.873		.271	.233	.184	.187	.181	.145
.40	-.309	-1.359	-1.186	-1.994	-1.991	-1.895		.245	.216	.169	.165	.155	.108
.45	-.370	-1.352	-1.174	-1.037	-1.986	-1.902		.205	.187	.147	.147	.140	.099
.50	-.358	-1.362	-1.115	-1.042	-1.993	-1.908		.195	.179	.131	.122	.120	.087
.55	-.362	-1.387	-1.045	-1.074	-1.005			.173	.156	.122	.118	.092	
.60	-.385	-1.396	-1.012	-1.019	-1.032	-1.899		.162	.144	.103	.100	.051	.068
.65	-.391	-1.406	-1.075	-1.117	-1.011			.142	.121	.079	.076	.048	
.66													
.70	-.391	-1.400	-1.053	-1.145	-1.116	-1.772		.119	.109	.044	.032	.015	.030
.75	-.396	-1.410	-1.025					.102	.082	.019			.026
.77													
.79													
.80	-.415	-1.405				-1.152	-1.702		.082	.069			.022
.81													-.019
.82													
.83													
.85	-.404	-1.398	-1.430	-1.082									
.87													
.88													
.89													
.90	-.418	-1.385											
.91	-.574												
.93	-.371	-1.369											
.95	-.370	-1.356											
.96													
.97													
.98													

 $\alpha = 14.3$

.00	-.099	-1.099	-1.946	-1.963	-1.977		.459	.411	.306	.202	.153	-.037	
.01	-.390	-1.045	-1.946	-1.975	-1.993	-1.998		.557	.463	.355	.293	.234	.045
.03	-.451	-1.070	-1.954	-1.990	-1.003	-1.891		.585	.471	.384	.343	.294	.158
.05	-.614	-1.098	-1.952	-1.991	-1.008	-1.983		.547	.443	.379	.336	.311	.201
.08	-.569	-1.196	-1.972	-1.991	-1.010	-1.986		.520	.433	.372	.339	.305	.213
.10	-.521	-1.261	-1.968	-1.991	-1.006	-1.988		.459	.391	.347	.312	.298	.202
.15	-.445	-1.290	-1.061	-1.999	-1.015	-1.992							
.19	-.1064	-1.131	-1.022	-1.034	-1.995								
.20	-.401												
.25	-.398	-1.650	-1.119	-1.042	-1.009	-1.990		.385	.326	.245	.253	.174	
.30	-.382	-1.482	-1.098	-1.096	-1.027	-1.978		.347	.304	.271	.240	.233	.156
.35	-.402	-1.435	-1.117	-1.106	-1.054	-1.964		.324	.280	.246	.214	.210	.015
.40													
.45	-.362	-1.455	-1.148	-1.133	-1.082	-1.971		.294	.262	.225	.190	.184	.127
.45	-.423	-1.426	-1.174	-1.155	-1.108	-1.973		.252	.234	.206	.171	.170	.121
.50	-.417	-1.421	-1.251	-1.112	-1.146	-1.979		.242	.221	.185	.145	.159	.108
.55	-.424	-1.442	-1.226	-1.132	-1.162			.215	.198	.165	.140	.121	
.60	-.439	-1.455	-1.148	-1.064	-1.144	-1.987		.204	.182	.156	.120	.077	.086
.65	-.451	-1.478	-1.064	-1.126	-1.103			.182	.157	.133	.095	.018	
.68													
.70	-.455	-1.466	-1.085	-1.147	-1.100	-1.945		.160	.144	.107	.047	.043	.046
.73													
.75	-.461	-1.471	-1.860		-1.094			.138	.114	.070			.051
.77													
.80	-.476	-1.462			-1.095	-1.723		.114	.099				
.81													
.82													
.83													
.85													
.87													
.88													
.89													
.90	-.474												
.91	-.433												
.93	-.433	-1.209											
.95	-.435	-1.423											
.96													
.97													
.98													

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TABLE L - Continued

X = 0.96

$\alpha = 17.8$

8.3.2019

TABLE L - Continued
PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 60° DELTA WING

$M = 0.96$

$\alpha = 22.1$

Upper surface							Lower surface						
Stations, fraction of semispan							Stations, fraction of semispan						
x/c	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	
.00	-.214	-1.346	-1.186	-1.131	-1.063		.482	.423	.275	.156	.077	-.143	
.01	-.350	-1.320	-1.188	-1.138	-1.063	-1.051	.679	.559	.379	.208	.105	-.021	
.03	-1.040	-1.413	-1.190	-1.157	-1.076	-1.032	.816	.618	.463	.357	.314	.107	
.05	-1.141	-1.420	-1.196	-1.152	-1.085	-1.025	.769	.614	.490	.424	.362	.195	
.08	-1.057	-1.390	-1.184	-1.135	-1.082	-1.029	.715	.614	.494	.446	.379	.224	
.10	-.902	-1.385	-1.187	-1.139	-1.076	-1.029	.647	.583	.492	.444	.398	.248	
.15	-.752	-1.393	-1.252	-1.132	-1.079	-1.031	.594	.471	.427	.407		.265	
.19	-.354	-1.324	-1.177	-1.087	-1.030								
.20	-.670												
.25	-.629	-1.237	-1.287	-1.121	-1.022	-1.018	.567	.521	.449	.412	.375	.248	
.30	-.582	-1.112	-1.264	-1.136	-1.010		.529	.497	.428	.390	.362	.233	
.35	-.605	-1.006	-1.253	-1.246	-1.175	-1.025	.503	.468	.409	.367	.342	-.017	
.40	-.583	-.929	-1.248	-1.228	-1.175	-1.031	.477	.450	.385	.343	.315	.222	
.45	-.662	-.870	-1.243	-1.212	-1.175	-1.027	.430	.419	.366	.325	.300	.212	
.50	-.651	-.815	-1.259	-1.152	-1.182	-1.017	.417	.401	.342	.296	.284	.200	
.55	-.655	-.781	-1.279	-1.190	-1.181		.385	.371	.316	.285	.252		
.60	-.655	-.745	-1.276	-1.099	-1.179	-1.002	.349	.340	.302	.241	.204	.175	
.65	-.638	-.735	-1.260	-1.181	-1.157		.345	.319	.271	.231	.146		
.68												.193	
.70	-.623	-.718	-1.243	-1.191	-1.163	-0.980	.307	.295	.234	.179	.164	.132	
.73												.186	
.75	-.618	-.710	-1.172		-1.163		.274	.254	.186		.161		
.77											.149		
.79											.146		
.80	-.656	-.703			-1.159	-0.960	.243	.225			.141	.080	
.81					-1.191						.138		
.82													
.83													
.85	-.642	-.693	-1.126	-1.130			.184	.114	.139	.130			
.87	-.662	-1.114	-1.222					.139	.126	.104			
.88	-.660							.130					
.89	-.647							.125					
.90	-.623												
.91	-.646												
.93	-.636	-1.058	-1.185					.113	.099	.068	.071	-.008	
.95	-.613	-.634			-1.154	-0.940	.055	.050			.005		
.97					-1.026								
.98	-.622												

$\alpha = 24.3$

.00	-.298	-1.314	-1.155	-1.069	-.979								
.01	-.625	-1.296	-1.149	-1.069	-.999	-.949	.412	.421	.273	.147	.063	-.182	
.03	-1.196	-1.277	-1.163	-1.072	-1.012	-.943	.688	.577	.395	.302	.198	-.034	
.05	-1.223	-1.326	-1.135	-1.063	-1.014	-.938	.845	.649	.482	.411	.321	.102	
.08	-1.145	-1.310	-1.147	-1.088	-1.012	-.941	.799	.653	.516	.447	.374	.197	
.10	-.898	-1.338	-1.139	-1.054	-1.006	-.941	.747	.654	.528	.472	.398	.232	
.15	-.848	-1.342	-1.235	-1.190	-1.013	-.941	.690	.628	.528	.474	.419	.260	
.19	-.572	-1.292	-1.238	-1.146	-1.064	-.938	.599	.510	.461	.435	.280		
.20	-.731												
.25	-.670	-1.208	-1.197	-1.150	-1.082	-.931	.607	.567	.491	.449	.406	.268	
.30	-.628	-1.117	-1.182	-1.136	-1.090	-.916	.567	.543	.468	.430	.395	.256	
.35	-.653	-1.028	-1.159	-1.104	-1.065	-.898	.543	.517	.455	.408	.373	-.071	
.40	-.633	-.956	-1.145	-1.083	-1.061	-.917	.522	.496	.431	.382	.349	.246	
.45	-.689	-.903	-1.136	-1.070	-1.042	-.928	.477	.463	.409	.365	.335	.240	
.50	-.697	-.856	-1.130	-1.027	-1.039	-.931	.457	.445	.385	.336	.320	.229	
.55	-.700	-.825	-1.165	-1.050	-1.025	-.929	.429	.416	.360	.327	.285		
.60	-.710	-.794	-1.154	-.991	-1.018	-.896	.409	.393	.344	.302	.241	.202	
.65	-.701	-.791	-1.137	-1.038	-1.030	-.861	.381	.358	.310	.271	.183		
.68													
.70	-.686	-.770	-1.133	-1.045	-1.003	-.857	.344	.332	.273	.218	.204	.160	
.73													
.75	-.682	-.768	-1.090			-.998	.312	.289	.227		.193		
.77													
.79													
.80	-.719	-.754			-1.012	-.987	-.813	.278	.256				
.81													
.82													
.83													
.85	-.699	-.748	-1.028		-1.007	-.975	.213	.145	.164				
.87	-.711	-1.017		-1.004				.168	.154	.135			
.88	-.710												
.89	-.700												
.90	-.673												
.91	-.704												
.93	-.709	-.977											
.95	-.653	-.695											
.96													
.97													
.98	-.669	-.925											

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TABLE L- Continued

X = 0.98

a = 0.8

6.3 342

TABLE I.- Continued

X = 0.95

$\alpha = -4.2$

Upper surface

Stations, fraction of semispan						
x/c	0.19	0.32	0.53	0.65	0.74	0.80
.00	-0.189	-0.119	-0.484	-0.708		
.01	-0.026	-0.363	-0.626	-0.670	-0.693	
.03	-0.019	-0.294	-0.664	-0.678	-0.668	-0.666
.05	-0.051	-0.188	-0.508	-0.681	-0.661	
.08	-0.079	-0.173	-0.420	-0.687	-0.669	
.10	-0.071	-0.169	-0.387	-0.692	-0.665	
.12	-0.094	-0.158	-0.391	-0.710	-0.665	
.15			-0.375	-0.732	-0.661	
.20		-0.001				
.22	-0.119	-0.167	-0.348	-0.648	-0.656	
.30	-0.119	-0.166	-0.339	-0.546	-0.422	
.35	-0.132	-0.178	-0.337	-0.402	-0.408	
.40	-0.107	-0.195	-0.341	-0.326	-0.326	
.45	-0.188	-0.196	-0.345	-0.248	-0.248	-0.655
.50	-0.178	-0.204	-0.332	-0.287	-0.287	-0.654
.55	-0.178	-0.219	-0.329	-0.298	-0.298	
.60	-0.198	-0.224	-0.304	-0.307	-0.307	
.65	-0.204	-0.239	-0.284	-0.285	-0.285	
.68				-0.270		
.70	-0.204	-0.232	-0.340	-0.298	-0.298	-0.610
.75				-0.283		
.75	-0.217	-0.240			-0.292	
.77				-0.306		
.79				-0.296		
.80	-0.233	-0.232			-0.280	-0.528
.83				-0.289		
.85	-0.216	-0.222			-0.299	
.87				-0.285		-0.271
.88				-0.285		
.89				-0.285		
.90	-0.243				-0.259	-0.421
.91		-0.205				
.93		-0.199				
.95	-0.202	-0.188			-0.199	
.96				-0.182		
.98		-0.157				

Lower surface

Stations, fraction of semispan					
0.19	0.32	0.53	0.66	0.74	0.90
+.251	.241		.231	.207	.096
+.251	.204		.198	.186	.178
+.226	.177		.174	.165	.166
+.226	.158		.154	.157	.129
+.176	.153		.123	.109	.107
+.149	.114		.082	.080	.067
	.090		.052	.077	.062
+.123					
+.167	.063		.034	.033	.028
.083	.046		.010	.016	.021
.074	.023		-.010	-.002	-.034
.044	.015		-.050	-.047	-.054
.008	-.006		-.041	-.027	-.075
.008	-.000		-.060	-.029	-.087
-.010	-.026		-.052	-.055	
-.010	-.032		-.053	-.080	-.087
-.026	-.048		-.065	-.115	
-.039	-.050		-.090	-.159	-.104
-.039	-.060			.133	
				-.134	
-.053	-.056		-.148		
			-.134		
-.059	-.104		-.129		
	-.102		-.126		
	-.102				
	-.106				
-.104	-.113		-.096	-.086	-.175
	-.125				
-.122	-.127			-.071	
	-.130		-.075		

6-3

TABLE I.- Continued
PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 60° DELTA WING

x = 0.98

TABLE I.- Continued
PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 60° DELTA WING

 $\kappa = 0.98$ $a = 12.4$

Upper surface							Lower surface						
Stations, fraction of semispan							Stations, fraction of semispan						
x/c	0.19	0.32	0.55	0.65	0.74	0.90	0.19	0.32	0.55	0.65	0.74	0.90	
.00	-0.166	-0.986	-0.857	-0.835	-0.852		.515	.418	.327	.250	.204	.038	
.01	-0.261	-0.911	-0.832	-0.834	-0.861		.522	.446	.365	.312	.264	.122	
.03	-0.342	-0.842	-0.833	-0.855	-0.868		.520	.442	.374	.347	.309	.177	
.05	-0.485	-0.684	-0.847	-0.851	-0.861		.505	.412	.363	.333	.309	.226	
.08	-0.427	-1.049	-0.865	-0.848	-0.863		.464	.399	.349	.330	.298	.226	
.10	-0.378	-1.077	-0.885	-0.857	-0.881		.417	.356	.320	.299	.284	.208	
.15	-0.356	-1.088	-0.916	-0.899	-0.888		.325	.293	.270	.254	.208		
.19	-0.310	-0.950	-0.934	-0.914	-0.877								
.20	-0.306												
.25	-0.310	-0.349	-0.996	-0.909	-0.913	-0.865	.347	.292	.268	.252	.239	.172	
.30	-0.291	-0.308	-1.015	-0.927	-0.933	-0.850	.313	.272	.247	.224	.221	.153	
.35	-0.312	-0.319	-1.058	-0.944	-0.941	-0.834	.291	.247	.223	.200	.197	.163	
.40	-0.281	-0.344	-1.137	-0.964	-0.950	-0.855	.264	.232	.206	.177	.171	.123	
.45	-0.345	-0.338	-1.131	-0.998	-0.948	-0.866	.219	.205	.187	.158	.159	.116	
.50	-0.336	-0.345	-1.079	-0.998	-0.955	-0.871	.212	.193	.167	.133	.146	.104	
.55	-0.340	-0.369	-1.013	-1.030	-0.967		.188	.172	.150	.131	.112		
.60	-0.361	-0.381	-0.890	-0.969	-0.997	-0.879	.180	.160	.143	.119	.067	.089	
.65	-0.370	-0.394	-0.759	-1.059	-1.017		.160	.133	.122	.088	.005		
.68												.048	
.70	-0.370	-0.388	-0.637	-1.100	-1.075	-0.776	.138	.125	.099	.045	.031	.052	
.73	-0.377	-0.397	-0.514				.122	.099	.080		.073		
.75	-0.377	-0.397	-0.514										
.77				-1.078									
.79				-1.078									
.80	-0.396	-0.394			-1.115	-0.684	.100	.084					
.81				-1.056									
.82				-0.431									
.83				-0.410	-1.051								
.85	-0.382	-0.384	-0.414		-1.108		.068	.008	.032				
.87	-0.363	-0.409	-0.409	-1.013			.027	.029	.017				
.88	-0.362												
.89	-0.353												
.90	-0.399	-0.368		-1.101	-0.688		.025		.017		.005	-0.060	
.93	-0.359	-0.359	-0.351	-0.974			.019			-0.003			
.95	-0.356	-0.351			-0.977		.020		.002	-0.002			
.96				-0.938							-0.044		
.97				-0.349							-0.038		
.98				-0.321							-0.033		
												-0.068	

 $a = 14.4$

.00	-0.117	-1.054	-0.917	-0.921	-0.945		.535	.424	.313	.223	.171	.013
.01	-0.350	-1.017	-0.929	-0.957	-0.963		.563	.477	.374	.310	.282	.084
.03	-0.423	-1.022	-0.924	-0.944	-0.970	-0.953	.591	.481	.394	.354	.311	.175
.05	-0.579	-1.051	-0.927	-0.943	-0.975	-0.949	.564	.455	.386	.350	.321	.218
.08	-0.557	-1.146	-0.944	-0.945	-0.975	-0.953	.526	.441	.377	.352	.315	.227
.10	-0.484	-1.201	-0.938	-0.951	-0.974	-0.955	.470	.402	.383	.383	.308	.219
.15	-0.412	-1.230	-1.008	-0.961	-0.980	-0.959	.371	.329	.300	.309	.221	
.19	-0.933	-1.078	-0.982	-0.997	-0.965							
.20	-0.374											
.25	-0.377	-0.619	-1.077	-1.012	-0.973	-0.954	.393	.336	.299	.281	.266	.189
.30	-0.358	-0.458	-1.058	-1.041	-0.992	-0.942	.356	.315	.279	.254	.247	.171
.35	-0.379	-0.410	-1.074	-1.045	-1.067	-0.931	.331	.290	.256	.231	.224	.184
.40	-0.352	-0.432	-1.112	-1.058	-1.043	-0.936	.301	.272	.238	.206	.199	.144
.45	-0.407	-0.419	-1.149	-1.094	-1.065	-0.939	.260	.244	.218	.189	.185	.135
.50	-0.409	-0.401	-1.203	-1.063	-1.093	-0.942	.250	.232	.195	.163	.169	.123
.55	-0.416	-0.421	-1.175	-1.089	-1.109		.228	.208	.177	.156	.158	
.60	-0.432	-0.437	-1.101	-1.012	-1.100	-0.956	.213	.195	.166	.137	.089	.106
.65	-0.440	-0.460	-1.022	-1.085	-1.068		.188	.168	.143	.110	.035	
.68				-1.057							.078	
.70	-0.440	-0.451	-0.950	-1.109	-1.057	-0.939	.167	.156	.118	.063	.055	.070
.73												
.75	-0.450	-0.459	-0.837		-1.067		.149	.125	.078			
.77					-1.124							
.79				-1.154								
.80	-0.469	-0.452			-1.071	-0.775	.122	.111				
.81					-1.177							
.82					-1.726							
.83				-0.681	-1.189							
.85	-0.450	-0.441	-0.663	-1.051			.086	.028	.044			
.87			-0.618	-0.641	-1.171		.048	.039	.023			
.88			-0.417									
.89			-0.409									
.90	-0.459	-0.587			-0.990	-0.701	.043	.028				
.91	-0.415			-1.151				.036				
.93	-0.415	-0.549			-0.896			.015	.003	.002		
.95	-0.426	-0.410			-1.129		.006	-0.012			-0.038	
.96				-0.523							-0.043	
.97				-0.374							-0.038	
.98											-0.063	

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TABLE L- Continued

$K = 0.98$

$\alpha = 17.8$

$a = 20.0$

TABLE L - Continued
PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 60° DELTA WING

 $M = 1.00$ $\alpha = 0.2$

Upper surface						Lower surface						
Stations, fraction of semispan						Stations, fraction of semispan						
x/c	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90
.00	.100	.309	.314	.315	.342		.057	.031	.041	.056	.001	-.189
.01	.063	.016	.007	-.041	-.100	-.227	.057	.022	.036	.021	-.022	-.058
.03	.038	-.003	-.016	-.045	-.094	-.195	.046	.051	.018	.006	-.030	-.042
.05	.013	.021	-.002	-.034	-.078	-.130	.046	.049	.008	-.003	-.026	-.103
.08	.006	.025	-.014	-.039	-.059	-.138	.028	.043	-.001	-.011	-.032	-.103
.10	.014	.019	-.010	-.024	-.063	-.143	.035	.033	-.013	-.026	-.049	-.103
.15	.017	.011	-.011	-.053	-.080	-.152	.025	.028	-.013	-.026	-.049	-.103
.19			-.014	-.079	-.145	-.159		.025	-.023	-.041	-.038	-.109
.20	.022						.033					
.25	.014	-.007	-.043	-.073	-.092	-.162	.035	.010	-.031	-.041	-.078	-.114
.30	.014	-.011	-.043	-.082	-.116	-.162	.026	.002	-.040	-.060	-.084	-.093
.35	.007	-.030	-.061	-.088	-.128	-.171	.027	-.028	-.054	-.080	-.103	-.138
.40	.021	-.035	-.080	-.103	-.128	-.174	.001	-.021	-.059	-.097	-.115	-.165
.45	-.042	-.040	-.093	-.110	-.116	-.189	-.035	-.043	-.076	-.108	-.118	-.172
.50	-.029	-.053	-.108	-.119	-.159	-.216	-.024	-.045	-.093	-.127	-.124	-.189
.55	-.036	-.075	-.118	-.131	-.175		-.044	-.053	-.101	-.121	-.144	
.60	-.055	-.084	-.100	-.113	-.175	-.245	-.038	-.061	-.099	-.118	-.154	-.182
.65	-.071	-.100	-.120	-.119	-.166		-.058	-.079	-.112	-.119	-.149	
.68												-.135
.70	-.073	-.097	-.128	-.160	-.188	-.233	-.071	-.084	-.112	-.134	-.175	-.189
.73												-.155
.75	-.086	-.109	-.113		-.168		-.071	-.095	-.114			-.167
.77												
.79												
.80	-.104	-.104					-.088	-.091				
.81												
.82												
.83												
.85	-.093	-.108	-.124	-.135			-.085	-.104	-.140			
.87	-.098	-.124	-.136		-.151		-.101	-.135	-.143			
.88												
.89												
.90	-.130		-.111		-.140	-.205	-.180		-.126		-.150	-.212
.91	-.098		-.116				-.121					
.93	-.100		-.100				-.139					
.94	-.104		-.090		-.120		-.134					
.96												
.97												
.98	-.087		-.076		-.103							

 $\alpha = 2.2$

a = 2.2												
.00	.098	.191	-.005	-.040	-.496		.127	.152	.188	.201	.184	.064
.01	.015	-.157	-.284	-.498	-.406	-.618	.127	.125	.166	.151	.142	.143
.03	-.014	-.138	-.208	-.332	-.379	-.533	.125	.115	.123	.109	.111	.109
.05	-.036	-.063	-.144	-.218	-.332	-.509	.125	.103	.099	.086	.087	.059
.08	-.048	-.050	-.132	-.188	-.294	-.491	.125	.103	.086	.074	.064	.042
.10	-.032	-.050	-.111	-.157	-.261	-.479	.098	.093	.086	.074	.064	.009
.15	-.031	-.048	-.098	-.166	-.230	-.471	.097	.086	.076	.049	.040	.009
.19			-.098	-.183	-.250	-.474		.068	.043	.017	.054	-.001
.20	-.023						.088					
.25	-.032	-.058	-.123	-.164	-.212	-.481		.088	-.048	-.028	-.004	-.003
.30	-.032	-.043	-.113	-.164	-.220	-.415		.069	-.035	-.021	-.017	-.014
.35	-.036	-.081	-.122	-.168	-.224	-.463		.068	-.015	-.004	-.039	-.030
.40	-.018	-.091	-.142	-.180	-.217	-.438		.039	-.008	-.016	-.052	-.049
.45	-.085	-.095	-.153	-.188	-.207	-.321		.003	-.013	-.032	-.066	-.095
.50	-.078	-.108	-.161	-.193	-.232	-.317		.004	-.016	-.050	-.088	-.061
.55	-.083	-.125	-.171	-.204	-.245			.011	-.032	-.055	-.083	-.082
.60	-.103	-.133	-.152	-.188	-.250	-.332		.011	-.040	-.055	-.078	-.096
.65	-.114	-.148	-.169	-.192	-.231			.028	-.057	-.067	-.085	-.104
.68												
.70	-.118	-.146	-.177	-.228	-.245	-.316		.043	-.062	-.067	-.101	-.139
.73												
.75	-.128	-.155	-.168		-.228			.042	-.072	-.074		-.129
.77												
.79												
.80	-.148	-.154			-.218	-.284		.058	-.067			
.81												
.82												
.85												
.88	-.136	-.152	-.167		-.210			.058	-.089	-.101		
.87									.091	-.098	-.119	-.110
.88										.093		
.89										-.098		
.90	-.173		-.193		-.202	-.233		.132		-.097	-.102	-.189
.91			-.141		-.158					-.106	-.109	
.93			-.141		-.138					-.123	-.102	
.95	-.139	-.132			-.161					-.112	-.127	-.092
.96												
.97			-.099		-.142					-.085	-.091	
.98			-.115							-.123		

TABLE I.- Continued
PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 60° DELTA WING

X = 1.00

443

Upper surface							Lower surface						
Stations, fraction of semispan							Stations, fraction of semispan						
x/c	0.19	0.32	0.53	0.65	0.74	0.80	0.19	0.32	0.53	0.65	0.74	0.90	
.00	+.112	=.147	=.505	=.474	=.602		+.199	=.266	=.288	=.279	=.247	=.125	
.01	=.032	=.370	=.528	=.564	=.609	=.640	+.213	=.225	=.259	=.245	=.227	=.205	
.03	=.067	=.322	=.529	=.615	=.614	=.616	+.219	=.204	=.222	=.221	=.207	=.193	
.05	=.089	=.195	=.525	=.519	=.618	=.615	+.207	=.197	=.191	=.185	=.179	=.160	
.08	=.111	=.169	=.533	=.427	=.422	=.618	+.184	=.186	=.172	=.168	=.147	=.139	
.10	=.075	=.156	=.506	=.393	=.645	=.617	+.172	=.153	=.139	=.127	=.121	=.100	
.15	=.086	=.138	=.375	=.377	=.563	=.616	+.128	=.110	=.096	=.118	=.096		
.19	=.129	=.226	=.366	=.603	=.612		+.154						
.20	=.069						+.149	=.105	=.086	=.080	=.073	=.064	
.25	=.092	=.129	=.281	=.351	=.550	=.607	+.127	=.090	=.074	=.056	=.058	=.057	
.30	=.078	=.124	=.158	=.346	=.452	=.603	+.118	=.071	=.049	=.053	=.031	=.045	
.35	=.085	=.136	=.162	=.341	=.349	=.609	+.094	=.061	=.035	=.015	=.020	=.039	
.40	=.059	=.152	=.191	=.340	=.265	=.623	+.055	=.036	=.018	=.002	=.010	=.048	
.45	=.136	=.152	=.207	=.351	=.256	=.623	+.059	=.031	=.002	=.017	=.004	=.061	
.50	=.122	=.161	=.226	=.316	=.297	=.624	+.039	=.014	=.007	=.010	=.018		
.55	=.128	=.178	=.232	=.304	=.274		+.039	=.008	=.007	=.012	=.041	=.067	
.60	=.147	=.185	=.201	=.279	=.438	=.617	+.022	=.010	=.013	=.029	=.082		
.65	=.157	=.205	=.223	=.246	=.229		+.022						
.68					=.250		+.087						
.70	=.162	=.201	=.236	=.306	=.245	=.587	+.004	=.010	=.025	=.046	=.129	=.080	
.73					=.238		+.084						
.75	=.179	=.208	=.226		=.240		+.004	=.024	=.040		=.122		
.77					=.284					=.108			
.80					=.262					=.108			
.81	=.197	=.202			=.234	=.510	+.012	=.022			=.093	=.104	
.82					=.264					=.108			
.85					=.209					=.098			
.86					=.197	=.253				=.100			
.87					=.211					=.074			
.88					=.178	=.213	=.227			=.093			
.89					=.174					=.079			
.90					=.167					=.081			
.91											=.084		
.93					=.176					=.088			
.95					=.173	=.183	=.207			=.100			
.96					=.175	=.169				=.089			
.97						=.150					=.066		
.99					=.146						=.110		

a = 6.3

TABLE L - Continued
PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 60° DELTA WING

 $M = 1.00$

$\alpha = 8 \cdot 3$						
Upper surface						Lower surface
x/c	0.19	0.32	0.53	0.65	0.74	0.90
.00	.143	-.674	-.689	-.715	-.670	
.01	-.109	-.683	-.588	-.705	-.670	-.719
.02	-.142	-.700	-.597	-.709	-.675	-.698
.03	-.179	-.726	-.692	-.763	-.724	-.773
.05	-.197	-.794	-.602	-.663	-.681	-.698
.08	-.218	-.818	-.629	-.640	-.683	-.704
.10	-.232	-.789	-.644	-.639	-.687	-.704
.13	-.187	-.692	-.692	-.692	-.701	-.703
.19	-.139	-.713	-.666	-.725	-.703	
.20						
.25	-.186	-.168	-.733	-.683	-.743	-.698
.30	-.171	-.179	-.726	-.692	-.764	-.695
.35	-.179	-.193	-.701	-.697	-.773	-.695
.40	-.146	-.218	-.680	-.705	-.784	-.702
.45	-.224	-.219	-.574	-.708	-.784	-.706
.50	-.212	-.227	-.469	-.684	-.787	-.703
.55	-.217	-.245	-.345	-.689	-.792	
.60	-.234	-.255	-.213	-.641	-.799	-.639
.65	-.240	-.269	-.213	-.627	-.772	
.68						
.70	-.244	-.266	-.242	-.641	-.776	-.577
.73						
.75	-.256	-.275	-.242	-.575		
.77						
.79						
.80	-.276	-.273				
.81						
.82						
.85	-.264	-.264	-.244	-.721		
.87	-.248	-.253	-.443			
.88						
.89						
.90	-.291	-.238				
.91	-.247	-.384				
.92	-.250	-.231				
.95	-.254	-.242				
.96						
.97						
.98	-.219	-.226				
$\alpha = 10 \cdot 3$						
.00	.168	-.878	-.748	-.758	-.757	
.01	-.156	-.747	-.719	-.753	-.768	-.808
.03	-.198	-.791	-.721	-.766	-.778	-.782
.05	-.237	-.858	-.722	-.761	-.782	-.779
.08	-.334	-.964	-.742	-.791	-.782	-.788
.10	-.266	-.1006	-.763	-.755	-.778	-.785
.15	-.247	-.724	-.796	-.782	-.790	-.786
.19	-.457	-.844	-.796	-.819	-.789	
.20						
.25	-.235	-.201	-.946	-.811	-.816	-.776
.30	-.224	-.213	-.930	-.828	-.831	-.768
.35	-.235	-.238	-.942	-.854	-.839	-.768
.40	-.206	-.264	-.968	-.868	-.862	-.787
.45	-.285	-.267	-.938	-.907	-.874	-.791
.50	-.269	-.279	-.851	-.890	-.895	-.797
.55	-.269	-.298	-.704	-.918	-.912	
.60	-.296	-.306	-.954	-.864	-.934	-.776
.65	-.303	-.321	-.408	-.998	-.927	
.68						
.70	-.304	-.319	-.324	-.917	-.955	-.650
.73						
.75	-.314	-.329	-.299			
.77						
.79						
.80	-.336	-.325				
.81						
.82						
.83						
.85						
.87	-.321	-.314	-.281	-.984		
.88	-.296	-.284	-.794			
.89						
.90	-.347	-.263	-.743	-.987	-.554	
.91	-.293	-.295	-.260			
.93						
.95	-.304	-.287				
.96						
.97						
.98	-.264	-.267				

TABLE L- Continued
PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 60° DELTA WING

X = 1.00

$\alpha = 12.3$

$\alpha = 14.4$

TABLE I.- Continued
PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 60° DELTA WING

X = 1.00

$\alpha = 17.8$

$\alpha = 20.0$

TABLE I- Continued

W = 1.93

TABLE I.- Continued
PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 60° DELTA WING

 $M = 1.05$ $\alpha = 4+2$

Upper surface						Lower surface						
Stations, fraction of semispan						Stations, fraction of semispan						
x/c	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90
.00	.074	-.083	-.444	-.391	-.653		.155	.186	.229	.235	.208	.102
.01	-.059	-.399	-.533	-.590	-.621	-.647	.155	.152	.202	.196	.187	.185
.03	-.092	-.314	-.512	-.610	-.621	-.621	.145	.132	.158	.173	.165	.170
.05	-.121	-.223	-.492	-.509	-.621	-.616	.145	.119	.127	.133	.138	.134
.08	-.142	-.191	-.510	-.619	-.624	-.616	.106	.117	.107	.122	.110	.114
.10	-.137	-.186	-.479	-.381	-.628	-.614	.096	.091	.081	.085	.079	.076
.15	-.134	-.174	-.268	-.374	-.630	-.610	.071	.057	.055	.079	.076	
.19			-.450	-.362	-.609							
.20												
.25	-.140	-.178	-.221	-.345	-.588	-.601	.077	.050	.034	.045	.033	.045
.30	-.140	-.175	-.208	-.341	-.520	-.592	.062	.039	.025	.021	.021	.055
.35	-.147	-.174	-.213	-.333	-.421	-.586	.050	.008	-.002	.000		.023
.40	-.114	-.192	-.229	-.332	-.332	-.589	.036	.010	-.002	.016		.040
.45	-.189	-.196	-.237	-.224	-.260	-.593	.014	-.010	-.005	-.017		.045
.50	-.176	-.198	-.249	-.310	-.253	-.590	.004	-.014	-.039	-.043	-.021	.054
.55	-.178	-.210	-.259	-.303	-.270		.020	.020	.043	-.040	-.043	
.60	-.189	-.212	-.226	-.280	-.288	-.581	.008	-.026	-.039	-.035	-.059	-.056
.65	-.197	-.225	-.242	-.264	-.262		.026	.043	-.047	-.040	-.085	
.68												
.70	-.191	-.219	-.249	-.313	-.279	-.559	.035	-.046	-.047	-.057	-.121	-.065
.73												
.75	-.200	-.225	-.235		-.259		.037	-.051	-.052			
.77												
.79												
.80	-.215	-.217										
.81												
.82												
.83												
.85	-.201	-.212	-.229									
.87												
.88												
.89												
.90	-.235	-.214	-.242		-.240	-.396	.080					
.91	-.193											
.92	-.196	-.206										
.93	-.187											
.94												
.95												
.96												
.97												
.98												

 $\alpha = 6+3$

Upper surface												Lower surface														
Stations, fraction of semispan												Stations, fraction of semispan														
x/c	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90		
.00	.066	-.412	-.696	-.728	-.668		.223	.266	.272	.258	.245	.137														
.01	-.123	-.617	-.594	-.647	-.652	-.692		.223	.239	.252	.245	.245														
.03	-.156	-.532	-.594	-.671	-.657	-.676		.226	.216	.225	.234	.238														
.05	-.183	-.523	-.583	-.653	-.660	-.676		.226	.208	.200	.203	.217														
.08	-.207	-.495	-.592	-.629	-.662	-.680		.189	.189	.180	.191	.195														
.10	-.197	-.369	-.586	-.664	-.664	-.680		.173	.156	.149	.154	.171														
.15	-.194	-.196	-.580	-.616	-.679	-.680		.173	.156	.149	.154	.171														
.19			-.570	-.614	-.702	-.677		.151																		
.20																										
.25	-.191	-.209	-.550	-.605	-.709	-.674		.141	.112	.103	.114	.127														
.30	-.186	-.207	-.468	-.595	-.668	-.668		.123	.097	.095	.094	.115														
.35	-.191	-.211	-.379	-.566	-.679	-.668		.117	.077	.076	.076	.100														
.40	-.157	-.229	-.331	-.543	-.643	-.672		.091	.071	.071	.065	.085														
.45	-.157	-.225	-.295	-.509	-.674	-.673		.049	.052	.060	.057	.077														
.50	-.149	-.225	-.285	-.509	-.668	-.671		.049	.054	.050	.046	.052														
.55	-.214	-.247	-.279	-.435	-.655	-.637		.053	.059	.049	.046	.054														
.60	-.235	-.246	-.287	-.395	-.560	-.586		.043	.046	.042	.040	.044														
.65	-.221	-.251	-.282					.047	.040	.020																
.68	-.218																									
.70	-.230	-.251	-.271	-.366	-.560	-.586		.043	.046	.042	.040	.044														
.73	-.237	-.257	-.257	-.395	-.558	-.530		.047	.040	.020																
.75	-.237	-.257	-.257	-.395	-.558	-.530		.047	.040	.020																
.77																										
.80	-.255	-.248	-.300	-.452	-.537	-.537		.044	.042																	
.81																										
.82																										
.83																										
.85	-.235	-.237	-.287	-.395	-.558	-.530		.031	.014	.021																
.87	-.221	-.251	-.282	-.395	-.558	-.530		.012		.018	.009															
.88	-.218							.009																		
.89	-.207							.028		.025																
.90	-.267	-.232	-.237	-.337	-.498		.003																			
.91	-.214							.006																		
.93	-.216	-.218						.013		.004																
.95	-.222	-.207						.028		.025																
.96								.004																		
.97								.046																		
.98																										

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TABLE I. - Continued

$\mu = 1.03$

c = 8.2

$$\alpha = 10^{-3}$$

TABLE I. - Continued

$K = 1.03$

a = 12.4

$$a = 14.4$$

TABLE L- Continued
PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 60° DELTA WING

X = 1.05

g = 2±2

TABLE I.- Continued

$$M = 1.05$$

a = 4.2

Upper surface						Lower surface						
Stations, fraction of semispans						Stations, fraction of semispans						
x/c	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90
.00	.102	-0.65	-0.411	-0.363	-0.629		.181	.192	.233	.239	.212	.102
.01	-0.031	-0.383	-0.510	-0.583	-0.597	-0.641	.181	.158	.213	.199	.190	.184
.03	-0.067	-0.300	-0.485	-0.600	-0.601	-0.612	.169	.139	.167	.177	.165	.167
.05	-0.097	-0.212	-0.461	-0.516	-0.598	-0.607	.169	.124	.138	.140	.141	.131
.08	-0.122	-0.185	-0.474	-0.429	-0.598	-0.607	.119	.121	.114	.127	.113	.111
.10	-0.122	-0.178	-0.447	-0.390	-0.598	-0.603	.106	.091	.077	.087	.082	.074
.15	-0.128	-0.167	-0.367	-0.378	-0.602	-0.604	.074	.057	.059	.082	.062	.049
.19			-0.264	-0.367	-0.620	-0.600						
.20							.088					
.25	-0.125	-0.171	-0.229	-0.361	-0.563	-0.594	.084	.055	.040	.046	.034	.038
.30	-0.132	-0.173	-0.212	-0.355	-0.511	-0.586	.064	.046	.030	.029	.020	.053
.35	-0.146	-0.168	-0.216	-0.341	-0.486	-0.577	.066	.008	.006	.002	.003	.020
.40	-0.111	-0.187	-0.227	-0.332	-0.370	-0.583	.042	.006	.002	.012	.017	.046
.45	-0.183	-0.192	-0.235	-0.318	-0.296	-0.582	.010	-0.005	-0.11	.023	.020	.051
.50	-0.172	-0.194	-0.247	-0.302	-0.281	-0.582	.012	-0.012	-0.31	.002	.028	.060
.55	-0.172	-0.208	-0.253	-0.301	-0.274		.018	-0.020	-0.040	-0.049		
.60	-0.187	-0.212	-0.224	-0.274	-0.294	-0.573	.007	-0.024	-0.036	-0.040	-0.064	.068
.65	-0.196	-0.224	-0.240	-0.256	-0.265		.023	-0.042	-0.048	-0.044	-0.053	
.68				-0.265							-0.076	
.70	-0.188	-0.216	-0.249	-0.307	-0.283	-0.558	.033	-0.045	-0.050	-0.063	-0.111	.082
.73					-0.273						-0.094	
.75	-0.199	-0.224	-0.238		-0.268		.032	-0.054	-0.056		-0.104	-0.102
.77					-0.312						-0.097	
.79					-0.267						-0.094	
.80	-0.218	-0.218		-0.264	-0.517		.047	-0.048			-0.094	-0.098
.81				-0.276							-0.090	
.82				-0.238							-0.097	
.83				-0.225	-0.268						-0.091	-0.088
.85	-0.202	-0.217	-0.236		-0.251		.047	-0.068	-0.086		-0.087	
.87				-0.205	-0.238	-0.274					-0.065	
.88				-0.200							-0.065	
.89				-0.191							-0.068	
.90	-0.250		-0.224		-0.251	-0.442	.087		-0.079		-0.083	-0.133
.91		-0.198		-0.264					-0.073	1.305		
.93		-0.203	-0.216						-0.092	-0.081		
.95	-0.208	-0.198			-0.245		.087		-0.103		-0.084	
.96				-0.261						1.307		
.97			-0.216							-0.090		
.98		-0.197								-0.115		

$$a = 6.3$$

TABLE I.- Continued

PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 60° DELTA WING

 $M = 1.05$ $\alpha = 8^\circ 1$

x/c	Upper surface						Lower surface					
	0.19	0.32	0.53	0.65	0.74	0.80	0.19	0.32	0.53	0.65	0.74	0.80
.00	+.069	-+.751	-+.724	-+.724	-+.694	-+.727	+.296	+.326	+.300	+.284	+.247	+.098
.01	-+.171	-+.717	-+.633	-+.690	-+.694	-+.727	+.296	+.304	+.304	+.261	+.261	+.194
.03	-+.211	-+.688	-+.636	-+.704	-+.697	-+.710	+.310	+.284	+.276	+.290	+.261	+.226
.05	-+.236	-+.719	-+.634	-+.687	-+.701	-+.709	+.310	+.265	+.253	+.260	+.245	+.221
.08	-+.265	-+.739	-+.652	-+.666	-+.701	-+.715	+.265	+.244	+.234	+.249	+.223	+.209
.10	-+.253	-+.718	-+.659	-+.669	-+.701	-+.714	+.240	+.211	+.200	+.210	+.196	+.179
.15	-+.240	-+.171	-+.687	-+.675	-+.713	-+.714	+.189	+.173	+.181	+.198	+.179	
.19			-+.704	-+.680	-+.732	-+.714						
.20	-+.218						+.212					
.25	-+.230	r+.228	-+.720	-+.684	-+.740	-+.707	+.204	+.161	+.149	+.162	+.147	+.158
.30	-+.221	-+.235	-+.708	-+.689	-+.763	-+.704	+.175	+.145	+.138	+.141	+.131	+.124
.35	-+.232	-+.249	-+.680	-+.686	-+.769	-+.703	+.164	+.125	+.113	+.114	+.114	+.061
.40	-+.192	-+.262	-+.642	-+.693	-+.778	-+.709	+.136	+.114	+.101	+.101	+.098	+.045
.45	-+.264	-+.271	-+.577	-+.691	-+.774	-+.715	+.097	+.094	+.085	+.088	+.089	+.038
.50	-+.261	-+.269	-+.478	-+.659	-+.780	-+.716	+.104	+.090	+.072	+.069	+.082	+.028
.55	-+.254	-+.284	-+.372	-+.666	-+.780		+.084	+.074	+.061	+.074	+.058	
.60	-+.269	-+.291	-+.424	-+.612	-+.786	-+.686	+.086	+.068	+.061	+.070	+.050	+.035
.65	-+.277	-+.300	-+.424	-+.612	-+.785		+.070	+.051	+.034	+.056	-.022	
.68												
.70	-+.270	-+.293	-+.268	-+.629	-+.768	-+.588	+.054	+.048	+.047	+.029	-.058	+.026
.73												
.75	-+.276	-+.301	-+.270				+.051	+.039	+.028			
.77												
.79												
.80	-+.295	-+.294					+.044	+.039				
.81												
.82												
.83												
.85	-+.279	-+.289	-+.248	-+.501			+.035	+.011	+.003			
.87	-+.271	-+.274	-+.448				+.015	+.023	+.016			
.88	-+.269						+.010					
.89	-+.261											
.90	-+.317	-+.266	-+.400	-+.692	-+.512		-.003	-+.010	-+.020		+.004	-.047
.91	-+.267						-.017	-+.021				
.93	-+.272	-+.257					-.023	-.029				
.95	-+.281	-+.268										
.96												
.97												
.98	-+.257							-.053				

 $\alpha = 10^\circ 3$

x/c	0.05	0.19	0.32	0.53	0.65	0.74	0.80	0.19	0.32	0.53	0.65	0.74	
.00	+.065	-+.753	-+.740	-+.752	-+.740			+.294	+.393	+.328	+.280	+.249	+.125
.01	-+.210	-+.791	-+.715	-+.732	-+.744	-+.765		+.326	+.393	+.328	+.280	+.249	
.03	-+.237	-+.816	-+.715	-+.760	-+.752	-+.748		+.384	+.381	+.339	+.314	+.285	+.194
.05	-+.307	-+.896	-+.715	-+.742	-+.756	-+.744		+.428	+.366	+.332	+.323	+.306	+.247
.08	-+.330	-+.938	-+.740	-+.728	-+.731	-+.749		+.415	+.336	+.311	+.304	+.297	+.236
.10	-+.310	-+.946	-+.759	-+.725	-+.720	-+.750		+.368	+.320	+.299	+.296	+.281	+.232
.15	-+.291	-+.709	-+.791	-+.743	-+.741	-+.753		+.330	+.285	+.268	+.261	+.264	+.226
.19	-+.455	-+.839	-+.758	-+.786	-+.755			+.257	+.238	+.238	+.262	+.233	
.20	-+.263							+.291					
.25	-+.267	-+.234	-+.916	-+.773	-+.780	-+.744		+.275	+.229	+.213	+.220	+.219	+.180
.30	-+.254	-+.250	-+.929	-+.786	-+.793	-+.736		+.243	+.211	+.199	+.196	+.205	+.155
.35	-+.265	-+.270	-+.912	-+.799	-+.808	-+.752		+.228	+.191	+.179	+.174	+.186	+.137
.40	-+.228	-+.285	-+.922	-+.811	-+.830	-+.744		+.201	+.179	+.165	+.158	+.166	+.132
.45	-+.200	-+.295	-+.905	-+.824	-+.844	-+.727		+.182	+.155	+.151	+.147	+.159	+.115
.50	-+.212	-+.501	-+.828	-+.839	-+.842	-+.781		+.162	+.134	+.128	+.122	+.121	+.113
.55	-+.284	-+.513	-+.689	-+.834	-+.859			+.142	+.137	+.130	+.131	+.121	
.60	-+.304	-+.521	-+.451	-+.780	-+.874	-+.755		+.147	+.133	+.120	+.125	+.080	+.128
.65	-+.308	-+.331	-+.408	-+.820	-+.867			+.134	+.117	+.119	+.107	+.007	
.68													
.70	-+.306	-+.324	-+.316	-+.843	-+.889	-+.624		+.119	+.119	+.105	+.068	-.007	+.101
.73								+.115	+.104	+.081			
.75	-+.310	-+.332	-+.298										
.77													
.79													
.80	-+.327	-+.327											
.81													
.82													
.85	-+.319	-+.322	-+.295	-+.771	-+.914			+.084	+.027	+.052		+.085	
.87	-+.303	-+.286	-+.741						+.035	+.061	+.072		
.88	-+.300								+.046				
.89	-+.292								+.055				
.90	-+.339	-+.267							+.055	+.069	+.063	+.068	+.008
.91	-+.295								+.057				
.93	-+.297	-+.264							+.054	+.060			
.95	-+.301	-+.292							+.026	+.040			
.96													
.97													
.98	-+.269								+.005	+.033	+.034		

TABLE L- Concluded

PRESSURE COEFFICIENTS ON A 3-PERCENT-THICK, 60° DELTA WING

 $M = 1.05$ $\alpha = 12^\circ 4$

Upper surface							Lower surface						
Stations, fraction of semispan							Stations, fraction of semispan						
x/c	0.10	0.32	0.53	0.65	0.74	0.90	0.10	0.32	0.53	0.65	0.74	0.90	
.00	.017	-.905	-.782	-.773	-.794	-.817	.381	.427	.352	.286	.245	.096	
.01	-.249	-.872	-.767	-.780	-.799	-.817	.453	.442	.377	.343	.302	.177	
.03	-.281	-.880	-.770	-.795	-.811	-.807	.535	.434	.389	.372	.345	.248	
.05	-.416	-.906	-.771	-.788	-.816	-.801	.504	.406	.376	.357	.341	.275	
.08	-.403	-.957	-.799	-.781	-.816	-.807	.462	.389	.364	.356	.330	.275	
.10	-.385	-.942	-.812	-.789	-.815	-.807	.412	.352	.338	.325	.320	.259	
.15	-.336	-.994	-.837	-.822	-.819	-.809	.323	.311	.300	.320	.259		
.19	-.372	-.864	-.844	-.811									
.20	-.292	F=320					.363						
.25	-.280	-.284	-.194	-.850	-.864	-.795	.340	.294	.287	.284	.279	.222	
.30	-.297	-.298	-.193	-.862	-.866	-.778	.312	.278	.273	.262	.262	.204	
.35	-.257	-.318	-.140	-.847	-.851	-.796	.293	.258	.252	.240	.240	.229	
.40	-.257	-.317	-.145	-.846	-.852	-.803	.270	.245	.240	.223	.220	.183	
.45	-.317	-.315	-.131	-.846	-.852	-.803	.257	.225	.220	.208	.208	.180	
.50	-.319	-.320	-.1002	-.899	-.878	-.811	.230	.218	.209	.188	.192	.172	
.55	-.316	-.344	-.054	-.928	-.897	-.811	.211	.202	.195	.184	.184		
.60	-.327	-.352	-.858	-.855	-.931	-.824	.210	.195	.191	.173	.120	.158	
.65	-.540	-.363	-.4744	-.900	-.957		.195	.175	.175	.150	.051		
.68											.117		
.70	-.340	-.355	-.622	-.954	-1.003	-.752	.178	.173	.155	.105	.097	.128	
.73											.153		
.75	-.342	-.362	-.486		-1.027		.169	.152	.121		.113		
.77													
.79													
.80	-.357	-.358					.156	.143					
.81													
.82													
.83													
.85	-.346	-.354	-.387				.128	.064	.106				
.87								.092	.104	.095			
.88								.099					
.89								.099					
.90	-.372	-.344					.103	.097	.096	.082	.085	.027	
.91	-.350												
.93	-.331	-.319											
.95	-.336	-.325											
.96													
.97													
.98	-.295												
a = 14° 4													
.00	-.100	-.986	-.817	-.812	-.844	-.856	.384	.461	.363	.280	.230	.061	
.01	-.257	-.930	-.814	-.814	-.848	-.856	.506	.501	.406	.362	.306	.149	
.03	-.340	-.929	-.820	-.827	-.862	-.850	.636	.503	.432	.406	.360	.242	
.05	-.473	-.923	-.823	-.831	-.867	-.846	.604	.480	.424	.402	.371	.282	
.08	-.465	-.975	-.842	-.824	-.867	-.848	.556	.464	.418	.403	.367	.290	
.10	-.424	-.1.047	-.842	-.838	-.865	-.848	.487	.426	.392	.379	.361	.281	
.15	-.367	-.1.115	-.889	-.859	-.868	-.852	.399	.369	.356	.363	.363	.284	
.19	-.843	-.938	-.886	-.887	-.887	-.856							
.20	-.315												
.25	-.315	-.612	-.962	-.917	-.880	-.851	.437						
.30	-.310	-.426	-.965	-.914	-.895	-.840	.410	.366	.344	.337	.321	.251	
.35	-.319	-.368	-.974	-.907	-.901	-.825	.376	.348	.326	.315	.309	.232	
.40	-.291	-.379	-.1.018	-.920	-.914	-.830	.356	.329	.306	.292	.286	.259	
.45	-.349	-.369	-.1.076	-.961	-.921	-.834	.295	.292	.277	.257	.250	.209	
.50	-.347	-.354	-.1.110	-.933	-.943	-.843	.290	.282	.256	.236	.236	.203	
.55	-.351	-.371	-.1.084	-.973	-.961	-.856	.272	.263	.241	.229	.204		
.60	-.358	-.382	-.1.035	-.895	-.965	-.857	.265	.255	.235	.213	.160	.187	
.65	-.365	-.398	-.976	-.986	-.946		.247	.233	.215	.188	.100		
.68											.158		
.70	-.365	-.391	-.911	-1.011	-.953	-.857	.227	.223	.192	.143	.132	.153	
.73											.174		
.75	-.374	-.397	-.794				.213	.196	.152				
.77													
.79													
.80	-.398	-.392					.194	.183					
.81													
.82													
.83													
.85	-.385	-.378	-.643	-.1.076									
.87													
.88													
.89													
.90	-.409	-.551											
.91													
.93													
.95	-.380	-.358	-.510										
.96													
.97													
.98	-.318	-.482	-.1.041										

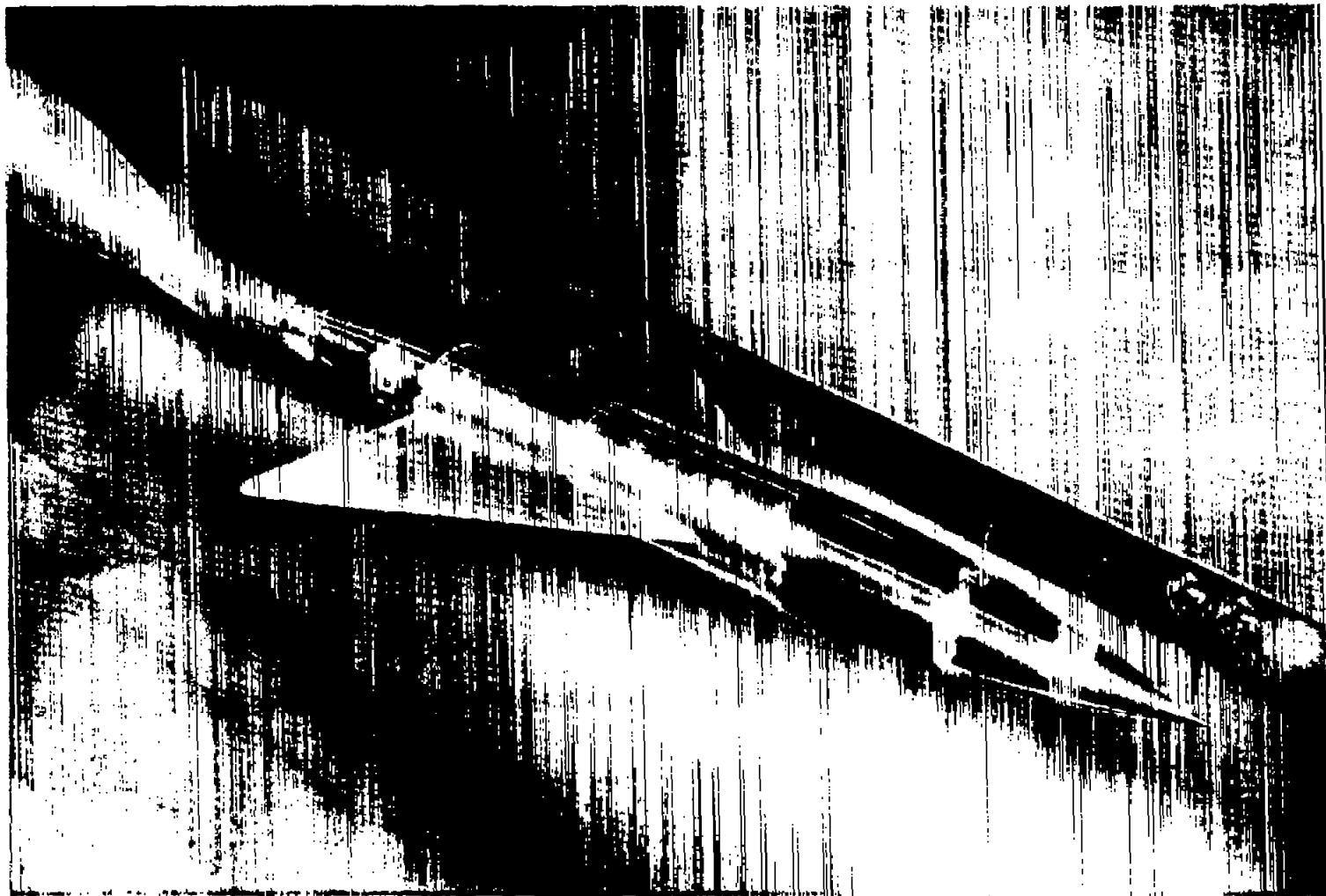
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TABLE II.- WING SECTION COEFFICIENTS

M	α , deg	$\frac{Y}{b/2} = 0.19$				$\frac{Y}{b/2} = 0.32$				$\frac{Y}{b/2} = 0.53$				$\frac{Y}{b/2} = 0.65$				$\frac{Y}{b/2} = 0.74$				$\frac{Y}{b/2} = 0.90$			
		c_n	$c_{\infty 0.35c}$	$\frac{x_{cp}}{c}$	c_n	$c_{\infty 0.35c}$	$\frac{x_{cp}}{c}$	c_n	$c_{\infty 0.35c}$	$\frac{x_{cp}}{c}$	c_n	$c_{\infty 0.35c}$	$\frac{x_{cp}}{c}$	c_n	$c_{\infty 0.35c}$	$\frac{x_{cp}}{c}$	c_n	$c_{\infty 0.35c}$	$\frac{x_{cp}}{c}$	c_n	$c_{\infty 0.35c}$	$\frac{x_{cp}}{c}$	c_n	$c_{\infty 0.35c}$	$\frac{x_{cp}}{c}$
.80	4.2	.0161	.001	0.344	0.172	0.011	0.288	0.230	0.024	0.248	0.269	0.018	0.282	0.316	0.040	0.224	0.482	-0.027	0.407	0.422	-0.012	.378			
	8.3	.358	.001	.346	.306	.028	.279	.537	.047	.262	.658	.005	.340	.842	-.005	.451	.422	-.012	.386						
	10.3	.451	-.003	.342	.518	.059	.275	.722	.047	.205	.894	-.035	.389	.995	-.129	.479	.465	-.017	.386						
	12.3	.512	.009	.333	.624	.053	.265	.883	.013	.395	.899	-.035	.512	.730	-.064	.437	.499	-.063	.477						
	14.4	.599	.010	.334	.745	.053	.279	1.113	-.070	.413	.883	-.065	.424	.733	-.077	.455	.544	-.044	.431						
	16.4	.718	-.001	.352	.918	.059	.308	1.130	-.062	.405	.976	-.089	.441	.733	-.077	.455	.544	-.044	.431						
	17.8	.785	.002	.347	1.003	.059	.311	1.147	-.072	.413	1.055	-.079	.426	.875	-.076	.437	.584	-.048	.431						
	19.8	.907	-.011	.362	1.186	.059	.334	1.177	-.080	.418	1.044	-.092	.438	.983	-.087	.445	.633	-.059	.443						
	25.9	1.231	-.053	.393	1.325	-.029	.372	1.295	-.101	.428	1.172	-.111	.445	1.064	-.111	.454	.732	-.083	.463						
.85	4.3	.167	.000	.350	.186	.010	.296	.241	.024	.250	.284	.018	.287	.328	.042	.222	.436	.008	.332						
	8.3	.351	-.002	.356	.402	.025	.268	.555	.043	.272	.694	-.002	.353	.861	-.089	.453	.445	-.014	.381						
	10.3	.521	.002	.346	.683	.044	.279	.913	-.004	.354	.879	-.055	.413	.708	-.073	.453	.542	-.045	.475						
	12.3	.570	-.021	.387	.678	.017	.324	.964	-.013	.364	1.156	-.127	.459	1.180	-.153	.480	.657	-.034	.437						
	14.4	.737	-.015	.370	.956	.025	.323	1.122	-.066	.409	.850	-.021	.375	.857	-.089	.454	.624	-.034	.450						
	19.8	.922	-.024	.376	1.192	.000	.350	1.167	-.088	.423	1.056	-.094	.439	.945	-.092	.448	.667	-.067	.471						
.90	4.2	.171	-.003	.366	.191	.006	.319	.242	.022	.259	.292	.016	.296	.342	.040	.234	.525	-.031	.409						
	8.3	.363	-.009	.375	.416	.036	.311	.564	.034	.289	.709	-.013	.368	.885	-.085	.446	.483	-.017	.386						
	10.3	.467	-.015	.382	.556	.020	.315	.768	.022	.322	.946	-.064	.417	1.066	-.129	.471	.572	-.025	.394						
	12.3	.570	-.021	.387	.678	.017	.324	.964	-.013	.364	1.156	-.127	.459	1.180	-.153	.480	.657	-.034	.402						
	14.4	.651	-.018	.378	.780	.025	.318	1.127	-.056	.408	1.037	-.077	.424	.877	-.082	.443	.613	-.060	.446						
	16.4	.747	-.029	.388	.915	.009	.340	1.153	-.077	.417	1.084	-.091	.439	.880	-.100	.464	.681	-.069	.446						
.92	4.2	.177	-.004	.373	.201	.003	.325	.257	.021	.269	.303	.017	.294	.353	.040	.237	.541	-.032	.409						
	8.3	.379	-.016	.392	.434	.009	.329	.586	.026	.306	.734	-.024	.383	.894	-.087	.447	.558	-.016	.385						
	10.3	.597	-.034	.407	.709	.003	.343	1.008	-.034	.384	1.193	-.147	.473	1.167	-.146	.475	.701	-.042	.404						
	12.3	.703	-.047	.417	.849	-.004	.355	1.186	-.086	.423	1.277	-.158	.474	1.198	-.129	.458	.839	-.031	.411						
	14.3	.845	-.047	.406	1.054	-.015	.364	1.205	-.097	.430	1.069	-.110	.453	.992	-.104	.455	.734	-.077	.455						
	26.1	1.280	-.101	.429	1.437	-.090	.413	1.457	-.149	.452	1.353	-.150	.461	1.270	-.150	.468	.996	-.126	.477						
.94	.3	.010	.004	-.050	.009	.005	-.206	.012	.005	-.067	.016	.006	-.025	.026	.006	.119	.054	.006	.174						
	4.3	.182	-.008	.391	.206	.001	.344	.262	.017	.286	.314	.012	.312	.366	.037	.248	.549	-.024	.393						
	6.3	.287	-.016	.406	.321	.002	.345	.431	.014	.340	.512	.005	.340	.636	-.009	.354	.585	-.029	.400						
	8.3	.390	-.026	.418	.446	-.002	.354	.606	.010	.333	.756	-.041	.404	.892	-.092	.453	.629	-.027	.392						
	10.3	.493	-.036	.423	.584	-.003	.358	.813	-.013	.366	.987	-.101	.452	1.045	-.131	.475	.762	-.049	.414						
	12.3	.596	-.042	.421	.705	-.008	.362	1.009	-.053	.403	1.169	-.153	.481	1.155	-.153	.482	.877	-.059	.429						
	14.4	.698	-.054	.427	.828	-.013	.369	1.179	-.100	.435	1.272	-.172	.485	1.207	-.145	.470	.962	-.083	.436						
	17.8	.836	-.046	.405	1.040	-.013	.364	1.254	-.108	.436	1.146	-.111	.447	1.013	-.117	.463	.821	-.093	.463						
	20.0	.964	-.068	.421	1.192	-.045	.388	1.312	-.125	.446	1.198	-.134	.462	1.131	-.186	.462	.860	-.098	.464						
	22.1	1.076	-.080	.424	1.308	-.065	.400	1.359	-.139	.453	1.262	-.141	.468	1.188	-.135	.464	.923	-.111	.470						
	24.1	1.184	-.093	.429	1.388	-.083	.410	1.427	-.150	.455	1.334	-.153	.465	1.253	-.147	.467	.996	-.122	.472						
	26.2	1.287	-.108	.434	1.454	-.098	.418	1.510	-.164	.459	1.398	-.162	.466	1.318	-.157	.470	1.050	-.132	.476						

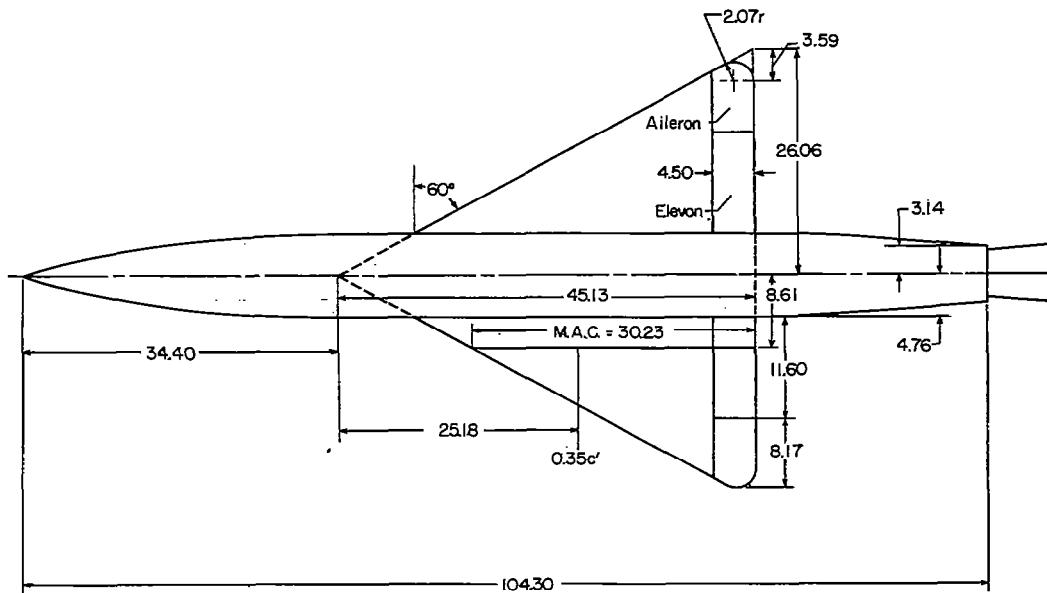
TABLE II.- WING SECTION COEFFICIENTS - Concluded

M	α , deg	$\frac{y}{b/2} = 0.19$			$\frac{y}{b/2} = 0.32$			$\frac{y}{b/2} = 0.53$			$\frac{y}{b/2} = 0.65$			$\frac{y}{b/2} = 0.74$			$\frac{y}{b/2} = 0.90$		
		c_n	c_m $0.35c$	x_{cp} c															
0.96	4.3	0.192	-0.015	0.428	0.216	-0.006	0.378	0.276	0.007	0.375	0.332	0.000	0.350	0.390	0.022	0.294	0.560	-0.028	0.400
	8.3	.393	-.032	.431	.406	-.008	.370	.614	-.004	.357	.755	-.053	.420	.879	-.097	.460	.708	-.052	.423
	12.3	.590	-.046	.428	.690	-.012	.367	1.031	-.091	.458	1.141	-.152	.483	1.141	-.158	.488	.905	-.086	.445
	17.8	.878	-.072	.432	1.071	-.039	.386	1.379	-.155	.462	1.350	-.172	.477	1.312	-.168	.478	1.102	-.129	.467
	20.0	.988	-.083	.434	1.208	-.058	.398	1.501	-.181	.471	1.456	-.191	.481	1.414	-.189	.484	1.200	-.156	.480
	26.4	1.277	-.112	.438	1.437	-.105	.423	1.527	-.173	.463	1.405	-.163	.466	1.341	-.163	.472	1.074	-.136	.477
.98	4.2	.188	-.016	.435	.208	-.007	.384			.315		-.004	.363	.387	-.012	.381	.556	-.035	.413
	8.3	.383	-.033	.436	.437	-.010	.373	.590	-.003	.355	.738	-.055	.425	.864	-.096	.461	.718	-.059	.432
	12.4	.582	-.047	.431	.687	-.014	.370	.974	-.055	.407	1.118	-.150	.484	1.124	-.158	.491	.898	-.089	.449
	14.4	.692	-.058	.434	.815	-.021	.376	1.142	-.099	.437	1.227	-.170	.489	1.186	-.151	.477	.997	-.106	.456
	17.8	.868	-.073	.434	1.064	-.041	.389	1.383	-.160	.466	1.364	-.180	.482	1.328	-.175	.482	1.133	-.145	.478
	22.2	1.088	-.098	.440	1.284	-.081	.413	1.477	-.177	.470	1.417	-.181	.478	1.385	-.179	.479	1.182	-.153	.479
1.00	4.3	.192	-.016	.432	.214	-.006	.379	.278	.005	.332	.539	-.003	.359	.368	.014	.312	.548	-.038	.419
	8.3	.372	-.031	.434	.425	-.009	.372	.573	-.003	.355	.712	-.054	.426	.829	-.092	.461	.704	-.060	.436
	10.3	.477	-.041	.435	.560	-.012	.371	.768	-.020	.376	.936	-.108	.464	.987	-.132	.484	.792	-.071	.440
	12.3	.547	-.049	.435	.677	-.016	.373	.962	-.057	.409	1.096	-.148	.485	1.111	-.158	.492	.894	-.091	.451
	14.4	.676	-.059	.437	.797	-.023	.379	1.119	-.100	.440	1.202	-.167	.489	1.165	-.149	.478	.988	-.107	.459
	17.8	.856	-.076	.439	1.038	-.043	.391	1.345	-.155	.466	1.329	-.176	.482	1.297	-.171	.482	1.113	-.151	.485
1.03	2.2	.094	-.008	.435	.100	-.004	.390	.128	-.002	.366	.151	-.001	.357	.193	-.002	.360	.283	-.002	.421
	4.2	.186	-.019	.452	.207	-.011	.403	.267	-.002	.357	.325	-.011	.384	.377	-.005	.337	.536	-.039	.423
	8.3	.388	-.038	.448	.437	-.018	.391	.589	-.008	.364	.729	-.056	.427	.862	-.098	.464	.726	-.062	.435
	10.4	.488	-.043	.438	.580	-.013	.372	.805	-.025	.381	.986	-.120	.472	1.024	-.142	.489	.811	-.074	.441
	12.3	.569	-.047	.433	.667	-.016	.374	.955	-.057	.410	1.090	-.146	.484	1.102	-.154	.490	.882	-.088	.450
	14.4	.658	-.054	.432	.800	-.019	.374	1.124	-.102	.441	1.196	-.168	.491	1.140	-.164	.476	.959	-.106	.461
1.05	2.2	.095	-.008	.431	.104	-.005	.399	.128	-.003	.370	.150	-.001	.359	.197	-.003	.365	.289	-.002	.357
	4.2	.185	-.018	.448	.205	-.011	.405	.267	-.003	.361	.325	-.009	.377	.374	-.005	.338	.531	-.042	.429
	8.2	.377	-.036	.446	.423	-.018	.392	.561	-.008	.363	.706	-.055	.428	.820	-.094	.465	.713	-.064	.440
	10.4	.477	-.045	.444	.559	-.017	.380	.773	-.028	.386	.926	-.112	.471	.969	-.133	.487	.802	-.079	.448
	12.4	.580	-.055	.445	.671	-.026	.381	.955	-.065	.418	1.080	-.148	.487	1.105	-.159	.494	.911	-.098	.457
	14.4	.668	-.064	.445	.802	-.030	.387	1.127	-.110	.448	1.190	-.171	.494	1.140	-.149	.480	.987	-.113	.465

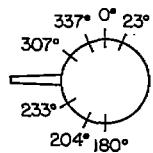


L-88122.1

Figure 1.- Model installed in test section of Langley 16-foot transonic tunnel.



Wing data		
	Actual	Theoretical
A	2.06	2.31
b/2	24.54	26.06
S	8.13 ft ²	8.17 ft ²
λ	0	0
c'	2.45	3.09 in.
Section	30.23 in. NACA 65A003	30.09 in. NACA 65A003



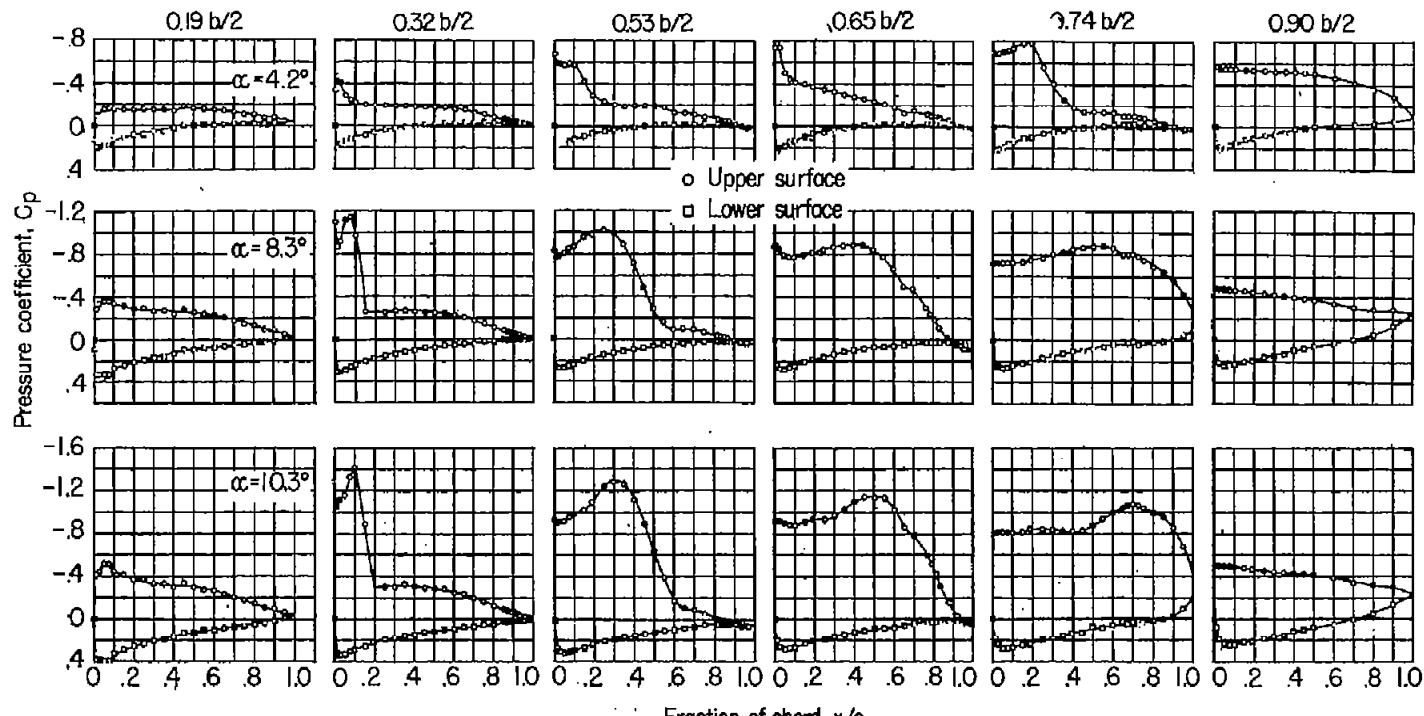
Body orifice meridians
looking upstream

Body ordinates	
x	r
0	0
2.00	.56
4.00	1.09
6.00	1.58
8.00	2.04
10.00	2.45
14.00	3.18
16.00	3.77
22.00	4.22
26.00	4.54
30.00	4.72
33.33	4.76
78.56	4.76
80.00	4.73
83.50	4.56
87.00	4.35
90.97	4.07
104.30	3.14

Wing orifice locations Station						
x/c	x/c	x/c	x/c	x/c	x/c	x/c
0.19 $\frac{b}{2}$	0.32 $\frac{b}{2}$	0.53 $\frac{b}{2}$	0.65 $\frac{b}{2}$	0.74 $\frac{b}{2}$	0.90 $\frac{b}{2}$	
0	0	0	0	0	0	
.01	.01	.01	.01	.01	.01	
.03	.03	.03	.03	.03	.03	
.05	.05	.05	.05	.05	.05	
.08	.08	.08	.08	.08	.08	
.10	.10	.10	.10	.10	.10	
.15	.15	.15	.15	.15	.15	
.20	.19	.19	.19	.19	.19	
.25	.25	.25	.25	.25	.25	
.30	.30	.30	.30	.30	.30	
.35	.35	.35	.35	.35	.35	
.40	.40	.40	.40	.40	.40	
.45	.45	.45	.45	.45	.45	
.50	.50	.50	.50	.50	.50	
.55	.55	.55	.55	.55	.55	
.60	.60	.60	.60	.60	.60	
.65	.65	.65	.65	.65	.65	
.70	.70	.70	.70	.70	.70	
.75	.75	.75	.75	.75	.75	
.80	.80	.82	.79	.75		
.85	.85	.83	.81	.80		
.90	.87	.85	.83	.85		
.95	.88	.87	.87	.90		
	.89	.90	.92	.95		
	.91	.93	.96			
	.93	.97				
	.95					
	.98					

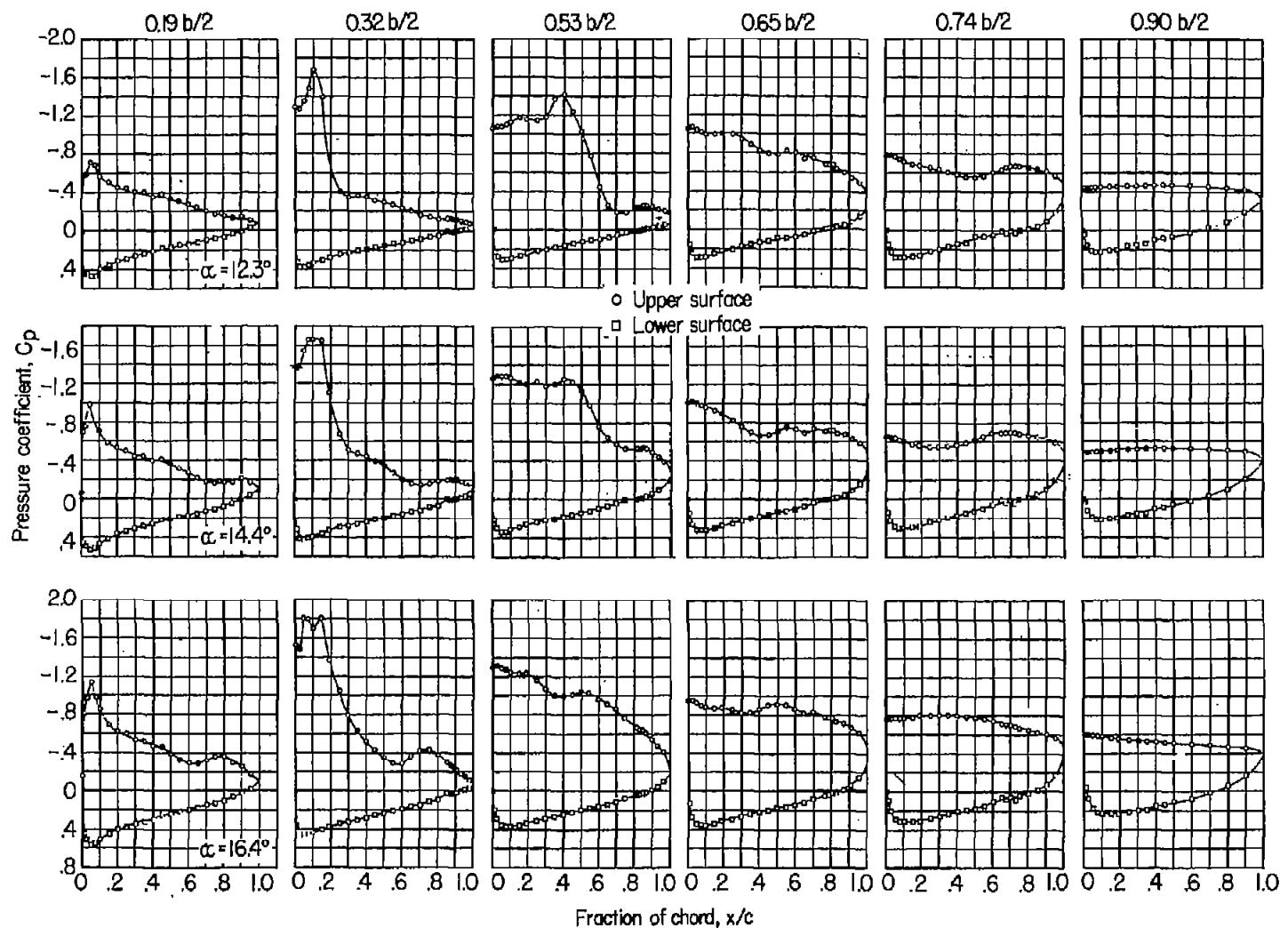
Body orifice locations Meridian						
x/l	x/l	x/l	x/l	x/l	x/l	x/l
0°	23°	180°	204°	233°	307°	337°
.02	.082	.002	.036	.036	.036	.036
.04	.06	.04	.40	.40	.40	.40
.06	.09	.08	.43	.43	.43	.43
.12	.14	.12	.47	.47	.47	.47
.20	.19	.20	.50	.50	.50	.50
.28		.28	.54	.54	.54	.54
.35		.35	.57	.57	.57	.57
.36		.36	.61	.61	.61	.61
.40		.40	.64	.64	.64	.64
.43		.43	.68	.68	.68	.68
.47		.47	.71	.71	.71	.71
.50		.50	.75	.75	.75	.75
.54		.54				
.57		.57				
.61						
.64						
.68						
.71						
.75						
.82						
.86						
.90						
.94						
.98						

Figure 2 - Model details and location of wing and body pressure orifices
All model dimensions are in inches unless otherwise noted.



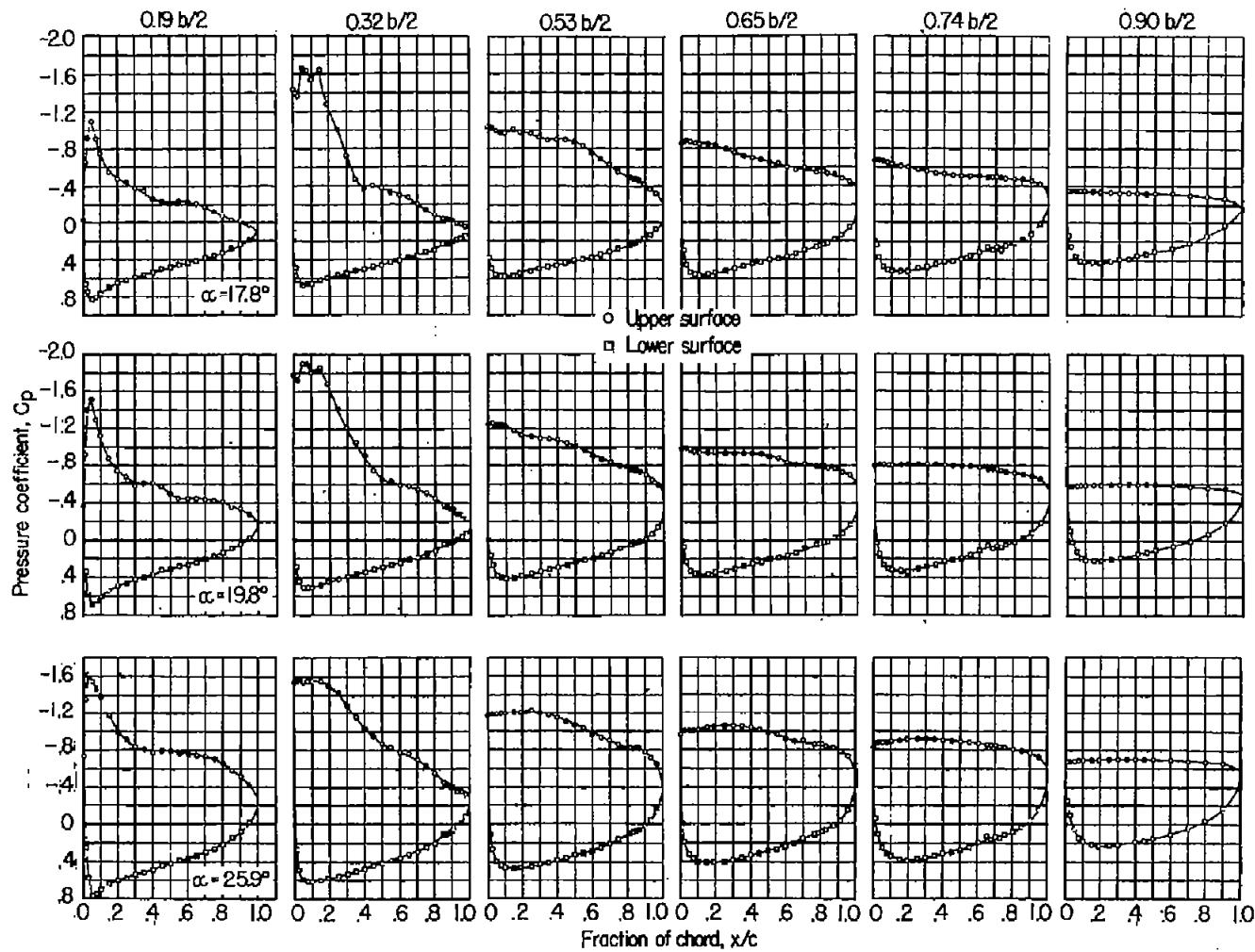
(a) $M = 0.80$; $C_{p,\text{sonic}} = -0.44$.

Figure 3.- Chordwise pressure distributions on a plane 60° delta wing.



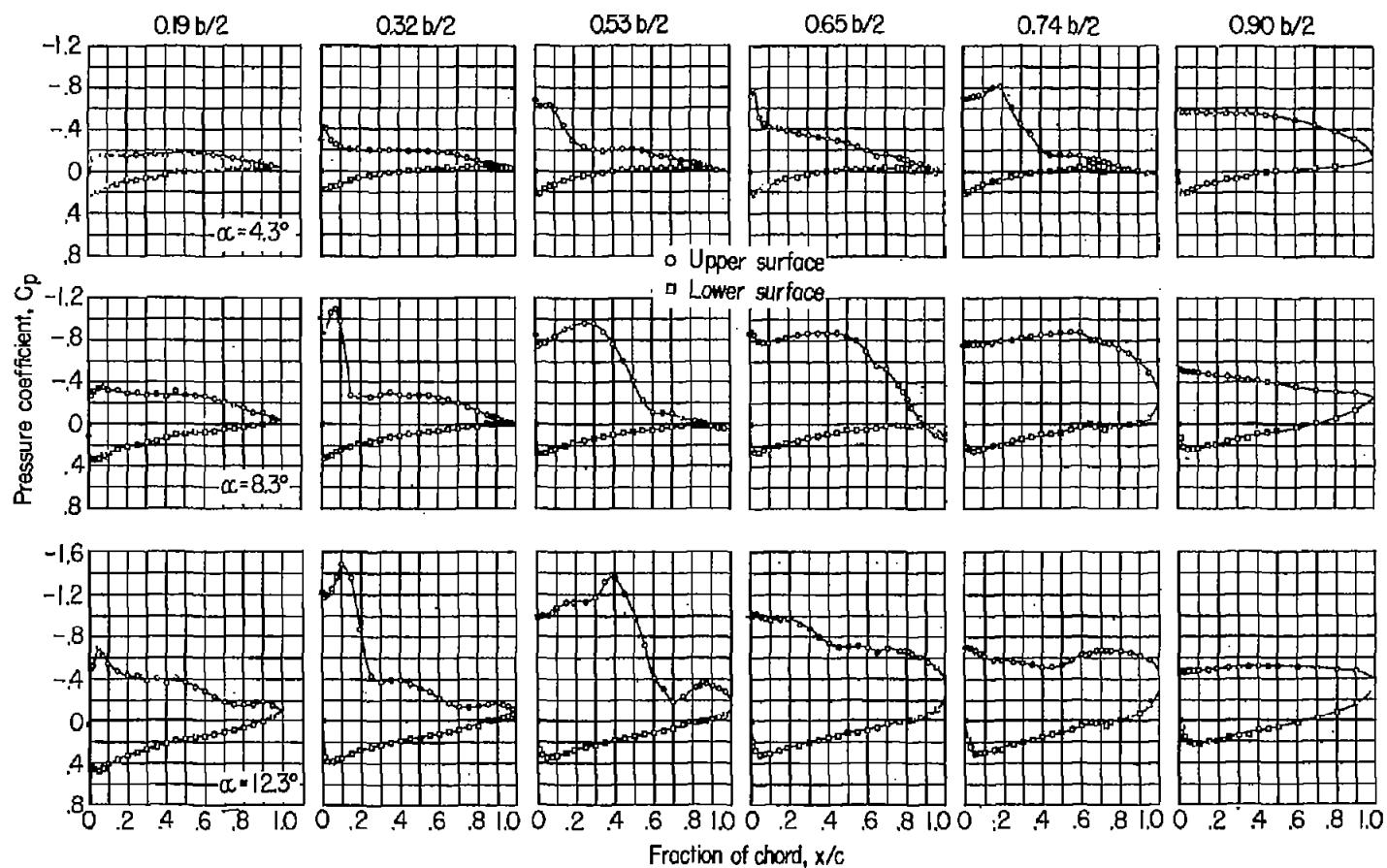
(a) Continued.

Figure 3.- Continued.



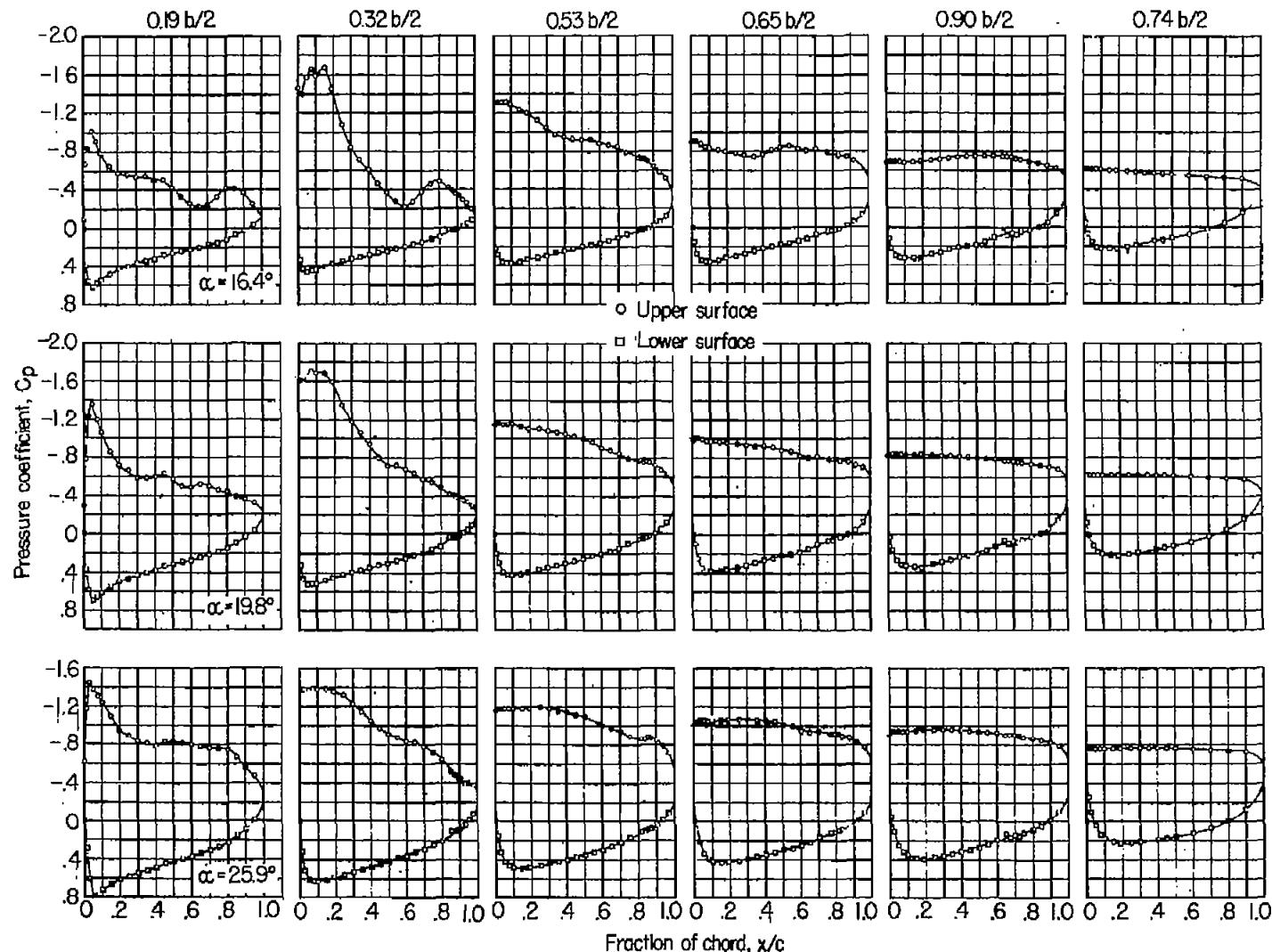
(a) Concluded.

Figure 3.- Continued.



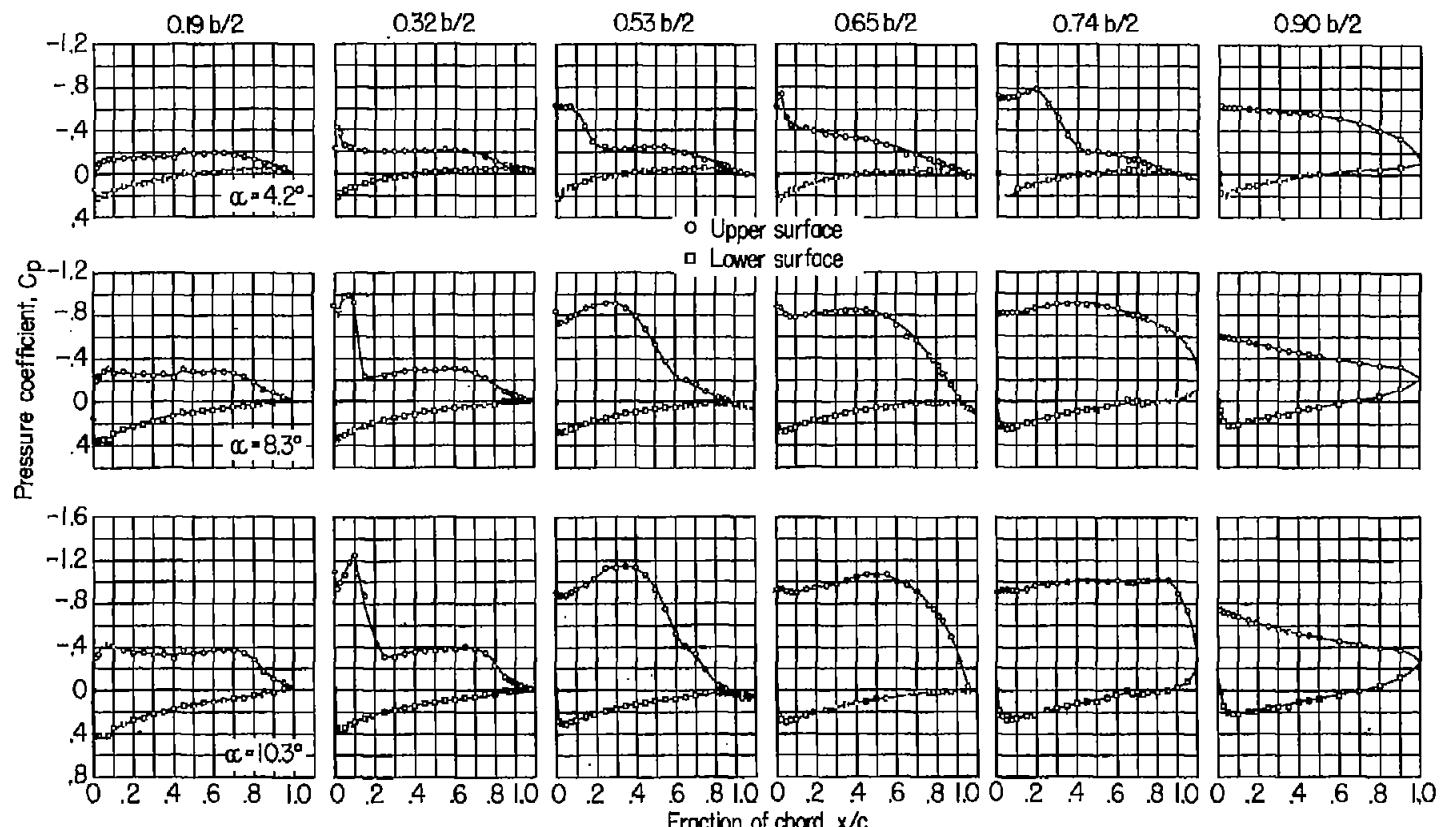
(b) $M = 0.85$; $C_{p,\text{sonic}} = -0.30$.

Figure 3.- Continued.



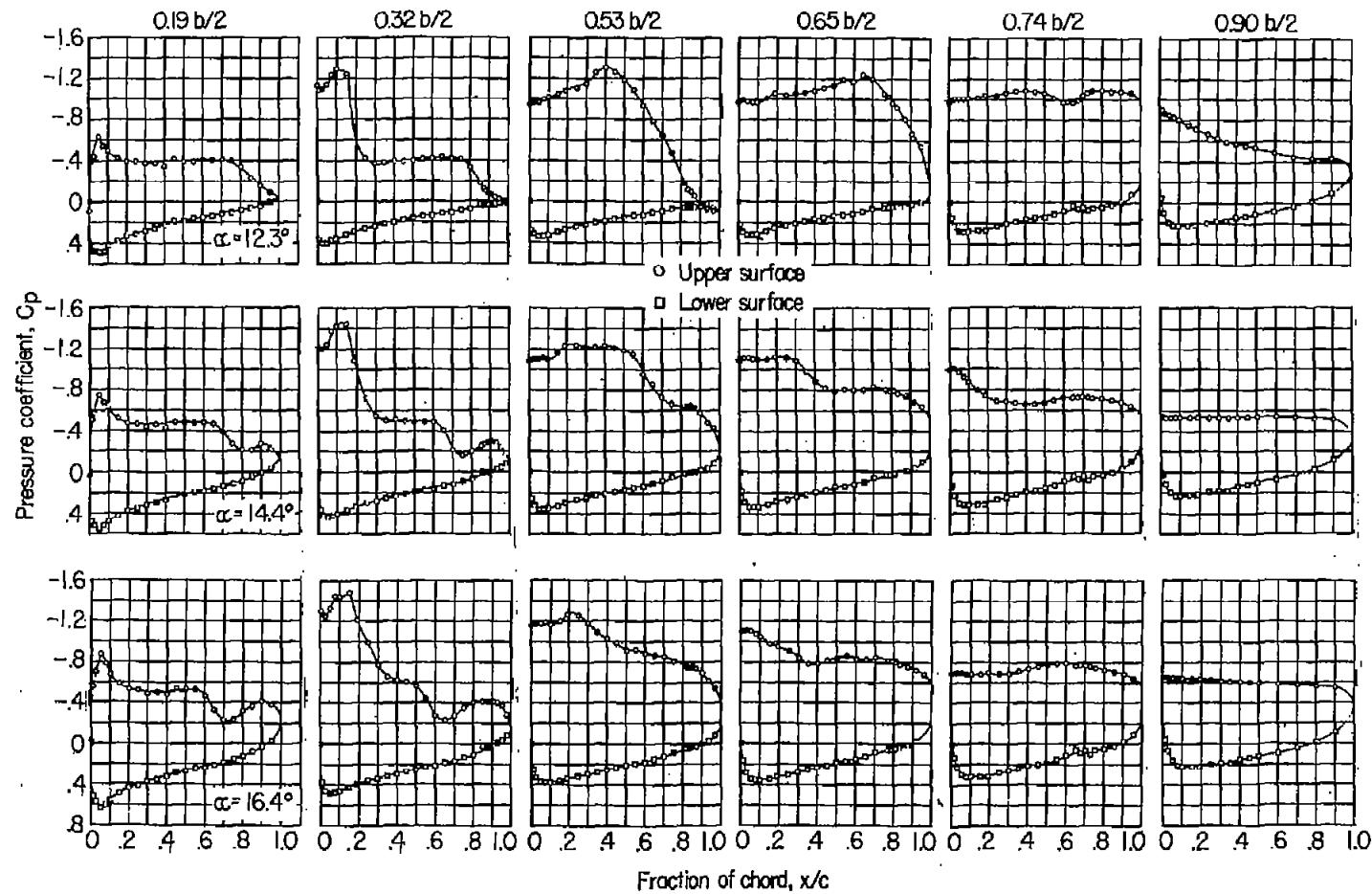
(b) Concluded.

Figure 3.- Continued.



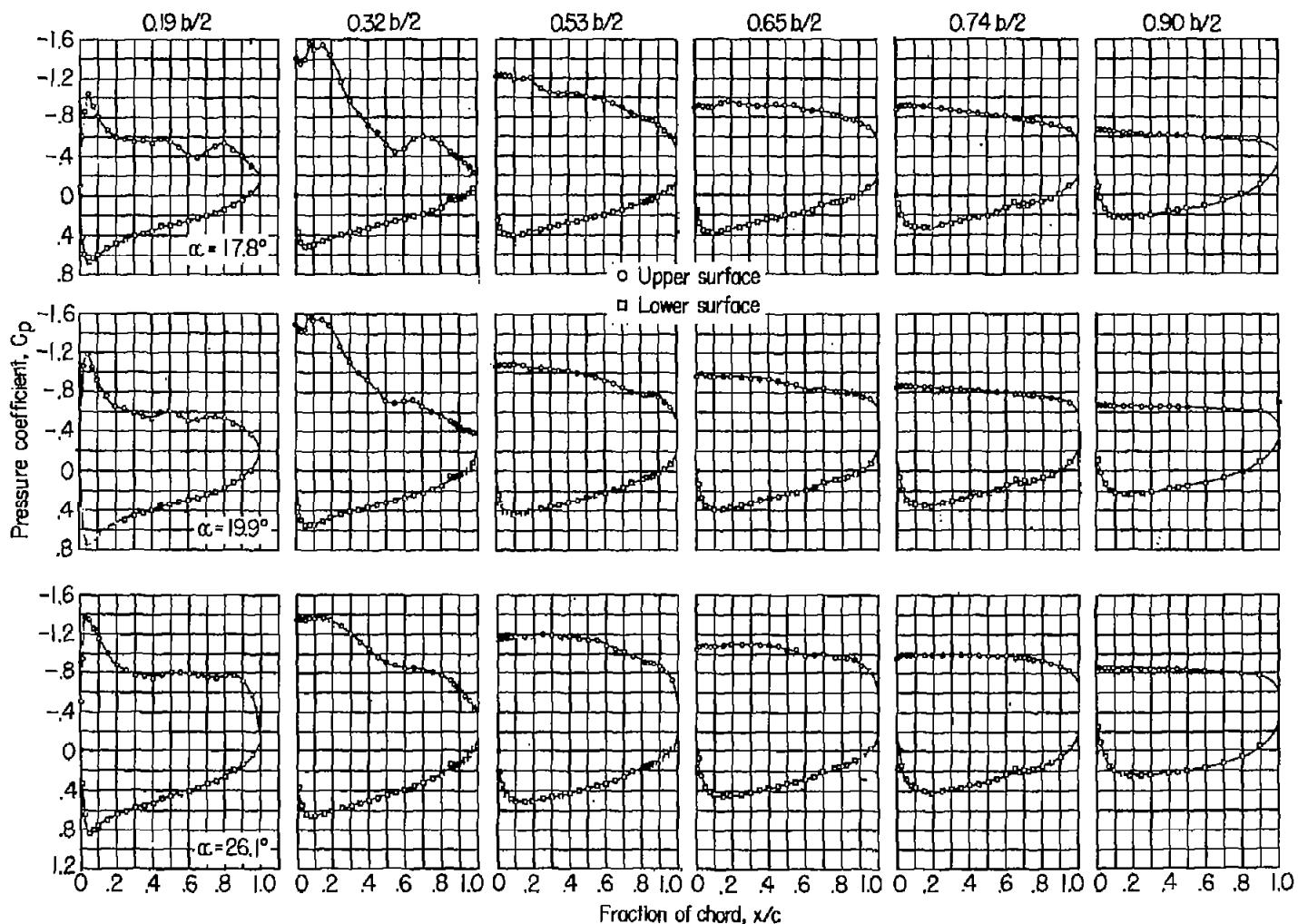
(c) $M = 0.90; C_{p,sonic} = -0.19.$

Figure 3.- Continued.



(c) Continued.

Figure 3.- Continued.



(c) Concluded.

Figure 3--Continued.

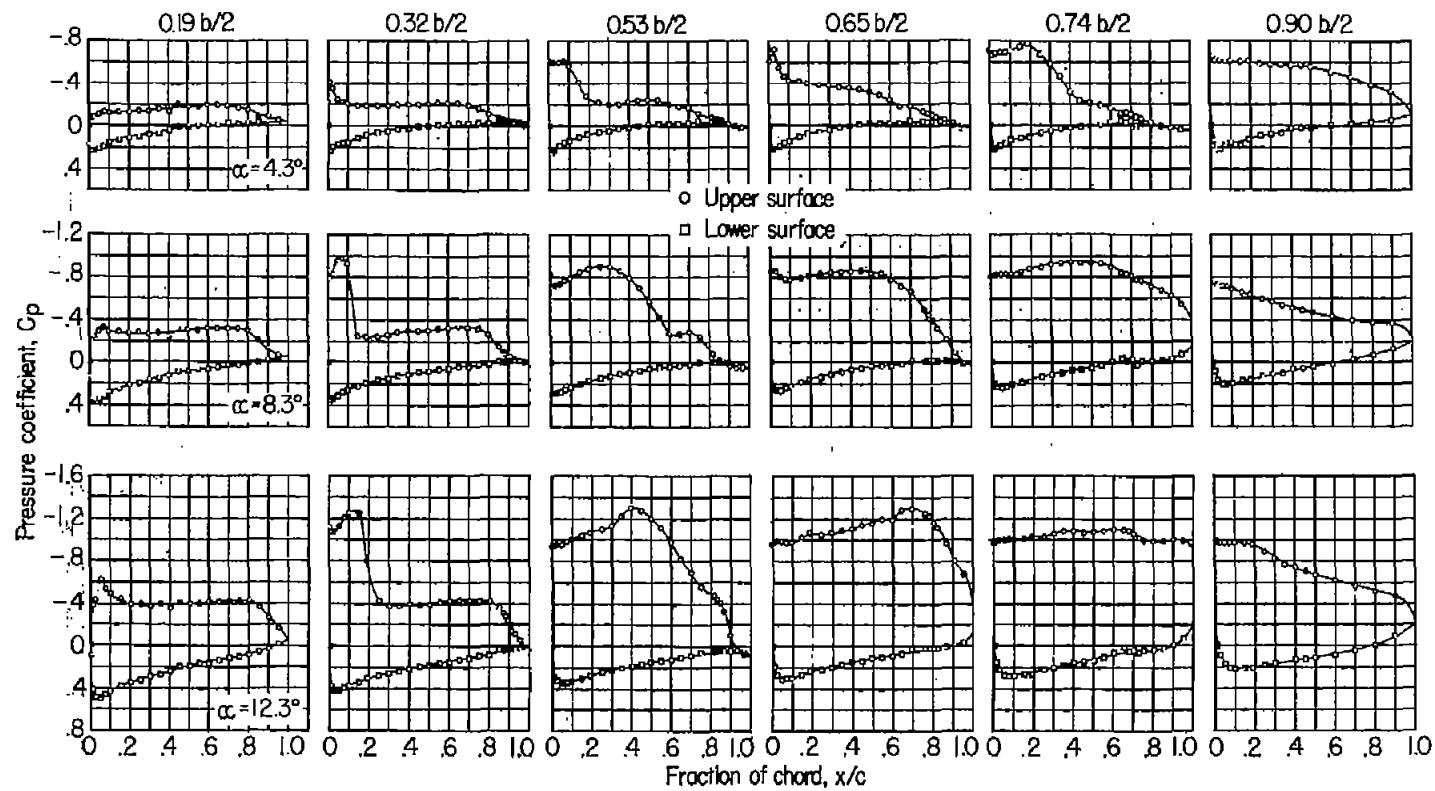
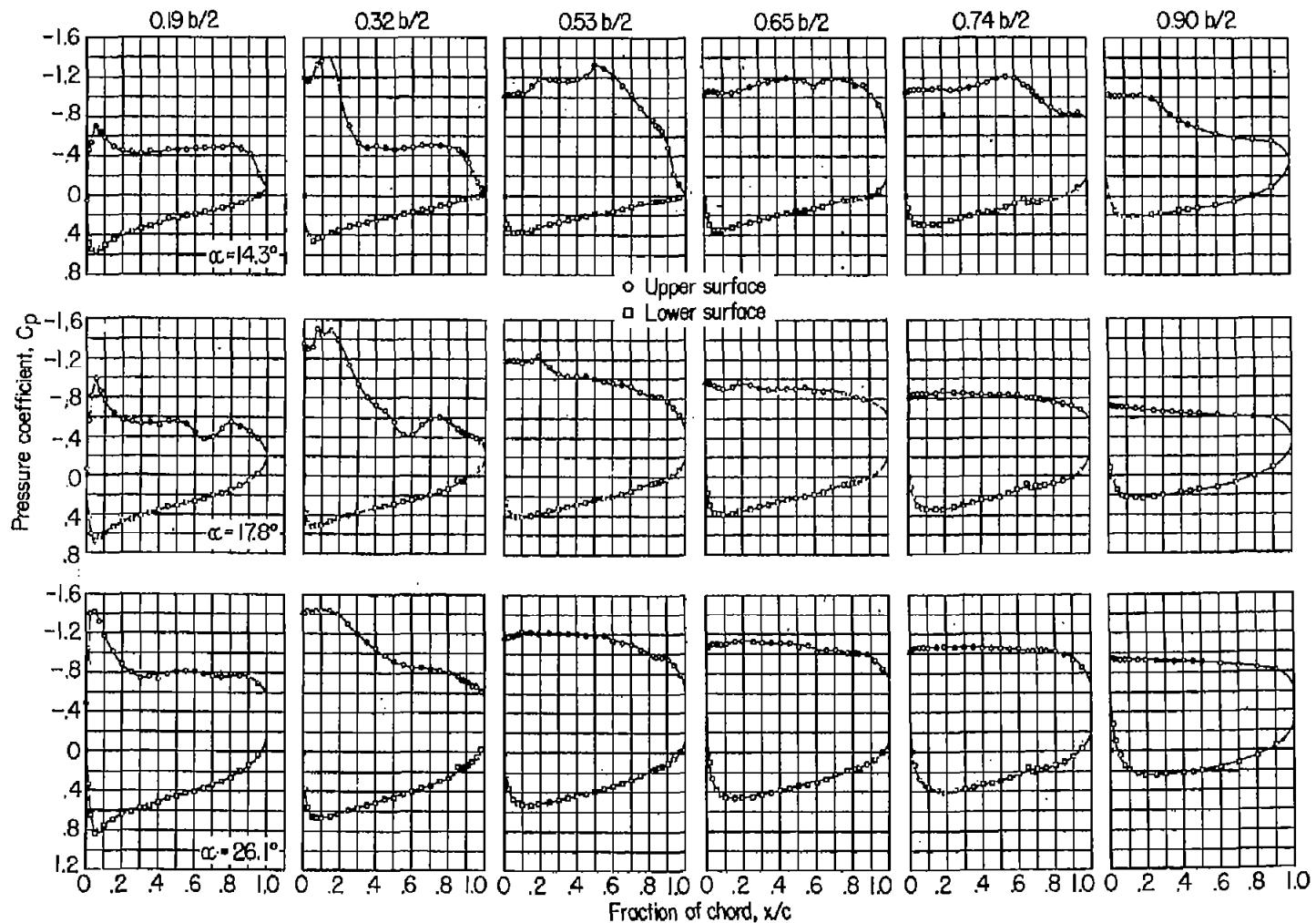
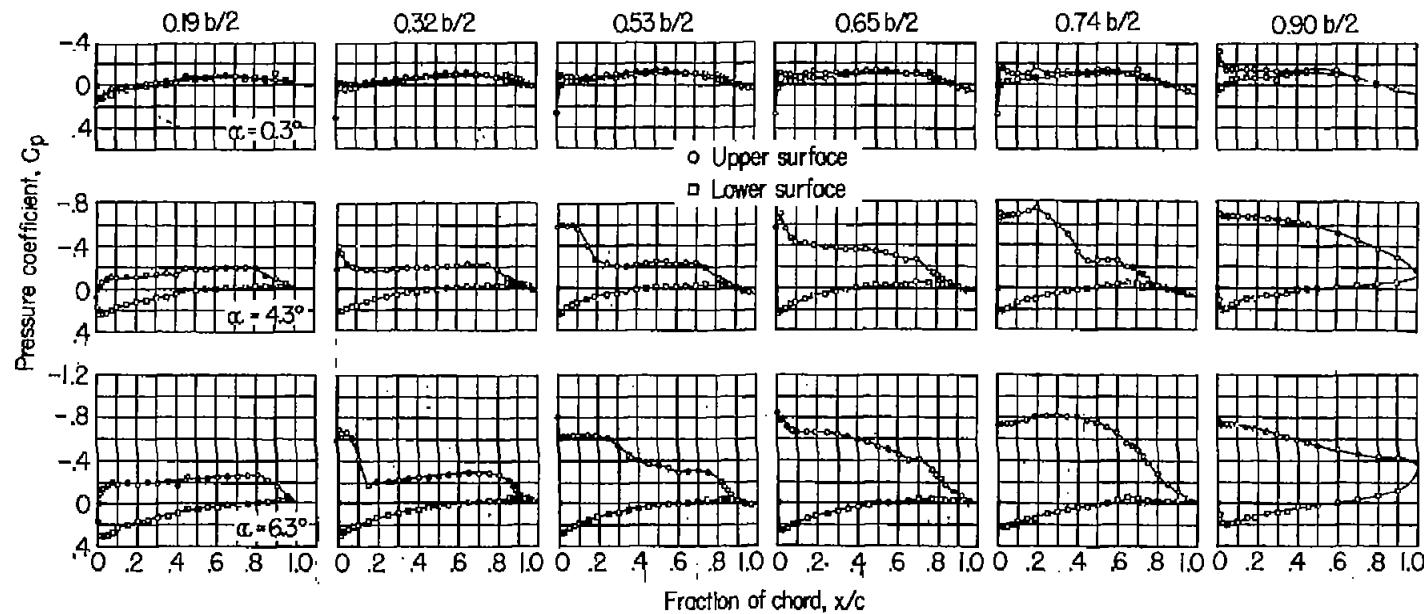
(d) $M = 0.92$; $C_{p,\text{sonic}} = -0.15$.

Figure 3.- Continued.



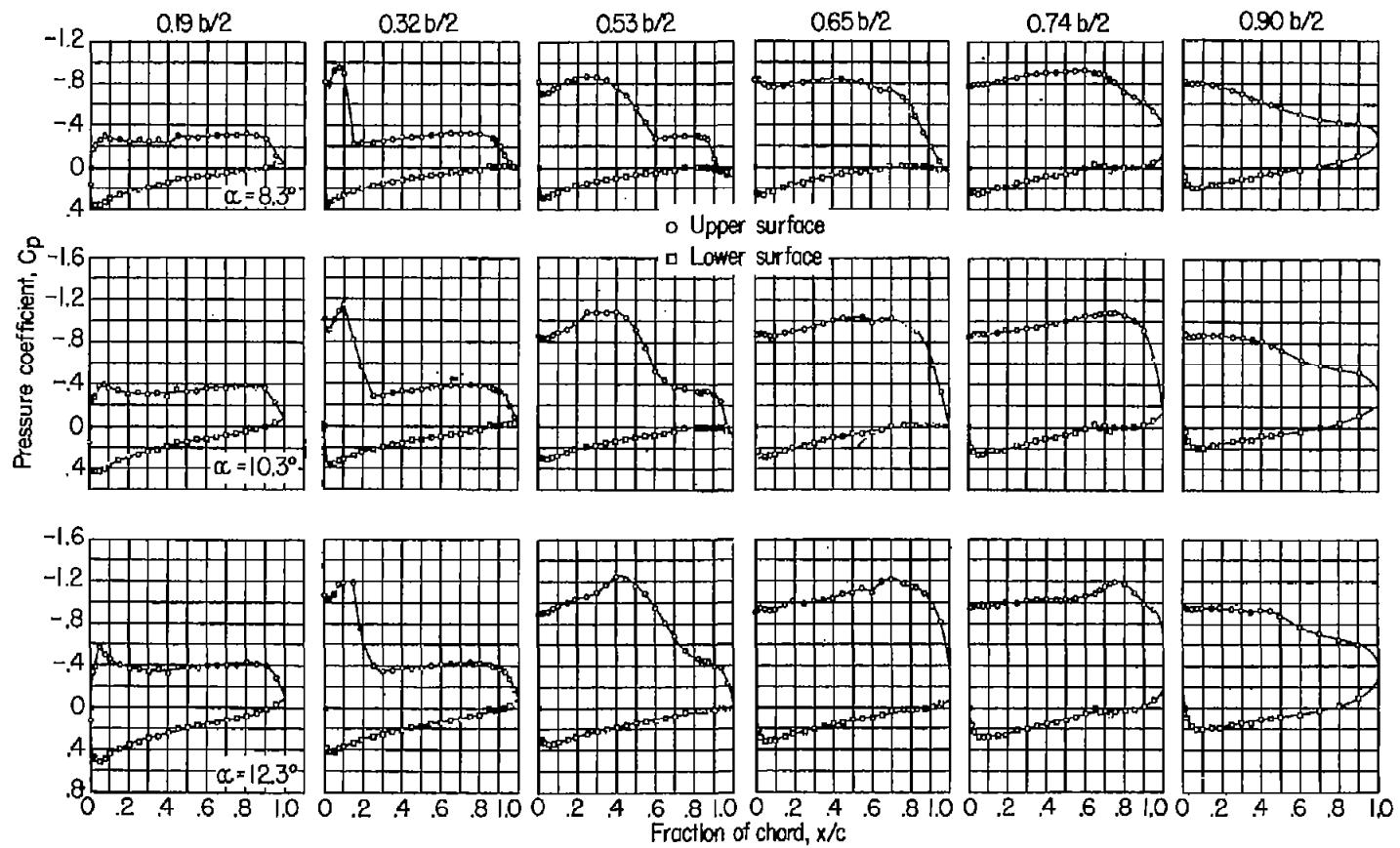
(d) Concluded.

Figure 3.- Continued.



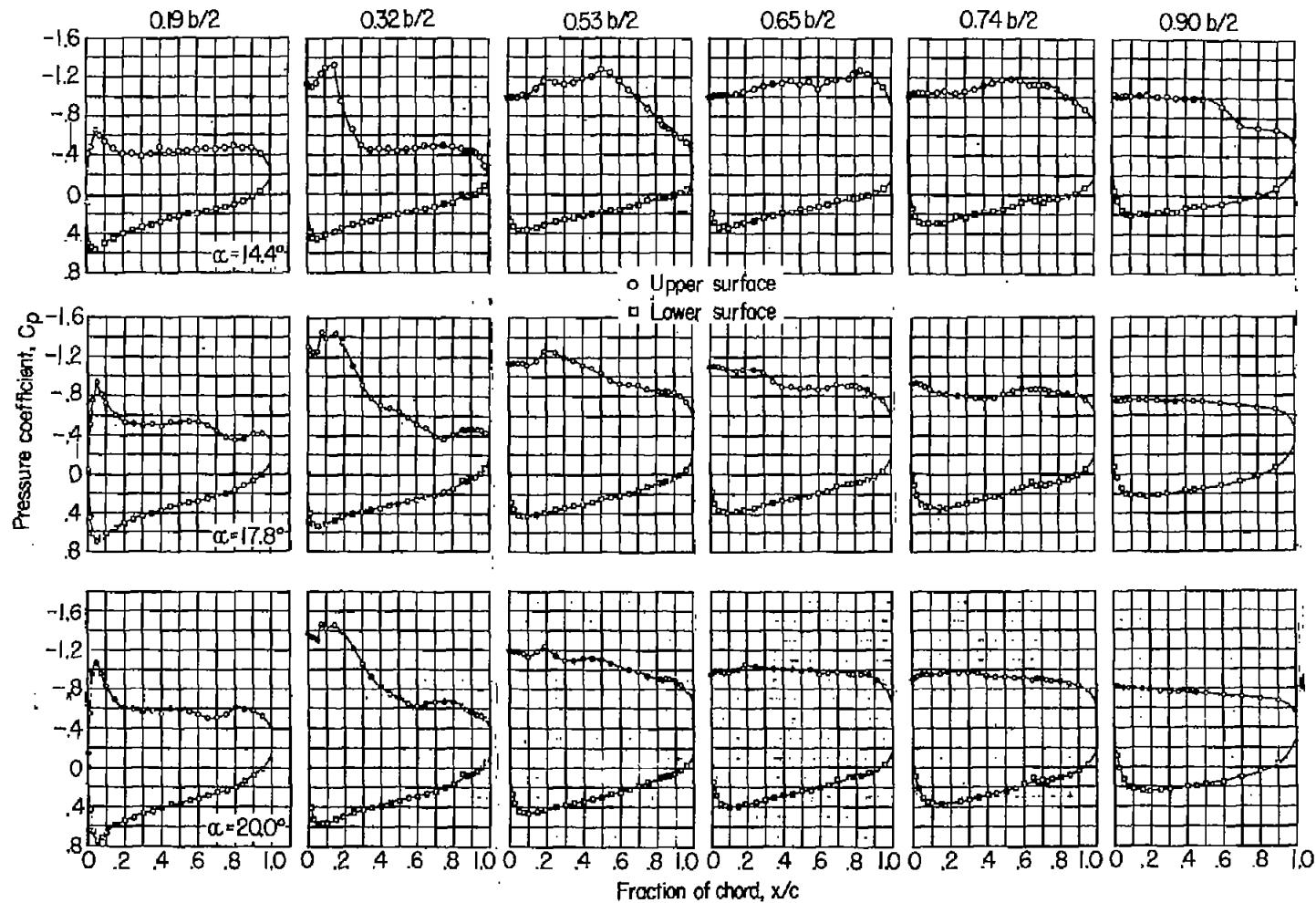
(e) $M = 0.94$; $C_{p, \text{sonic}} = -0.10$.

Figure 3.- Continued.



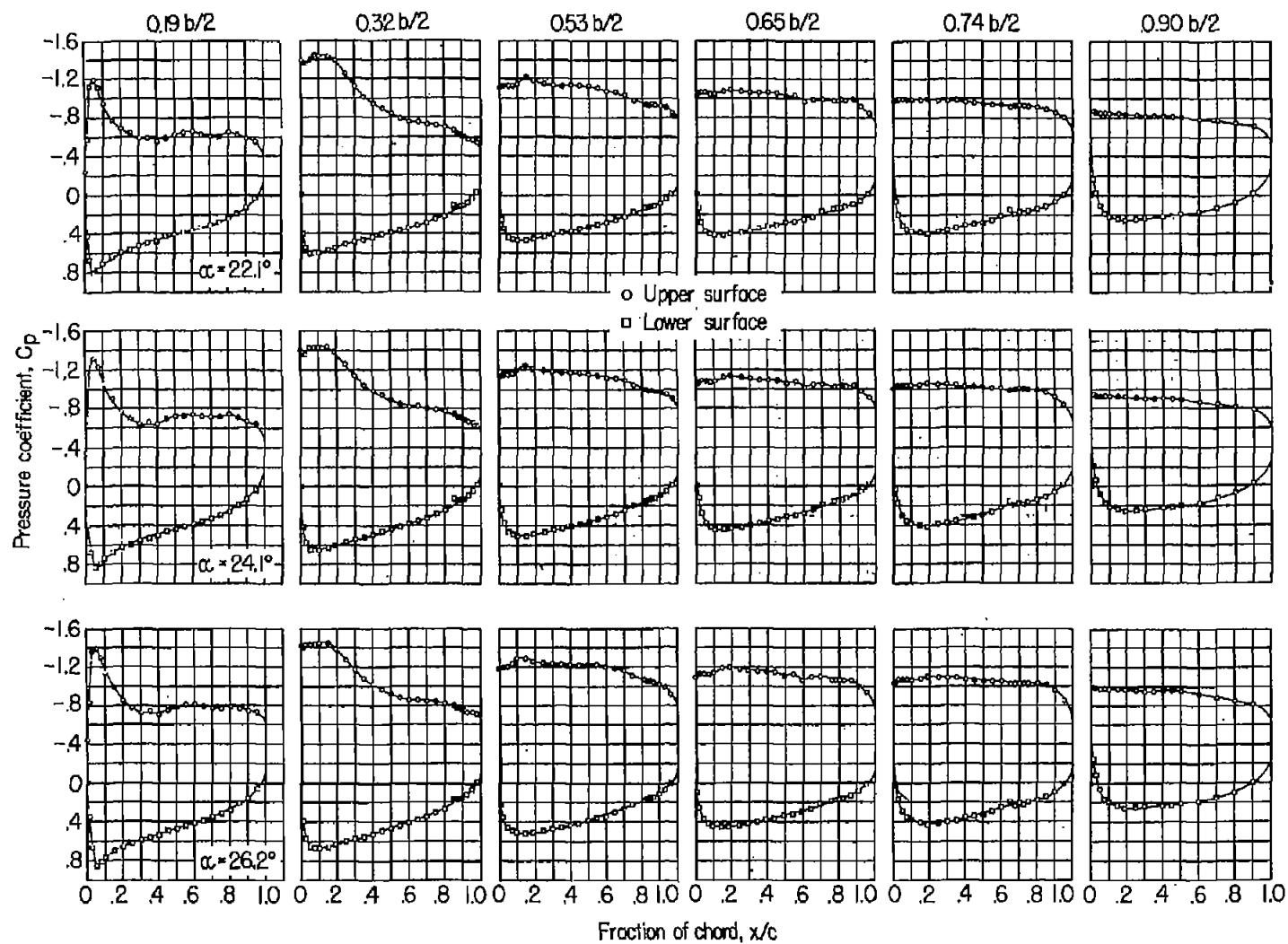
(e) Continued.

Figure 3.- Continued.



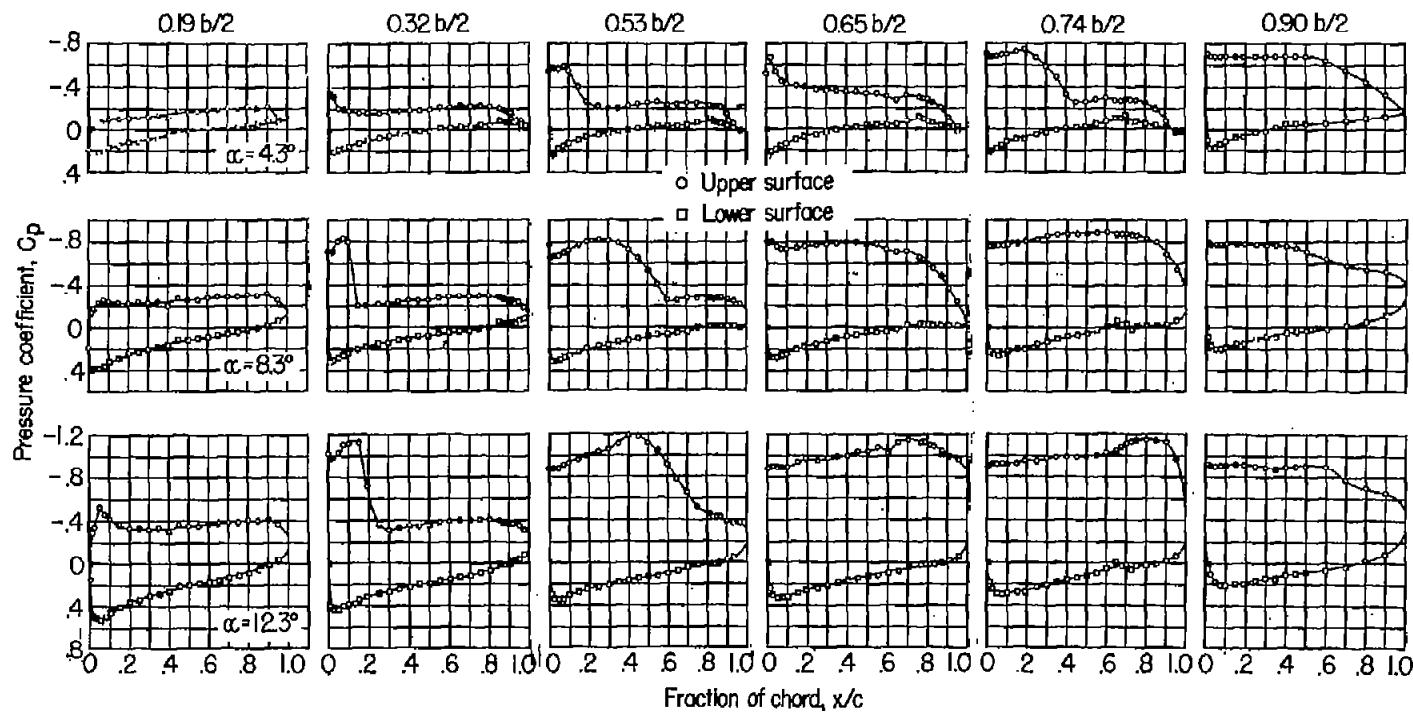
(e) Continued.

Figure 3.- Continued.



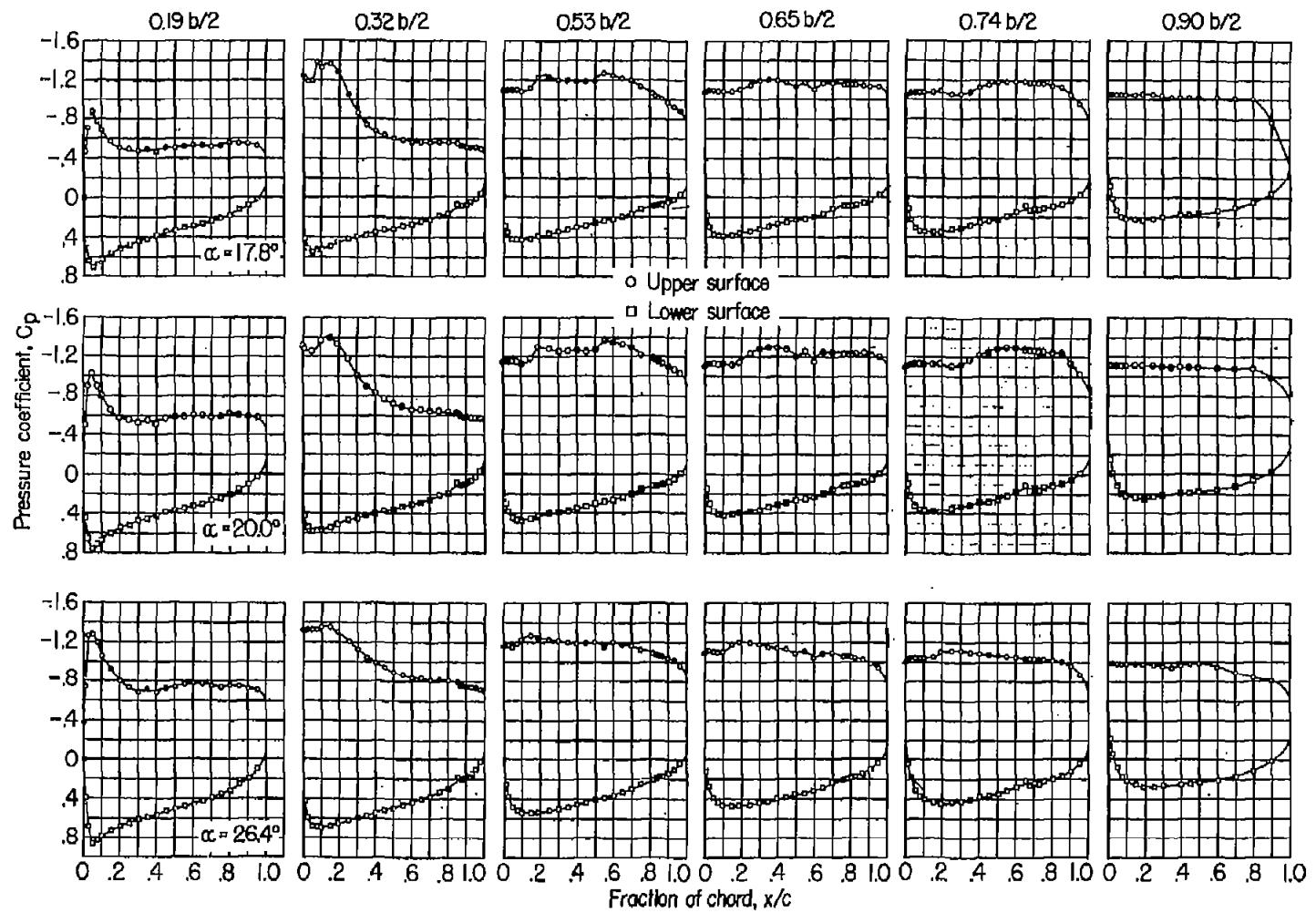
(e) Concluded.

Figure 3.- Continued.



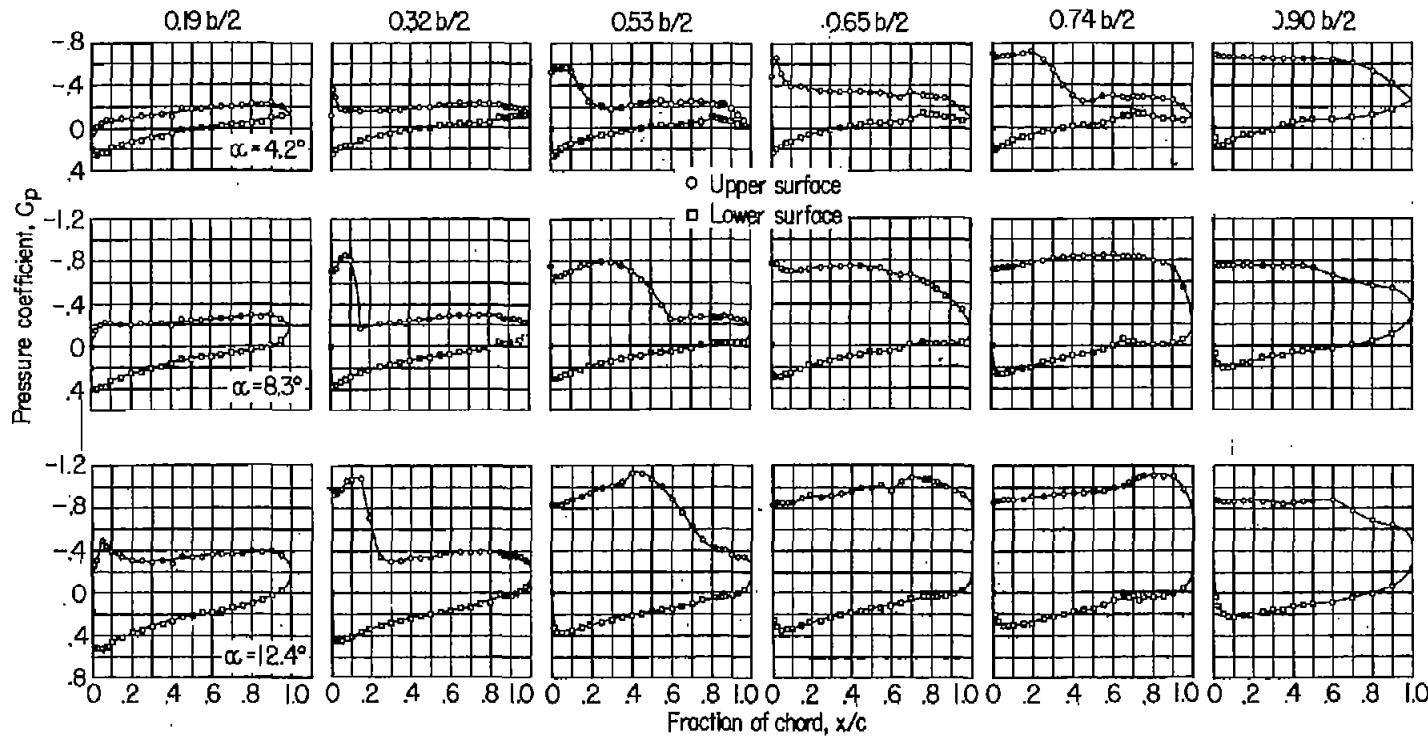
(f) $M = 0.96; C_{p,\text{sonic}} = -0.07$.

Figure 3.- Continued.



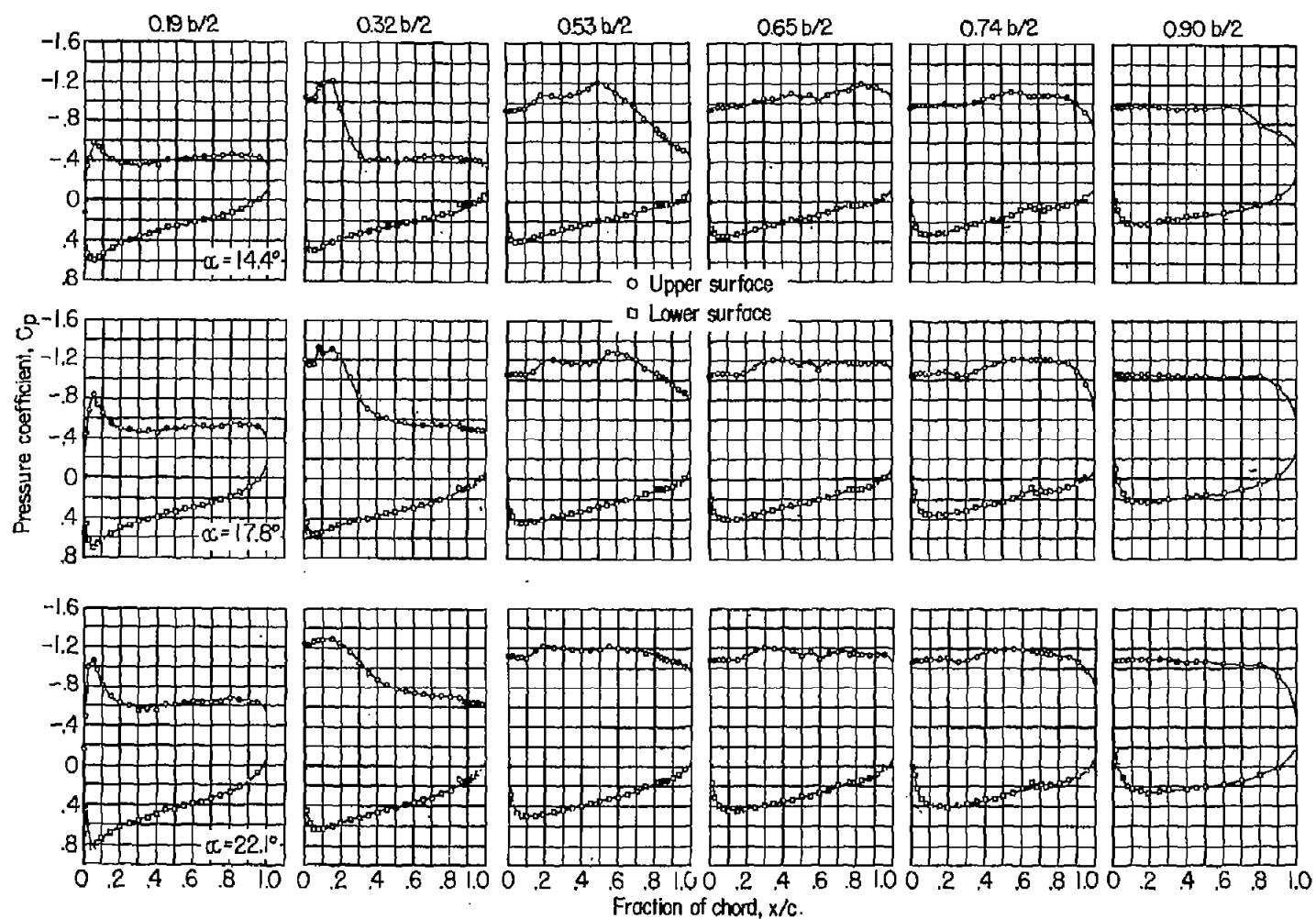
(f) Concluded.

Figure 3.- Continued.



(g) $M = 0.98$; $C_{p, \text{sonic}} = -0.04$.

Figure 3.- Continued.



(g) Concluded.

Figure 3.- Continued.

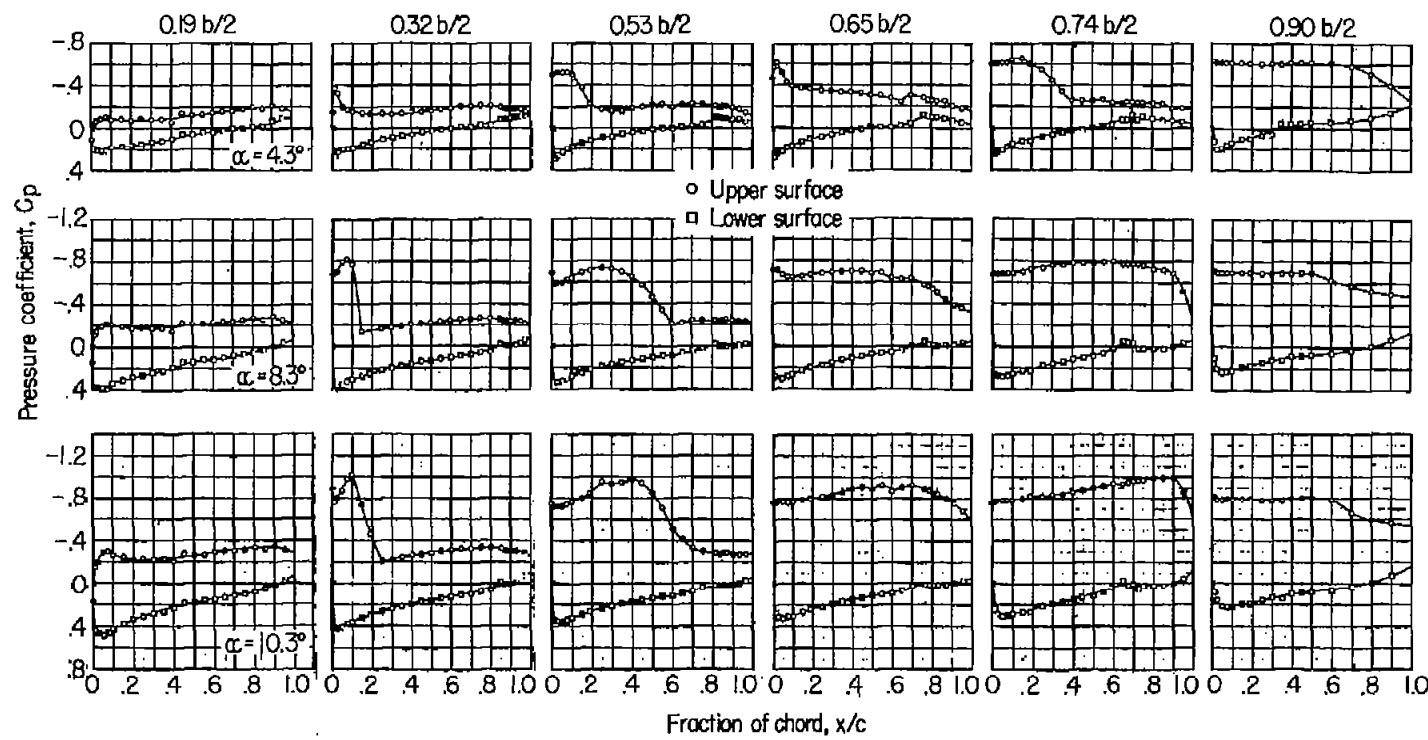
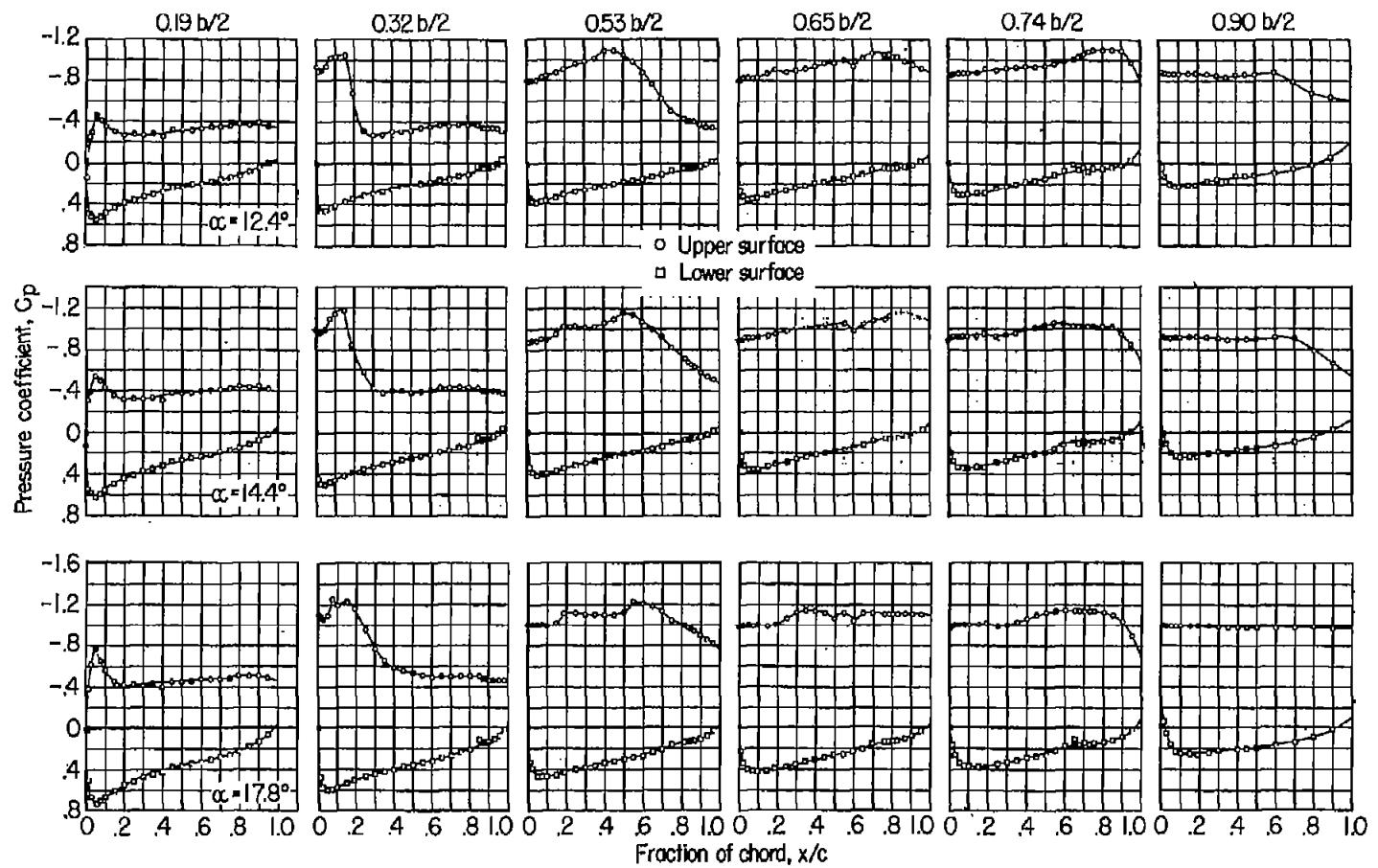


Figure 3.-- Continued.



(h) Concluded.

Figure 3.- Continued.

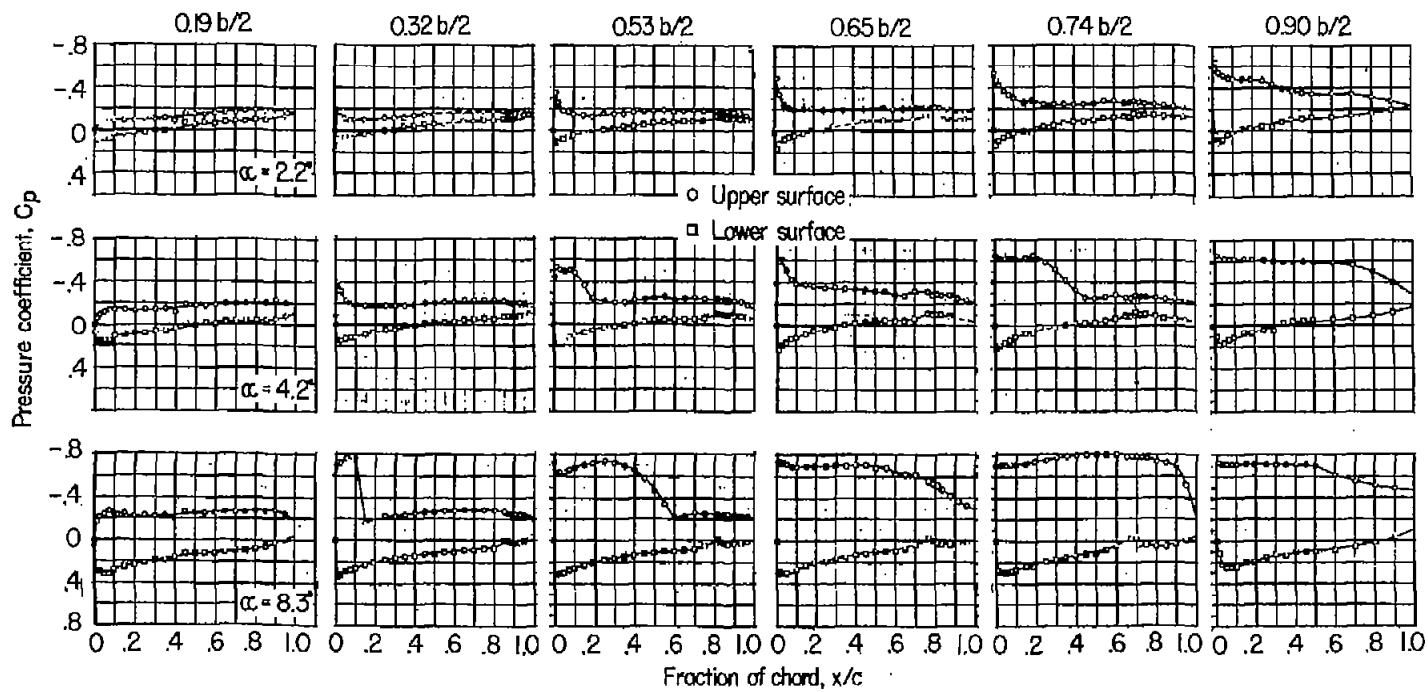
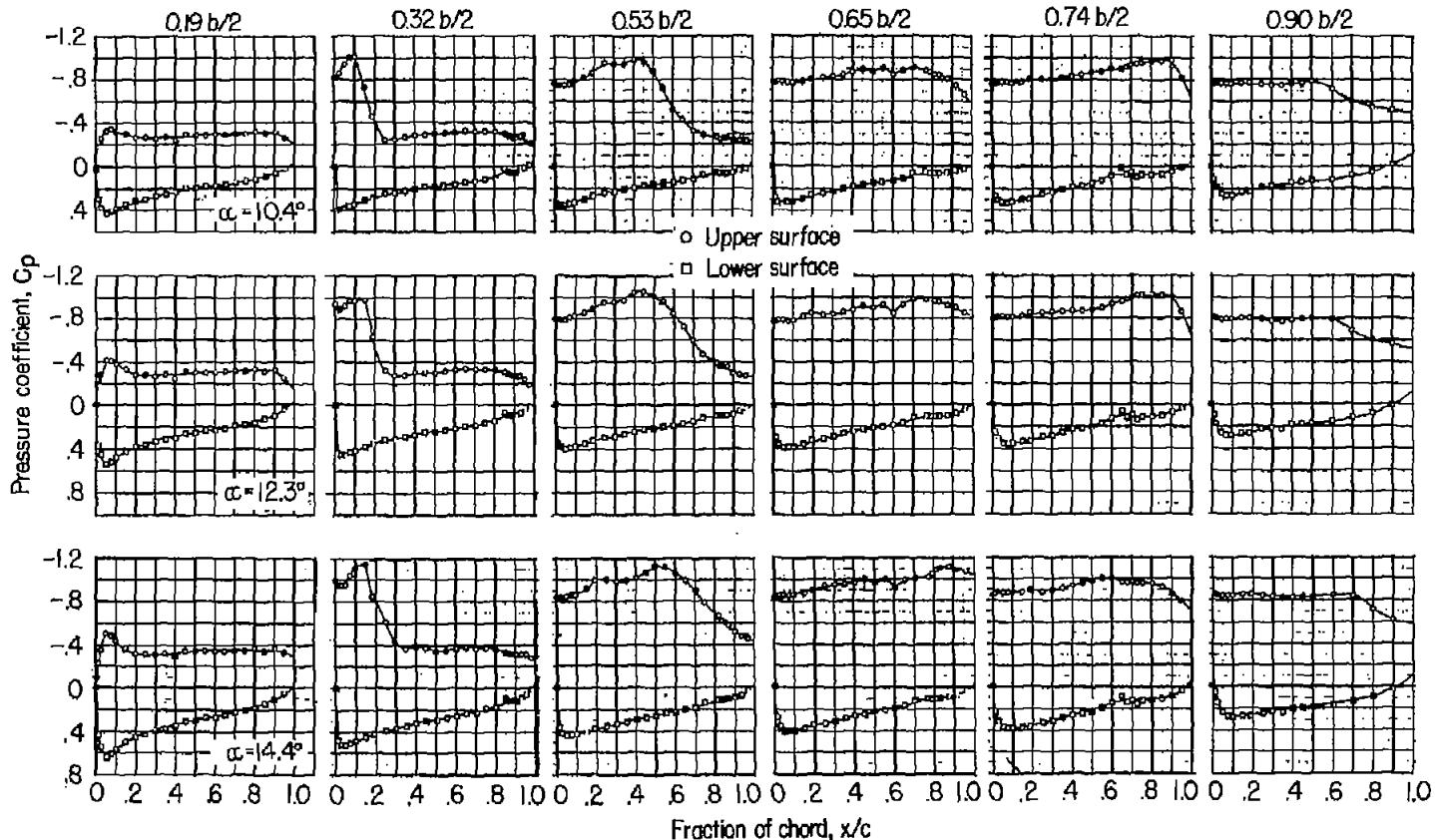
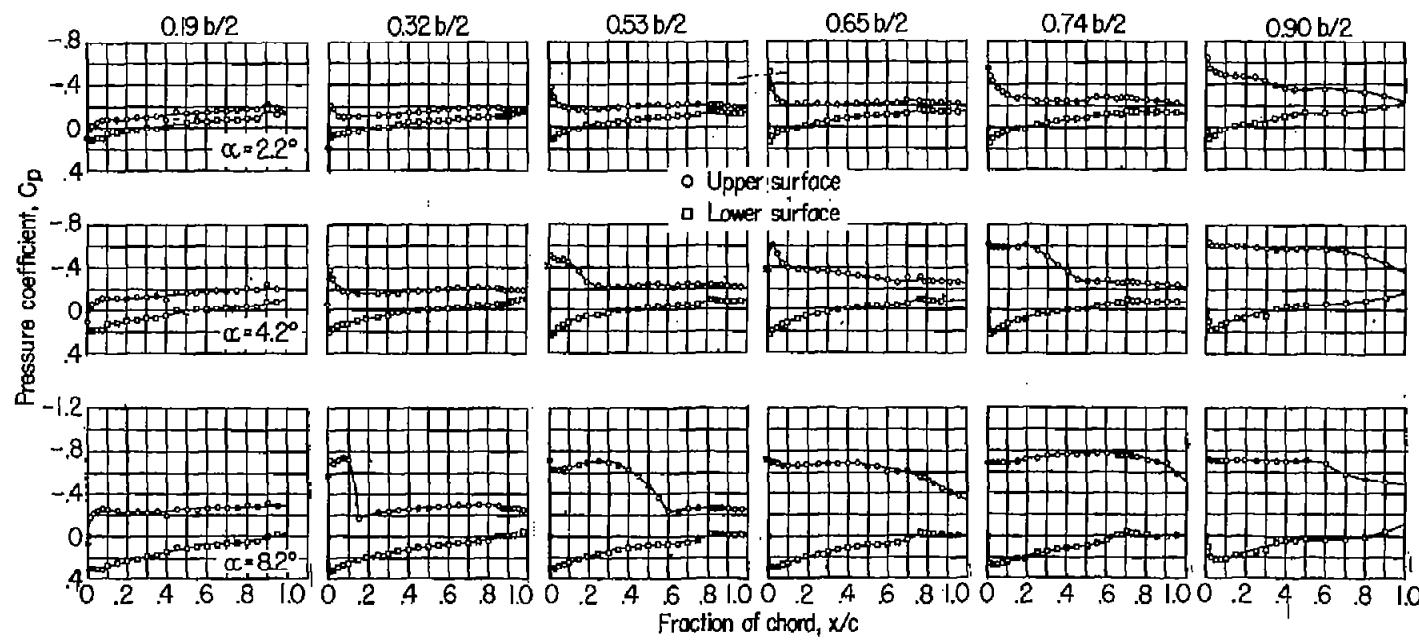
(1) $M = 1.03.$

Figure 3.- Continued.



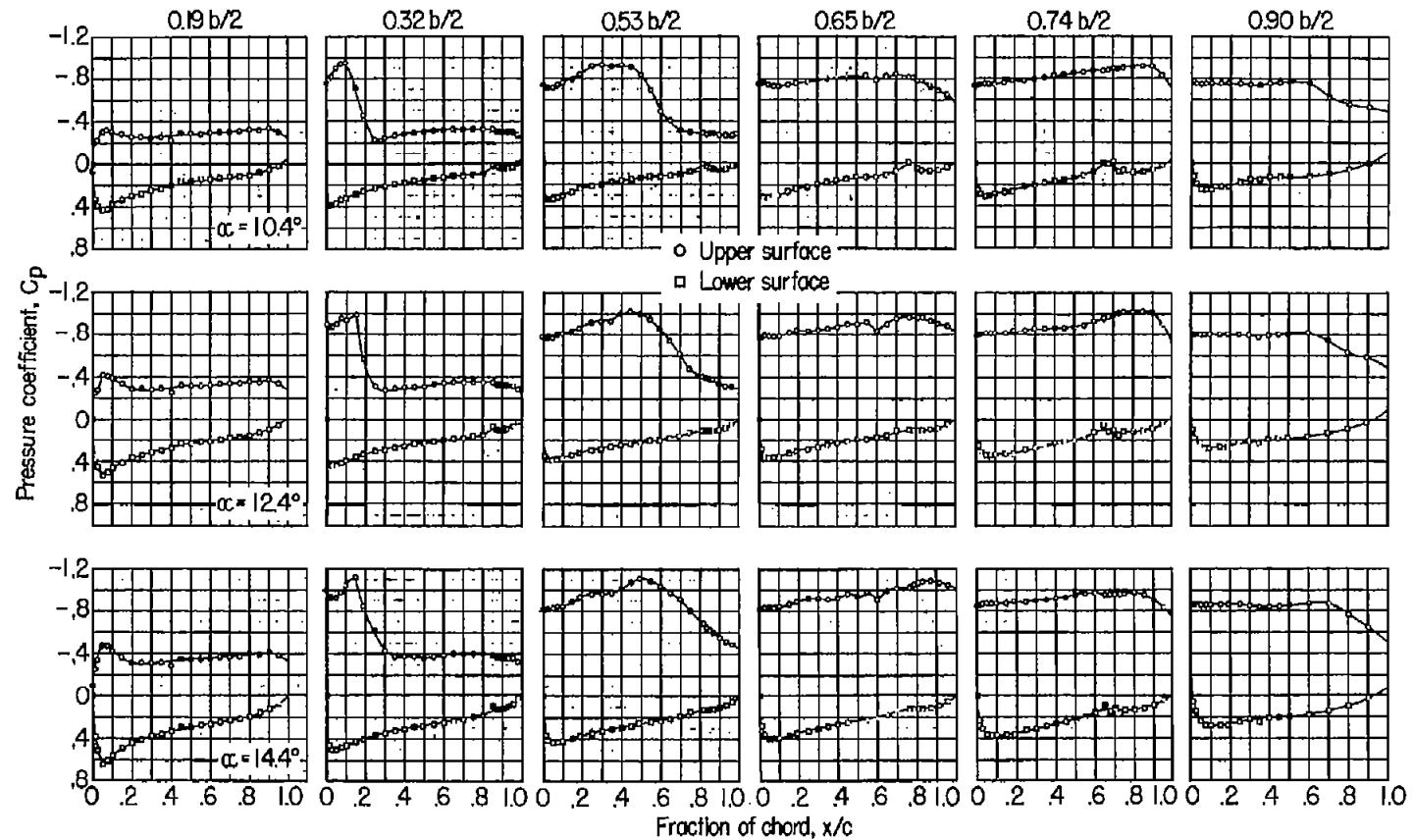
(i) Concluded.

Figure 3.- Continued.



(j) $M = 1.05$.

Figure 3.- Continued.



(j) Concluded.

Figure 3.- Concluded.

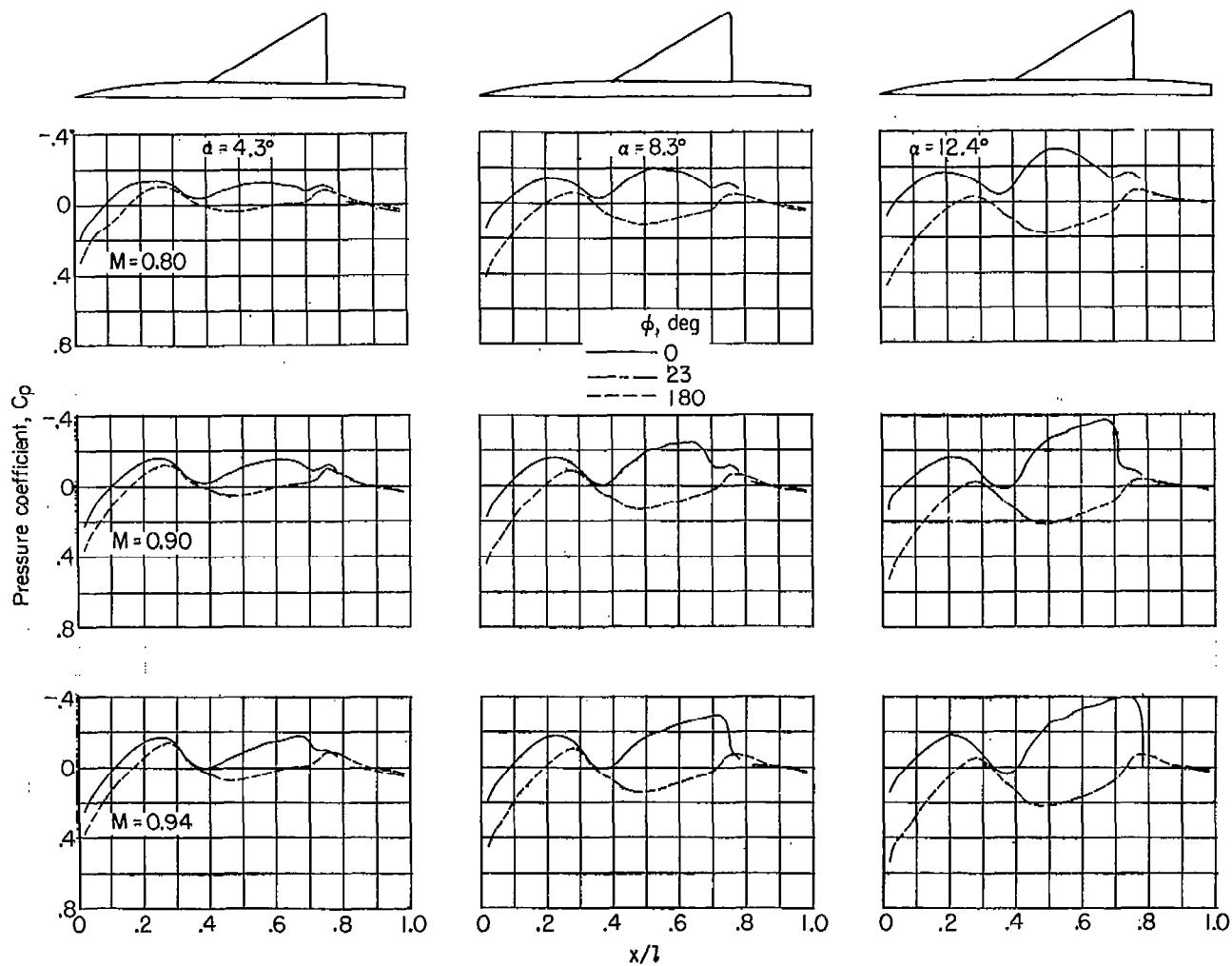
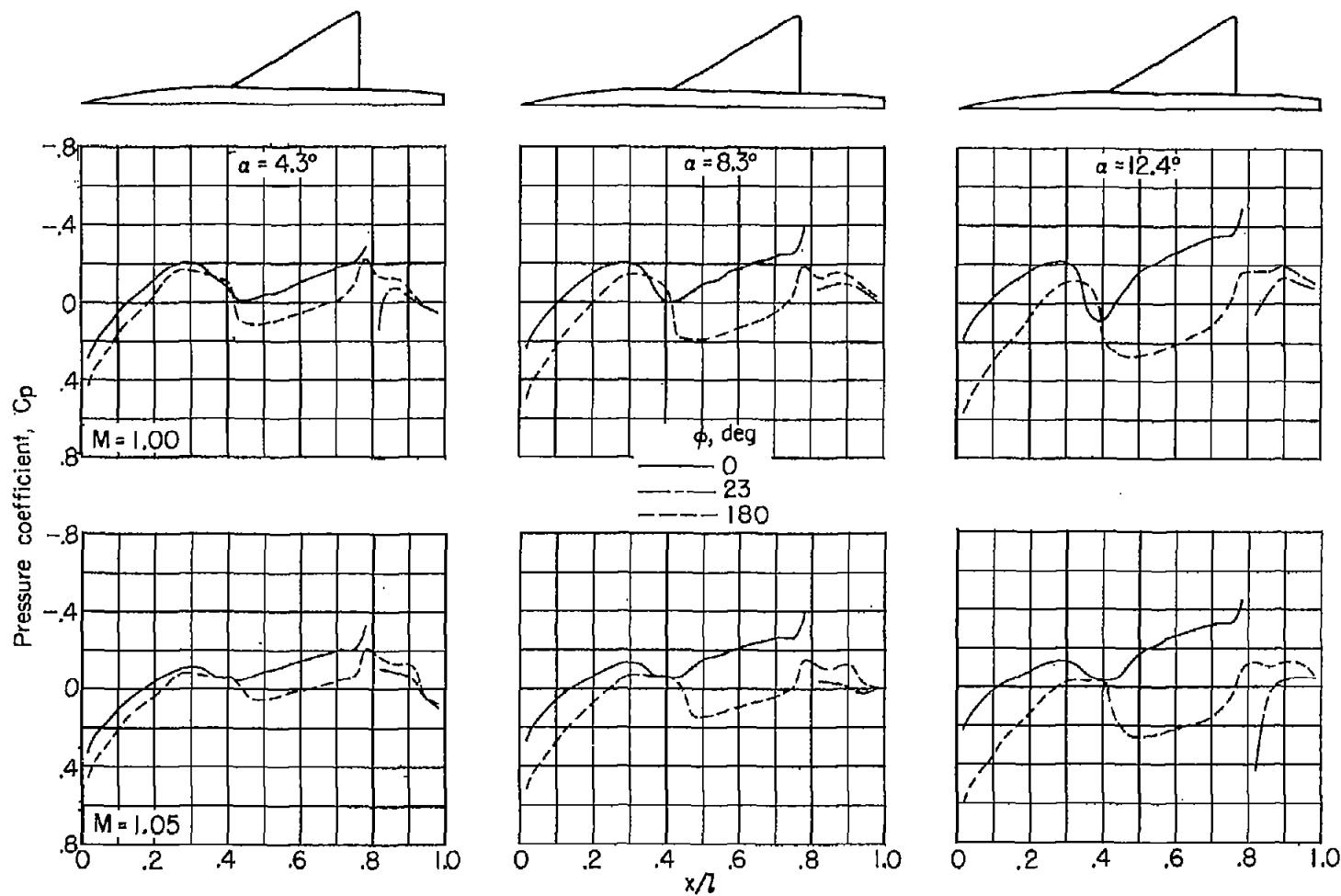
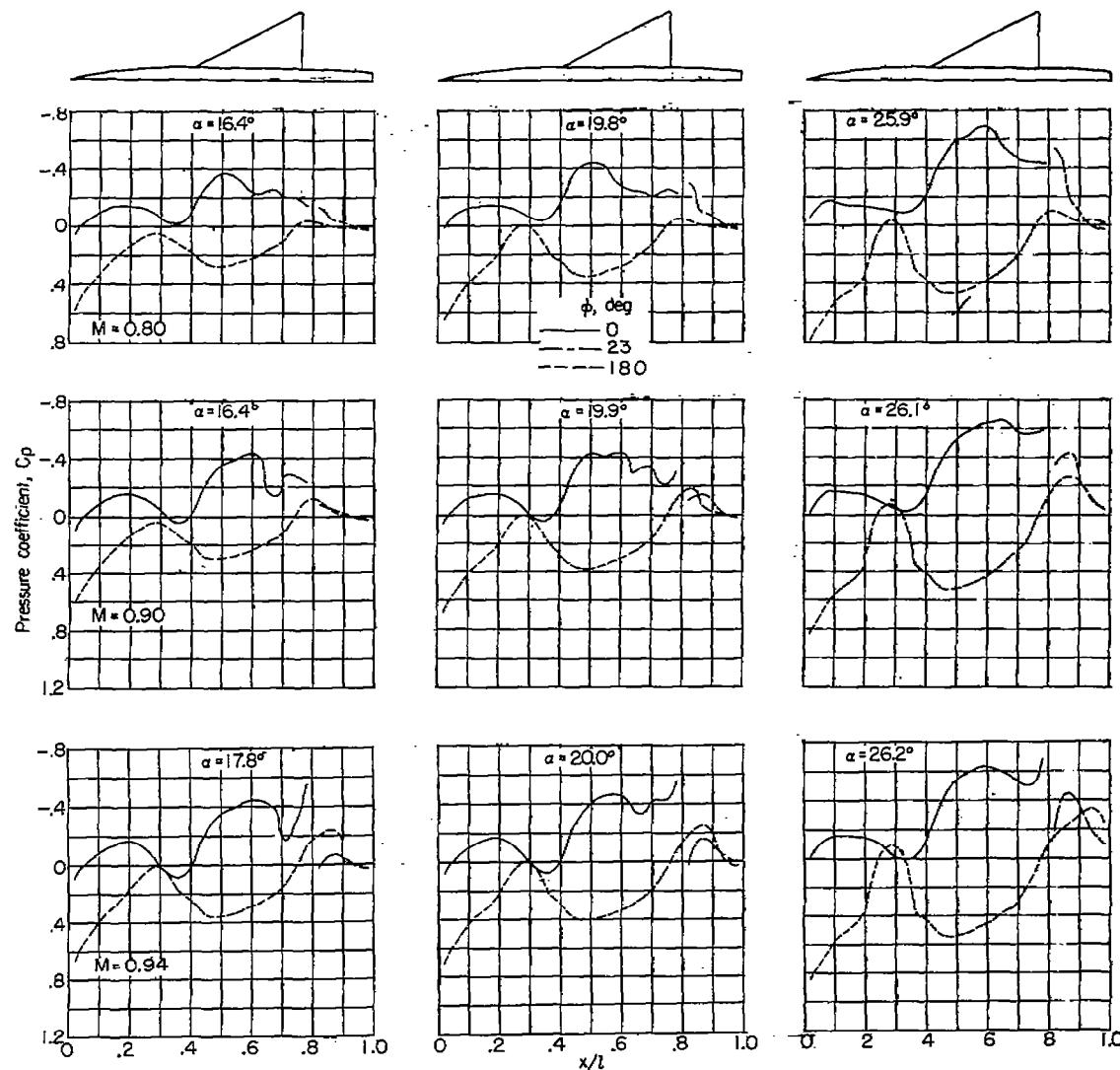
(a) $\alpha = 4.3^\circ, 8.3^\circ$, and 12.4° .

Figure 4.- Body longitudinal pressure distributions for several Mach numbers and angles of attack.



(b) $\alpha = 4.3^\circ$, 8.3° , and 12.4° .

Figure 4.- Continued.



(c) $\alpha = 16.4^\circ, 17.8^\circ, 20^\circ$, and 26° .

Figure 4.- Concluded.

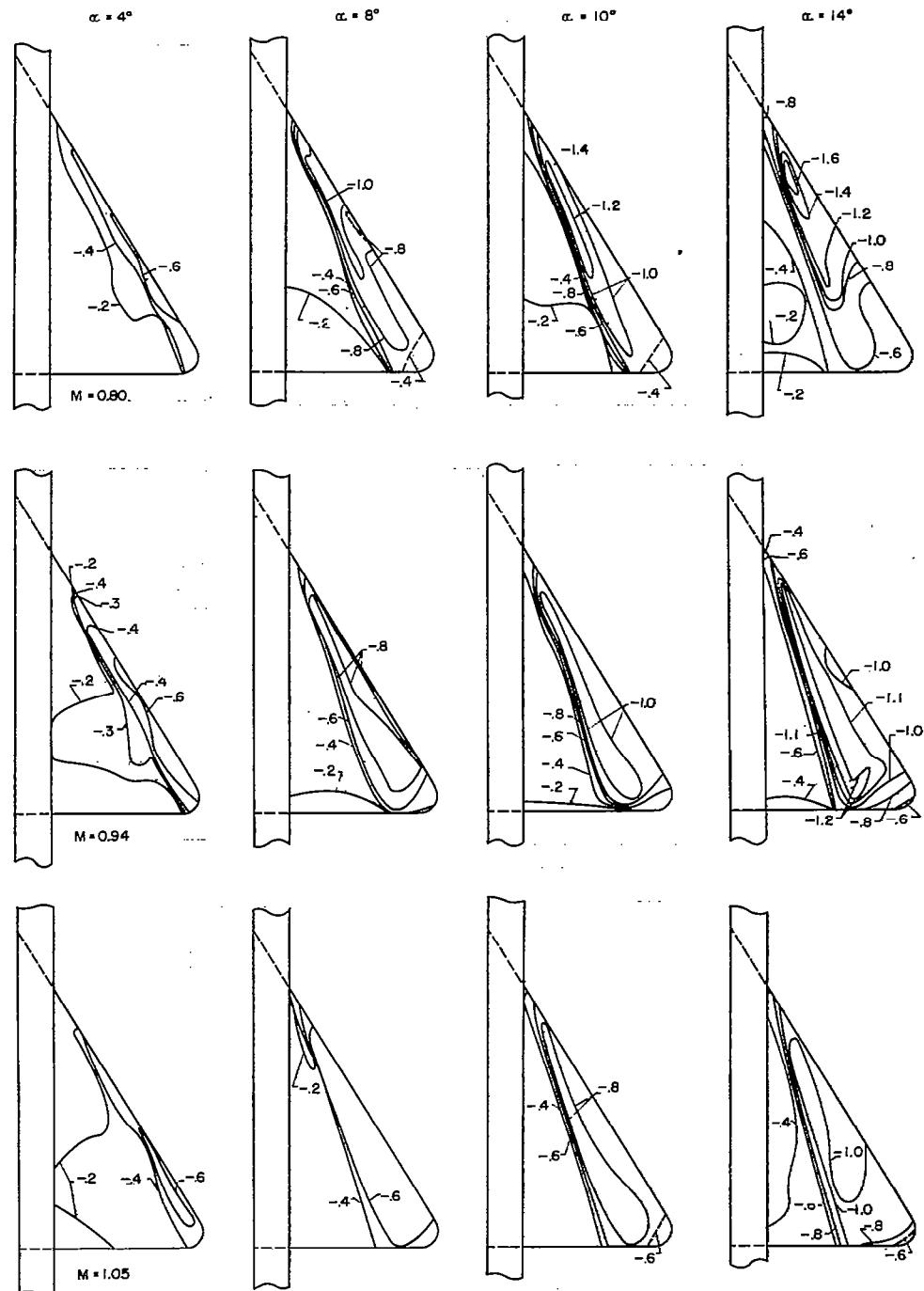


Figure 5.- Pressure contours on wing upper surface at three Mach numbers and four nominal angles of attack.

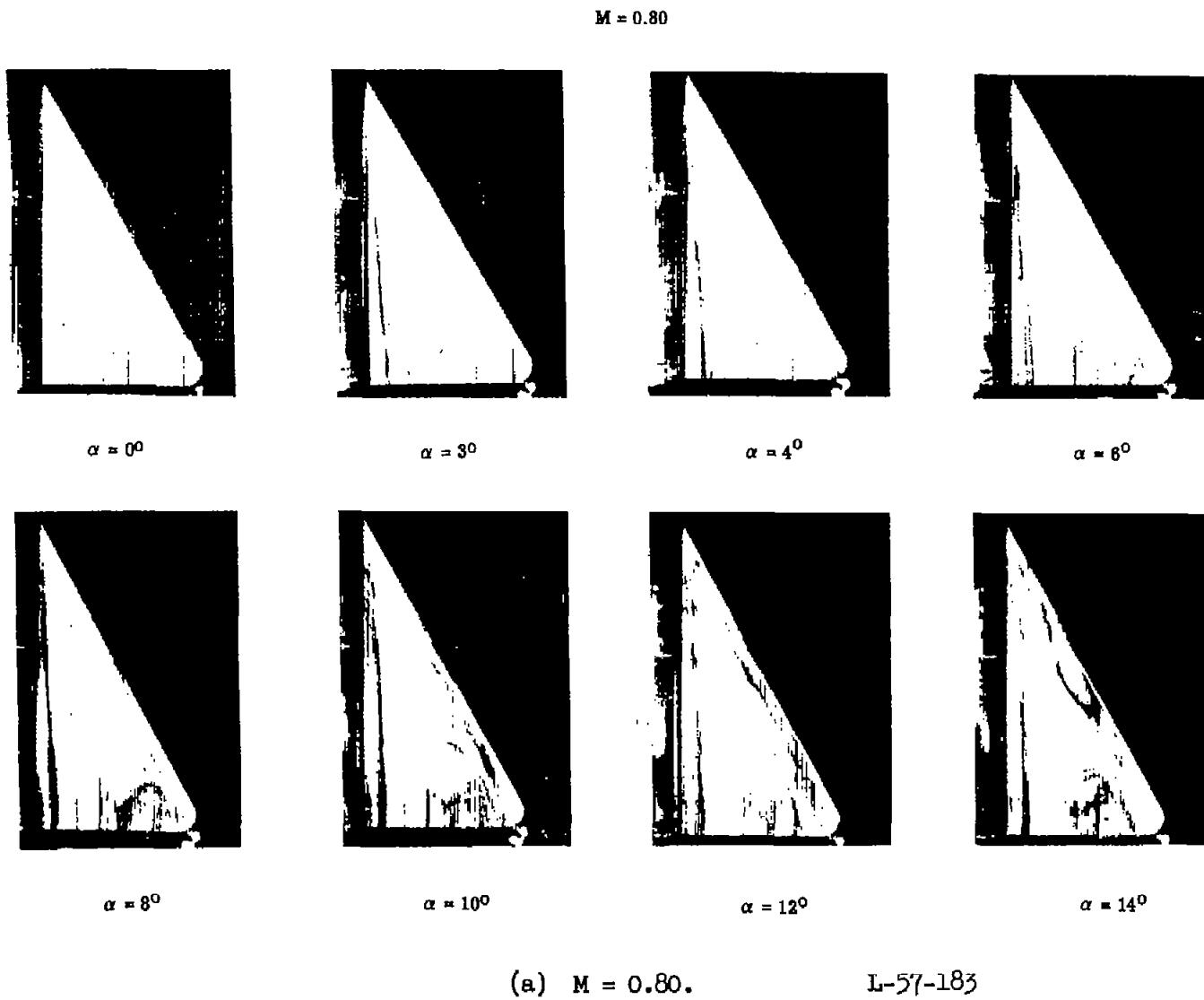
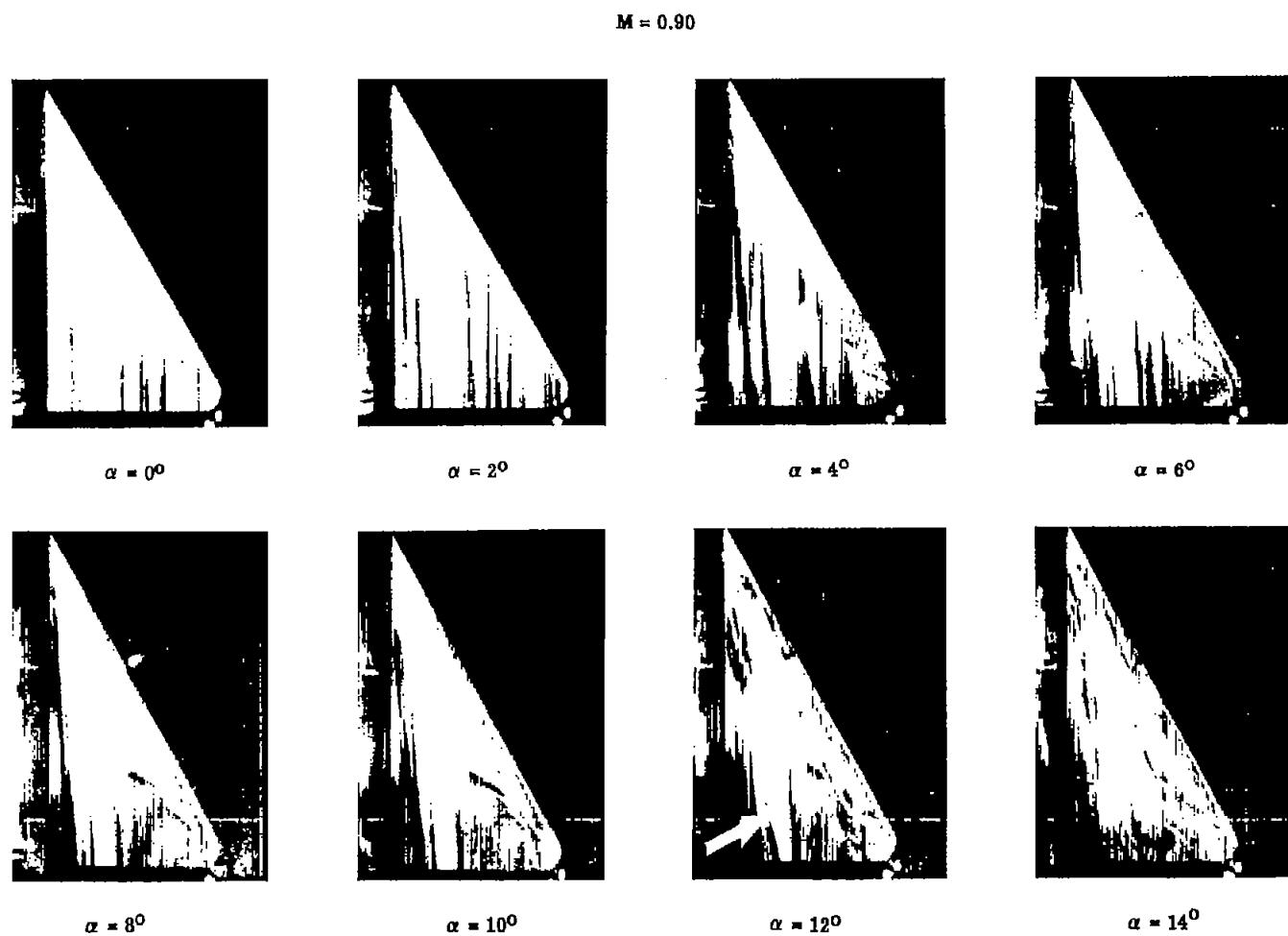


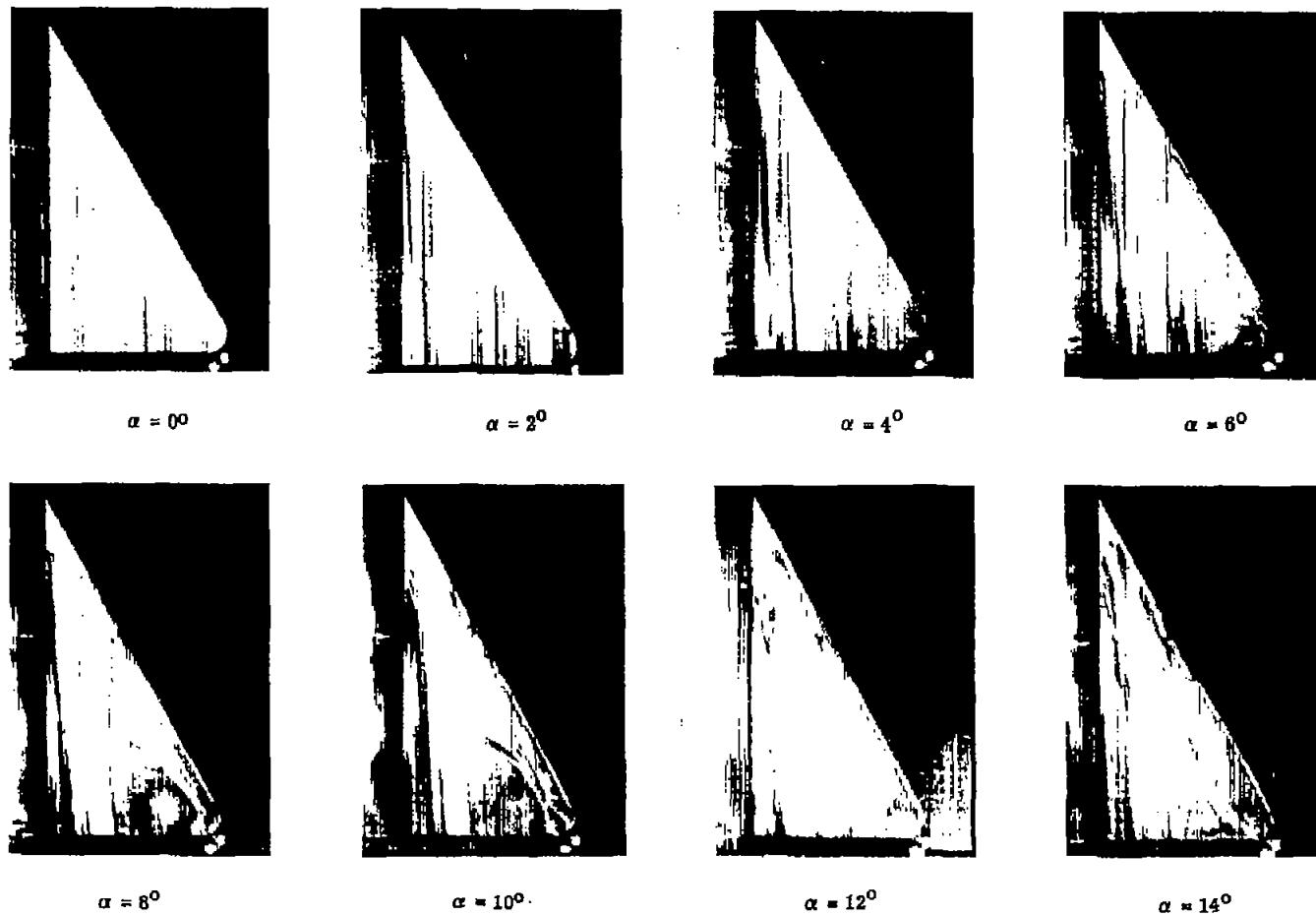
Figure 6.- Boundary-layer flow patterns on a 60° delta wing.

(b) $M = 0.90.$

L-57-184

Figure 6.- Continued.

M = 0.94



(c) M = 0.94.

L-57-185

Figure 6.--Continued.

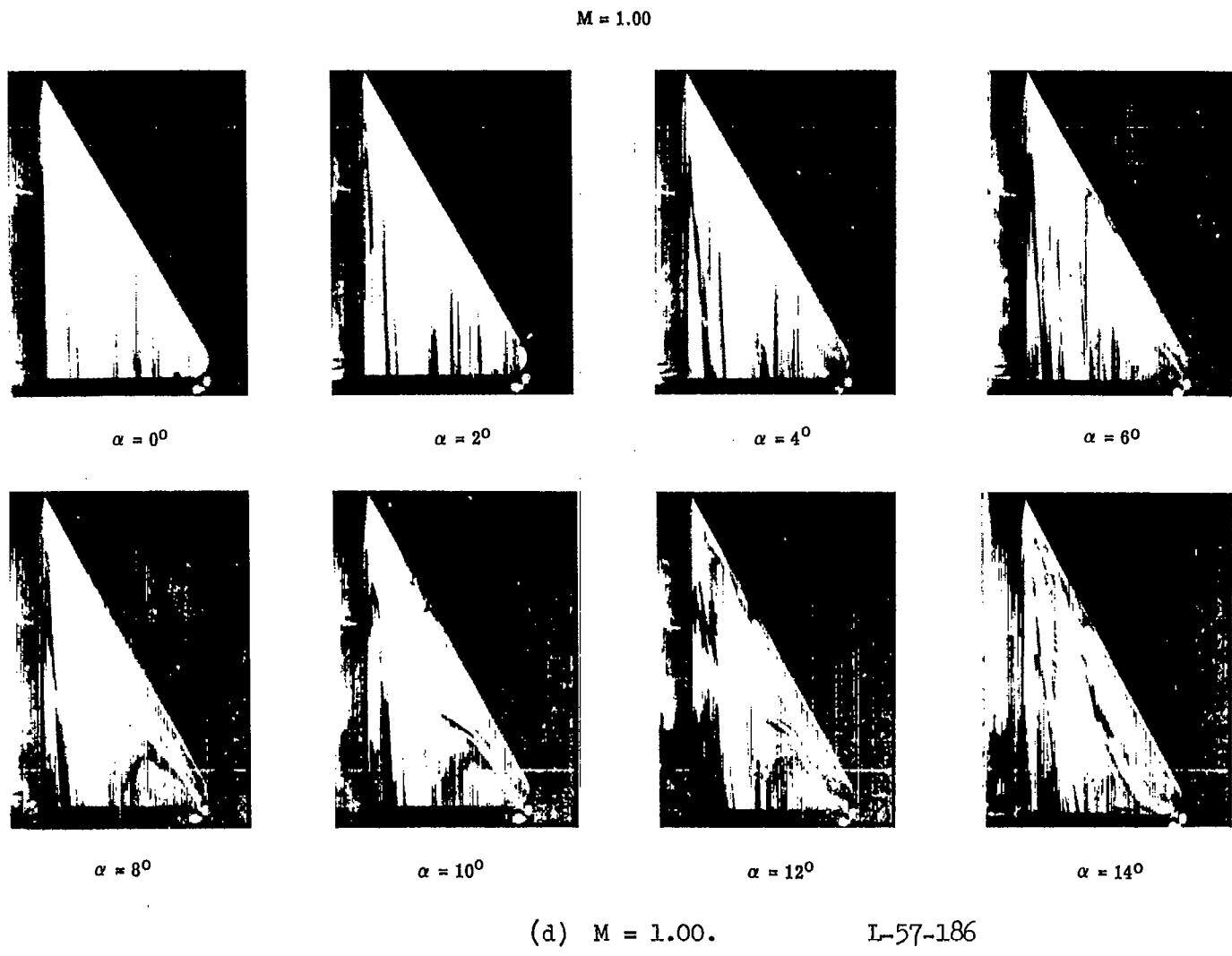


Figure 6.- Continued.

M = 1.05



$\alpha = 0^\circ$



$\alpha = 2^\circ$



$\alpha = 4^\circ$



$\alpha = 6^\circ$



$\alpha = 8^\circ$



$\alpha = 10^\circ$



$\alpha = 12^\circ$



$\alpha = 14^\circ$

(e) M = 1.05.

L-57-187

Figure 6.- Concluded.

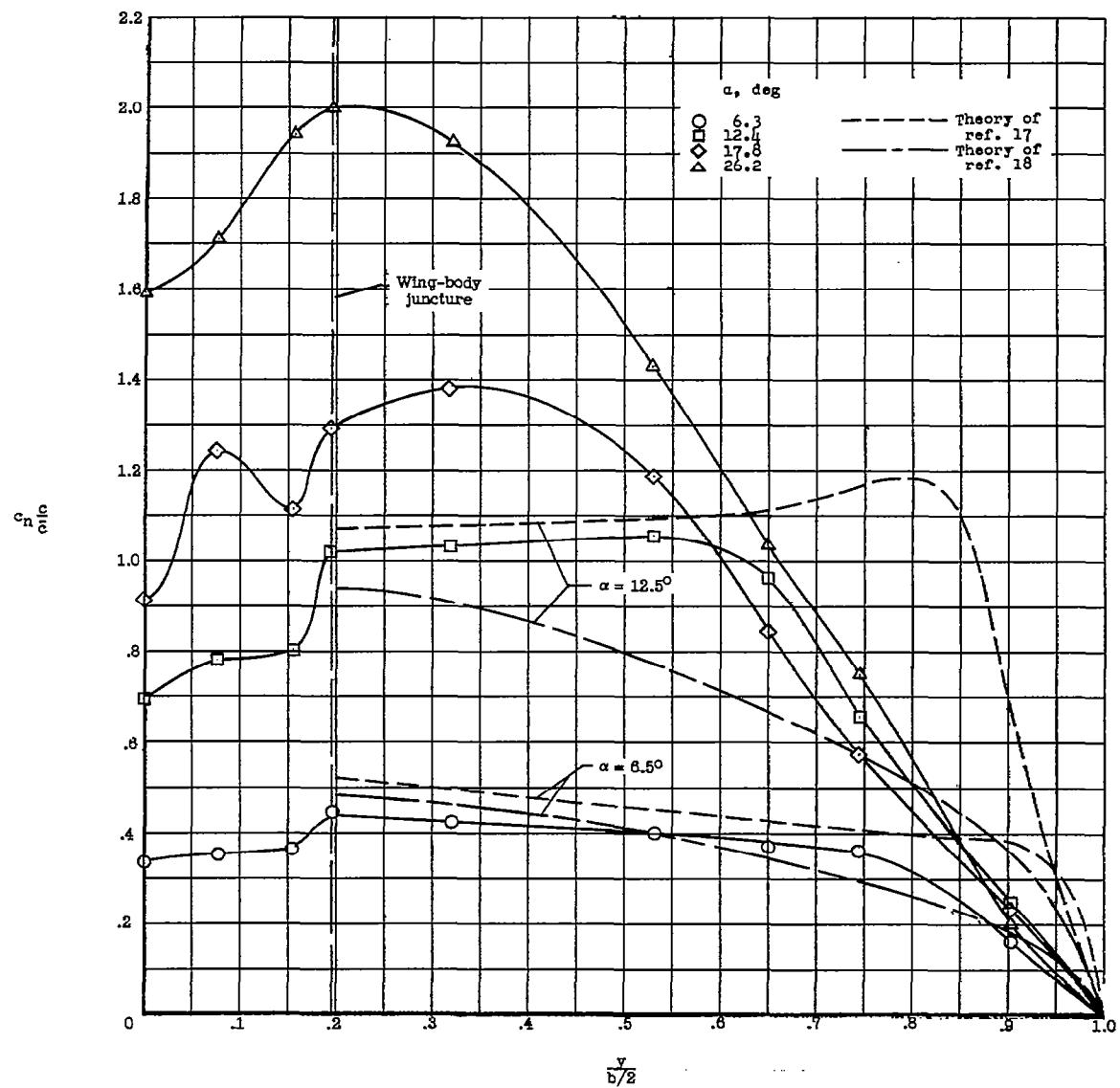


Figure 7.- Variation of section normal-load parameter with wing semispan including the fuselage carryover for several angles of attack at a Mach number of 0.94.

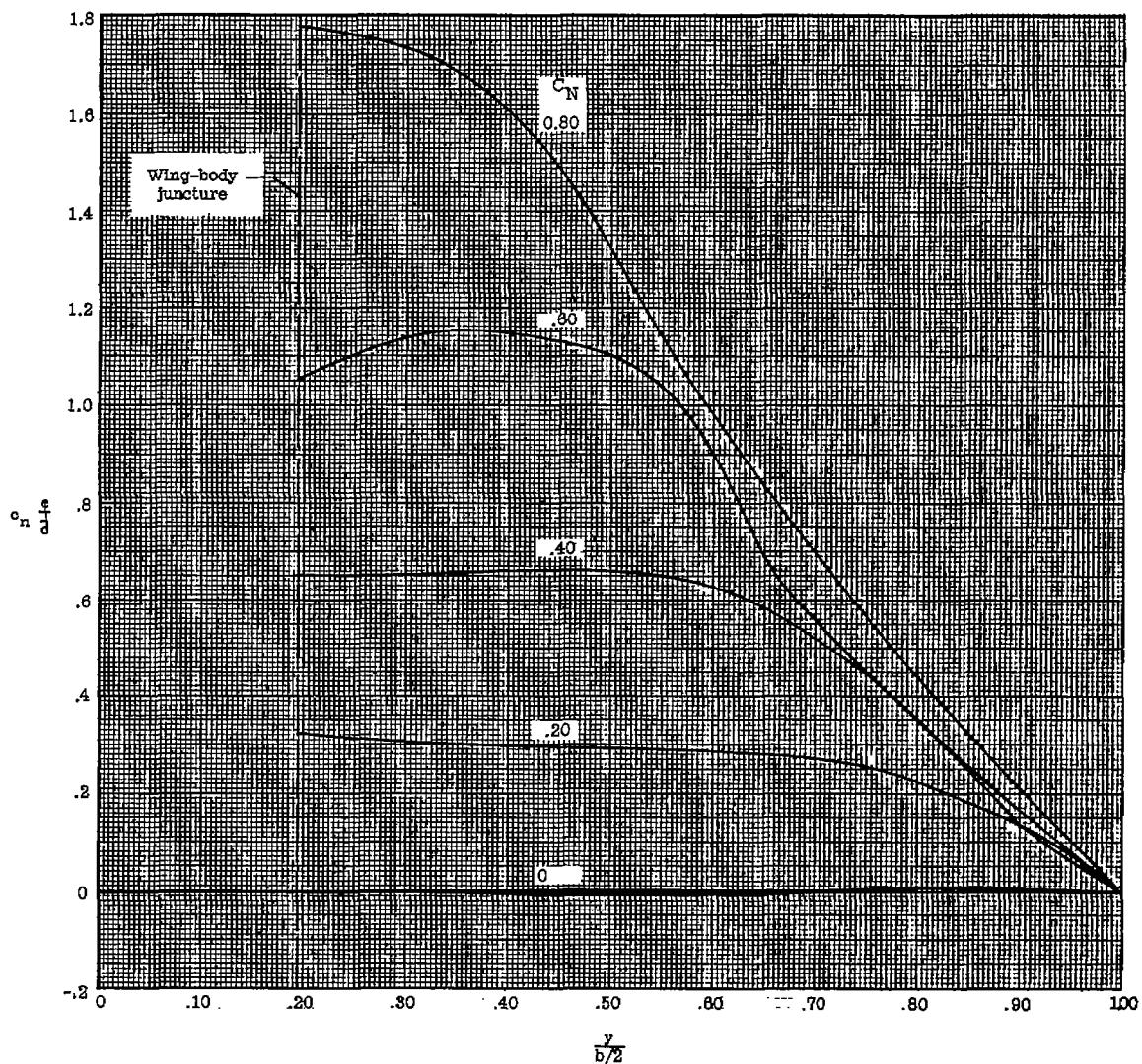
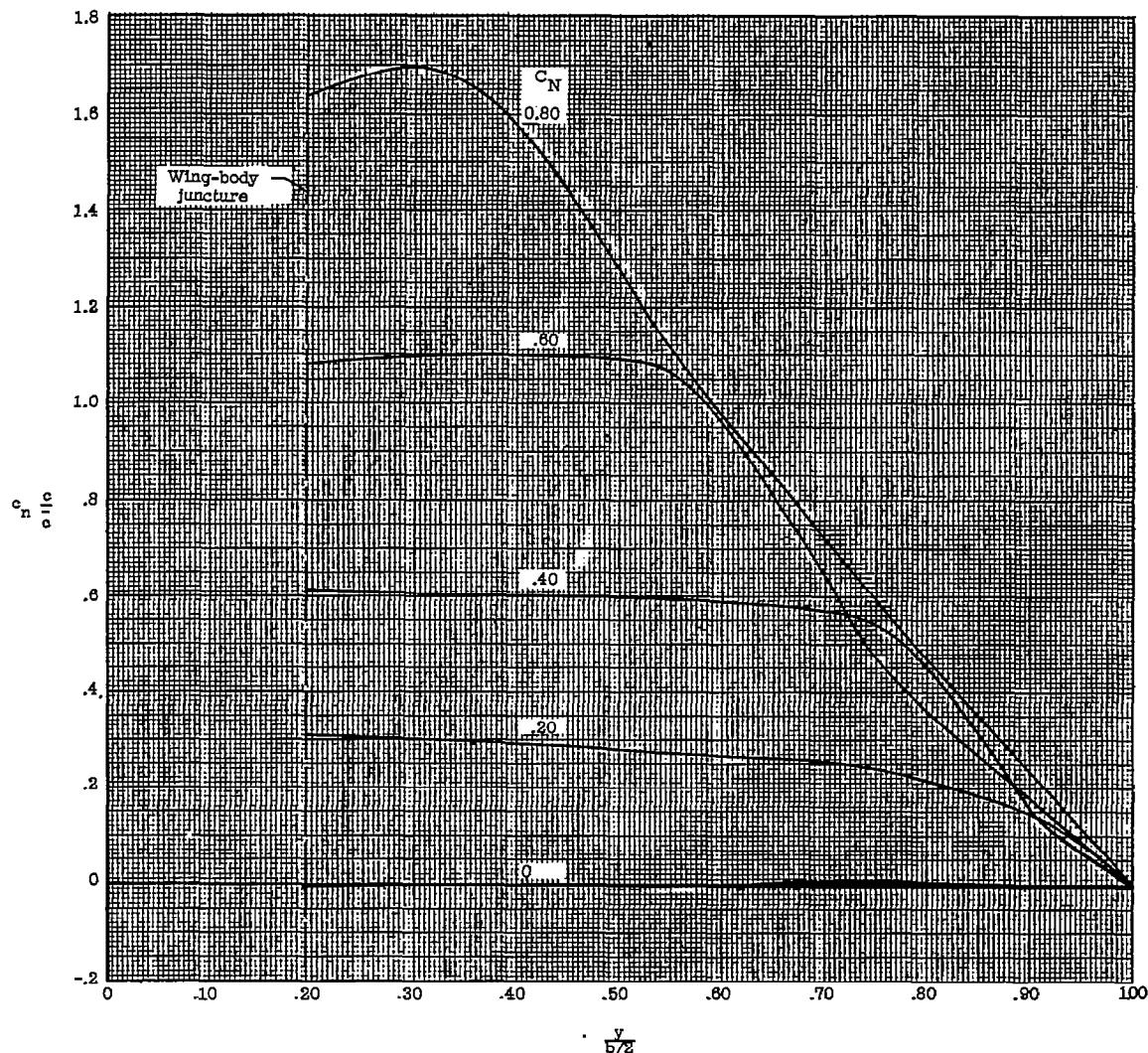
(a) $M = 0.80$.

Figure 8.- Variation of section normal-load parameter over the wing semi-span for constant values of wing normal-force coefficient at several Mach numbers.



(b) $M = 0.90.$

Figure 8.- Continued.

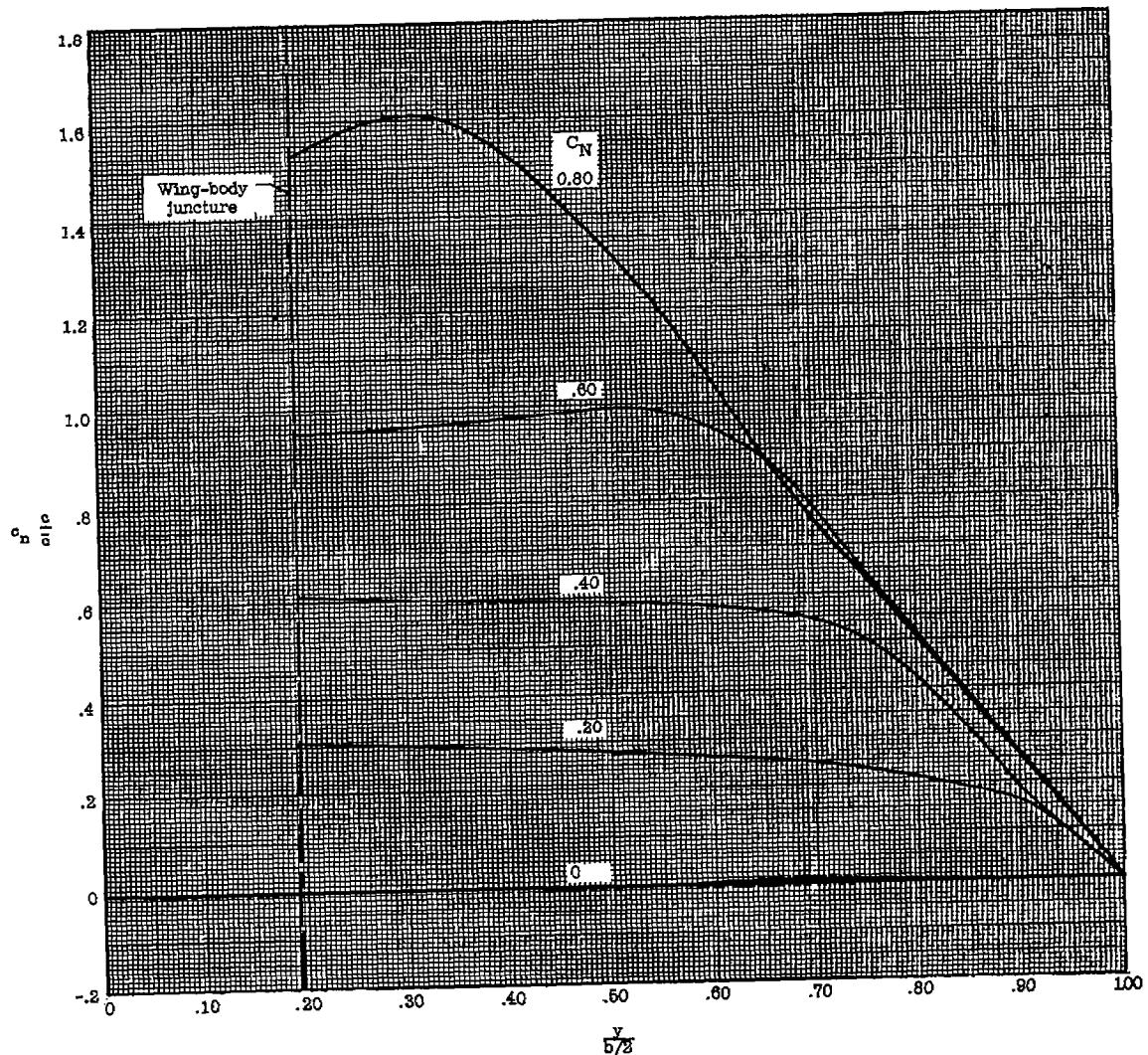
(c) $M = 0.94.$

Figure 8.- Continued.

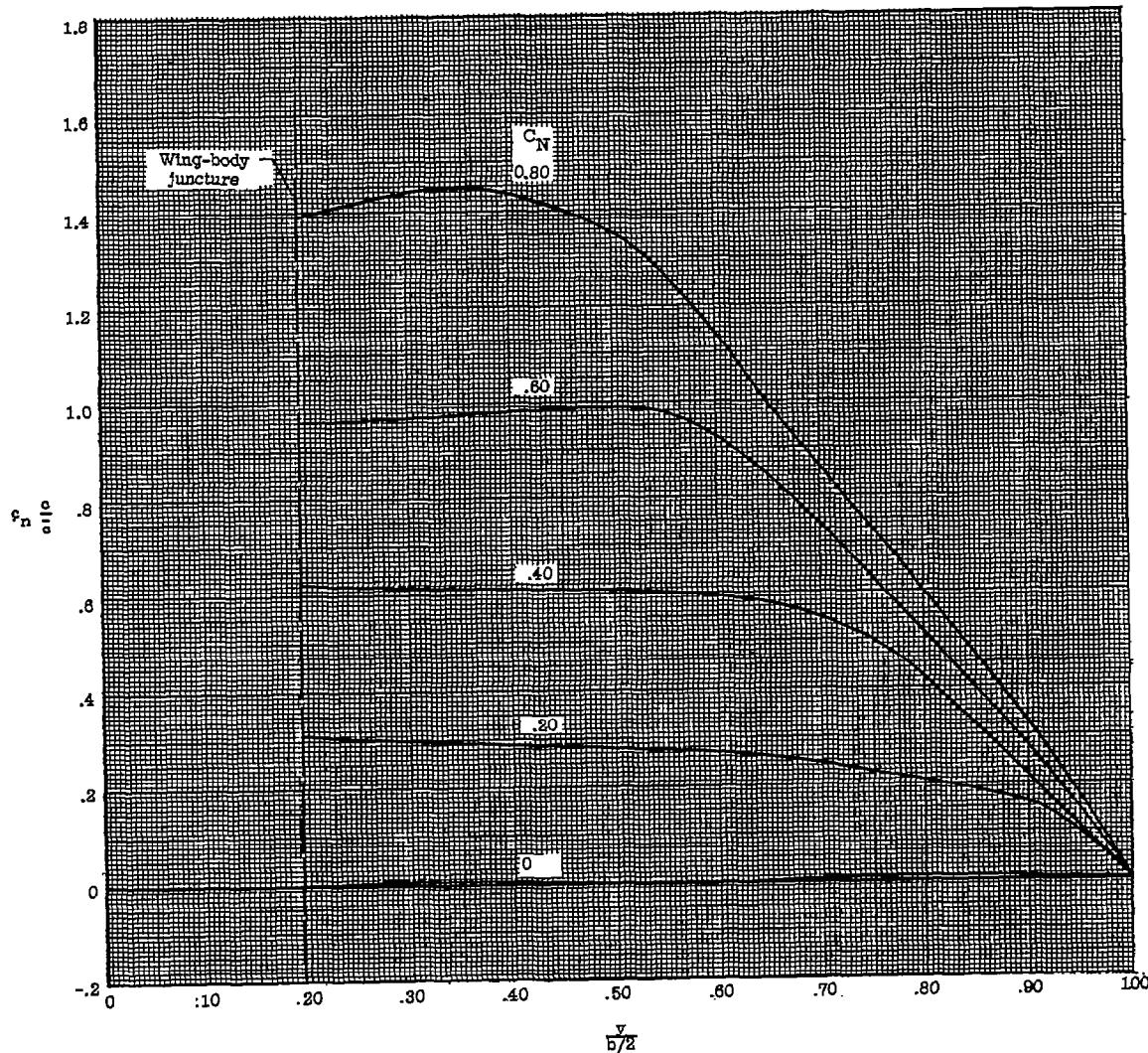
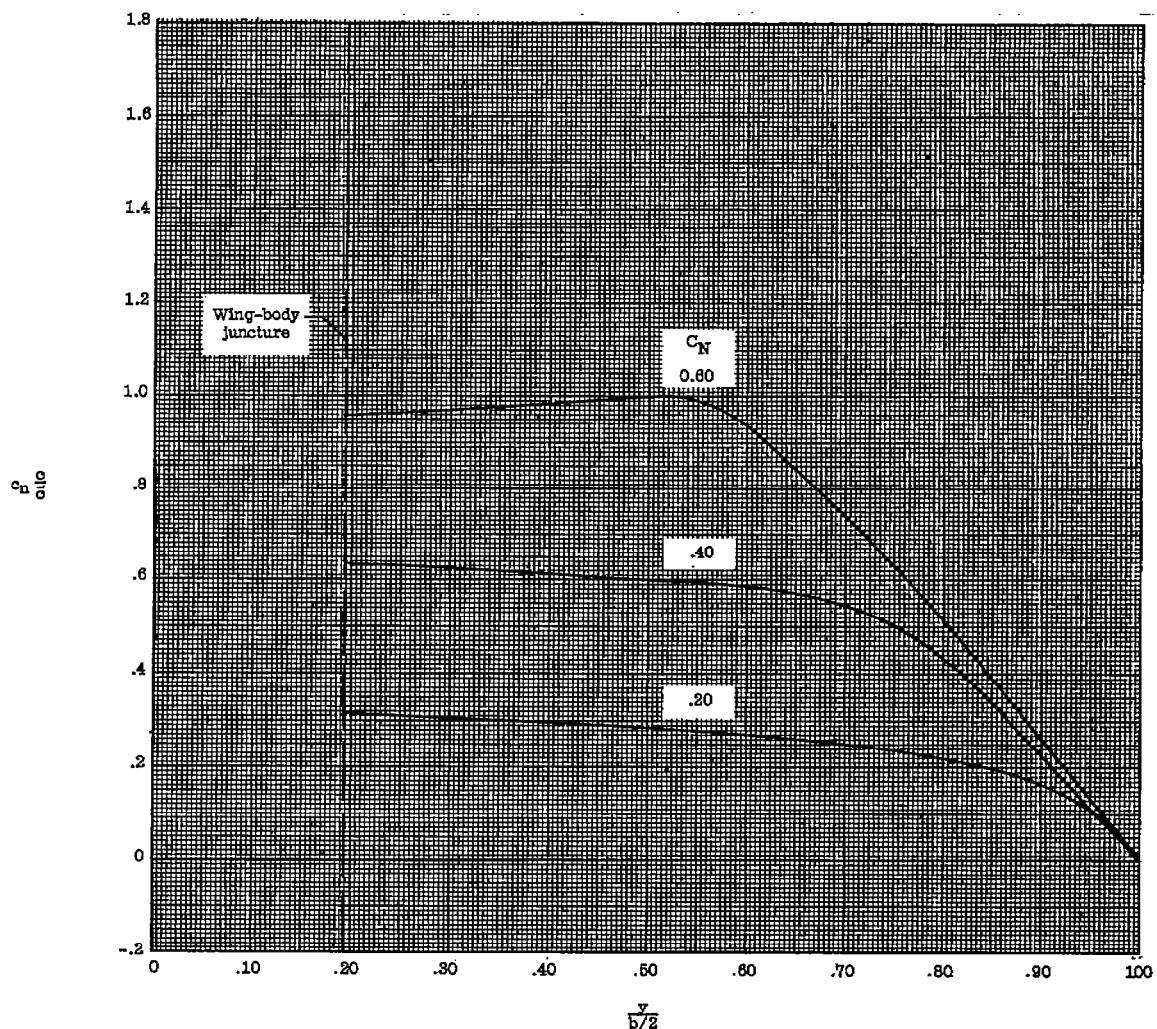
(d) $M = 1.00.$

Figure 8.- Continued.

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(e) $M = 1.03$.

Figure 8.- Continued.

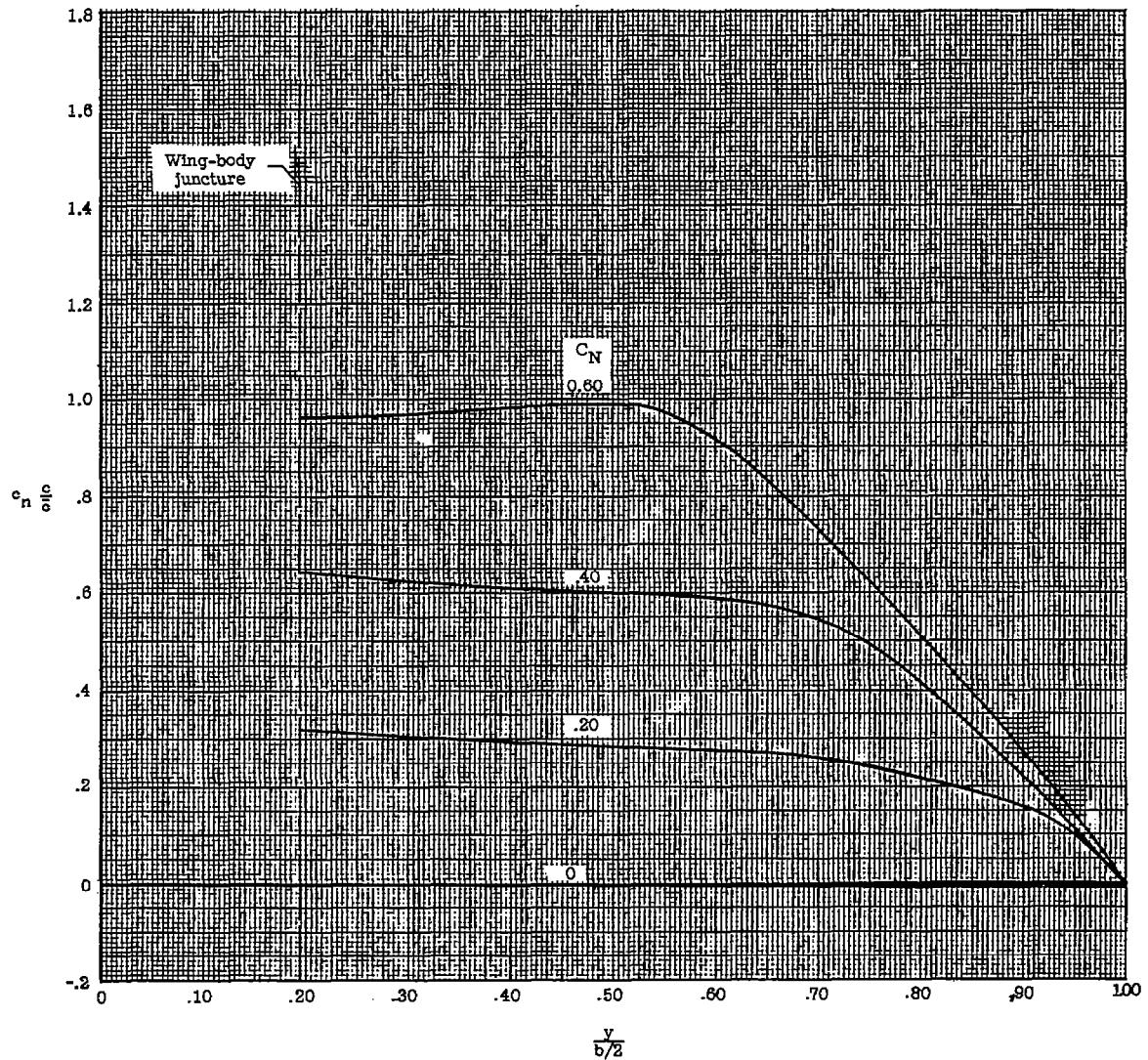
(f) $M = 1.05.$

Figure 8.- Concluded.

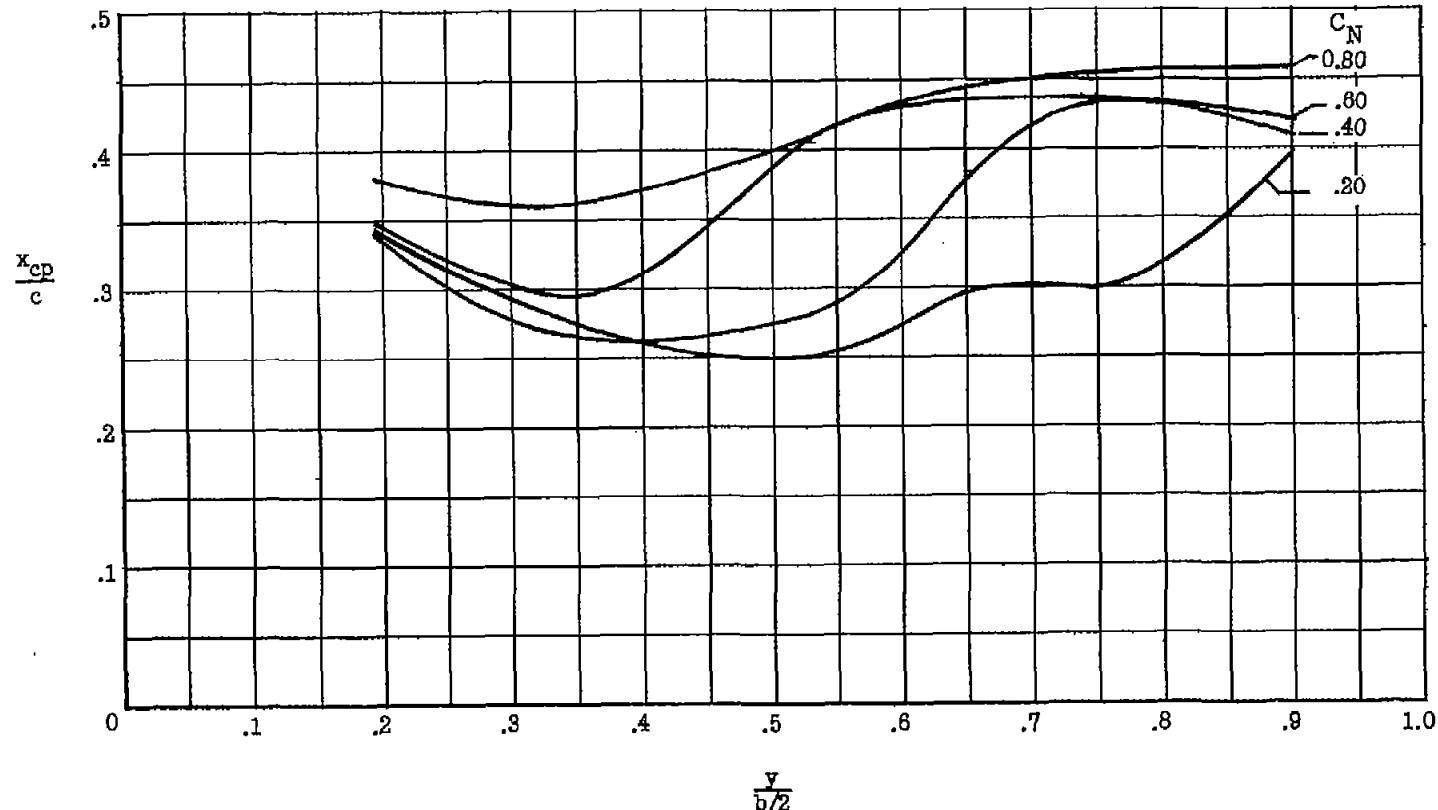
(a) $M = 0.80$.

Figure 9.- Spanwise variation of section center-of-pressure locations for constant values of wing normal-force coefficient at several Mach numbers.

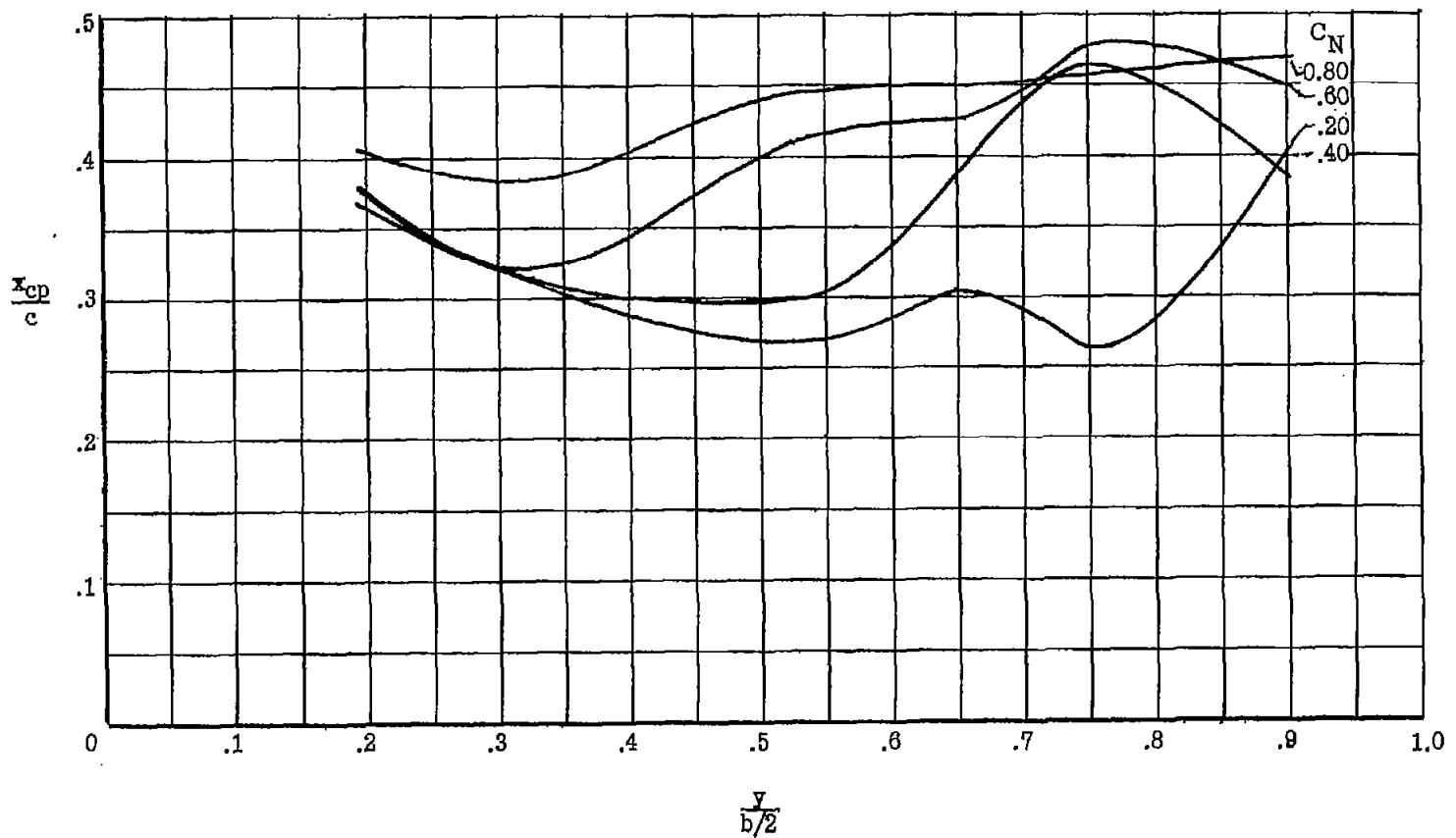
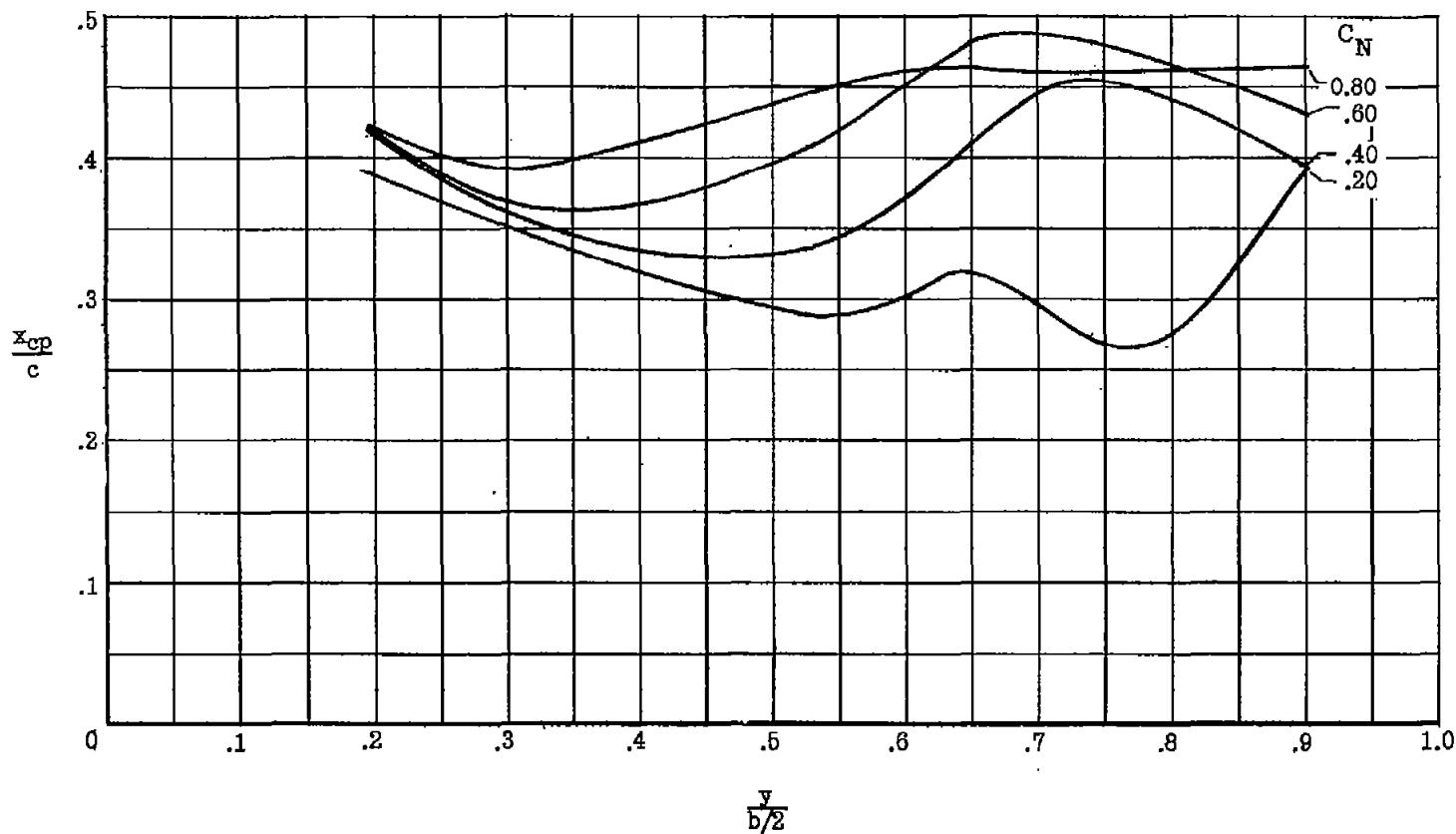
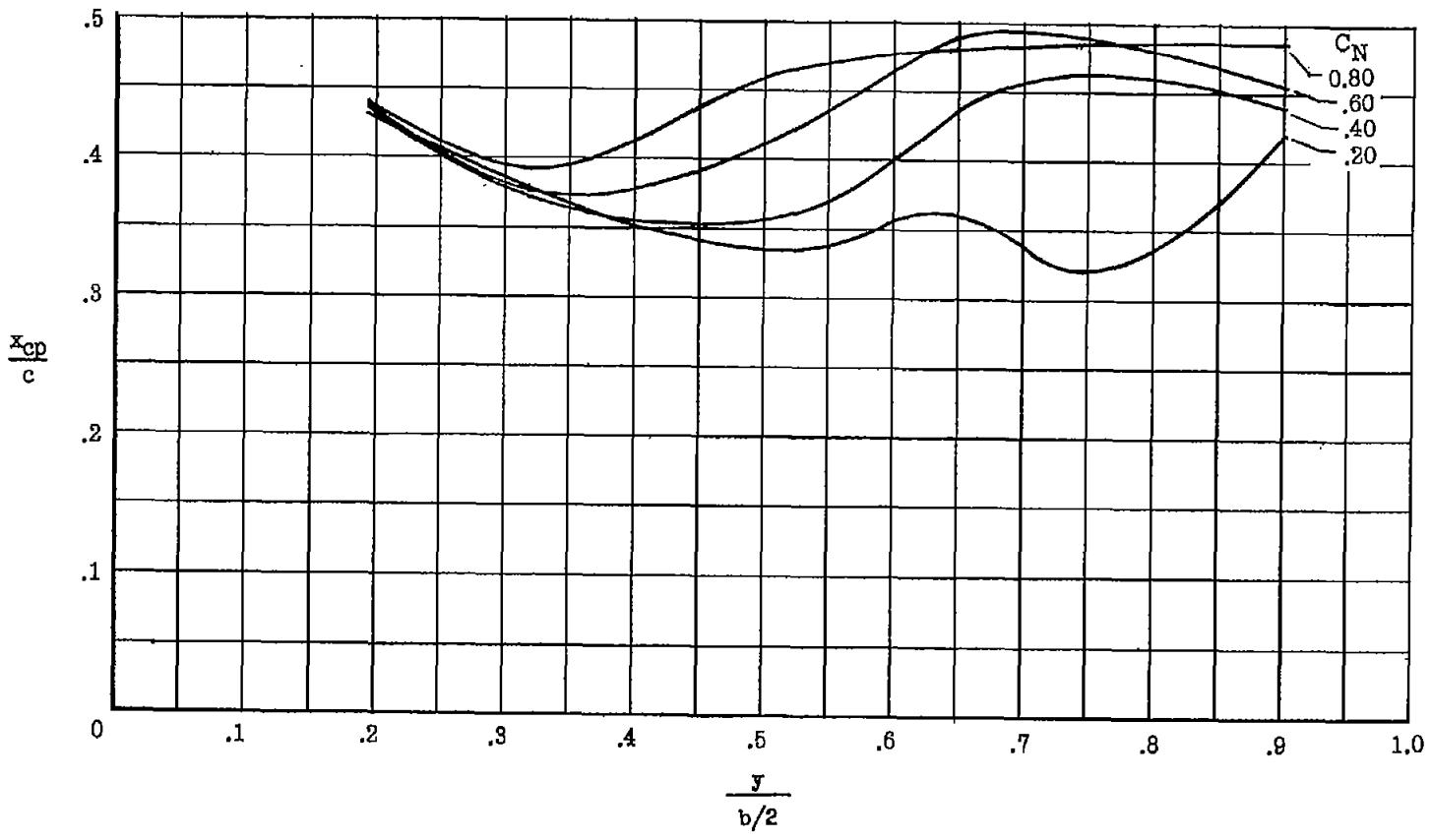
(b) $M = 0.90.$

Figure 9.- Continued.



(c) $M = 0.94.$

Figure 9.- Continued.

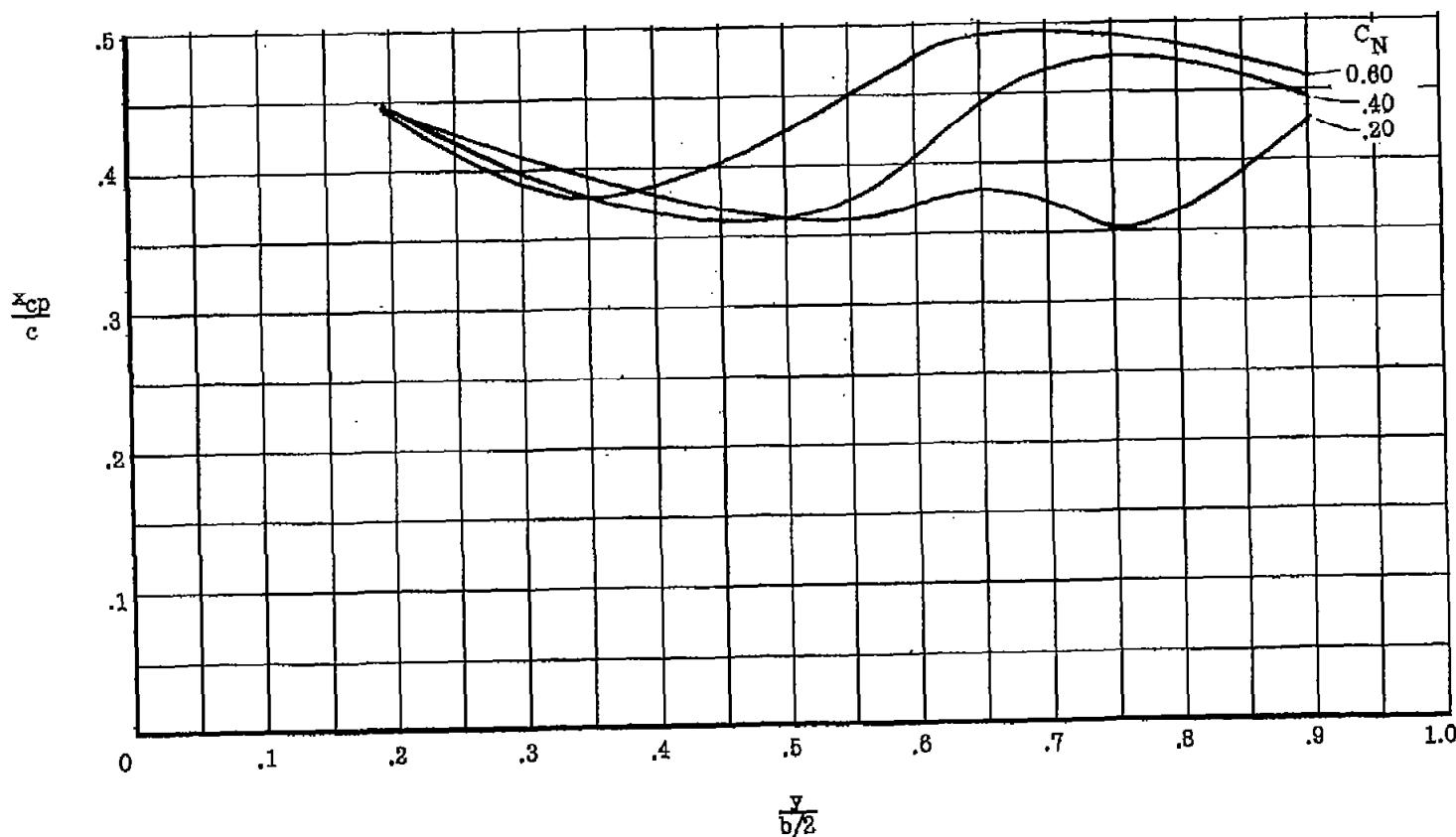


(d) M = 1.00.

Figure 9.-- Continued.

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(e) $M = 1.05$.

Figure 9.- Concluded.

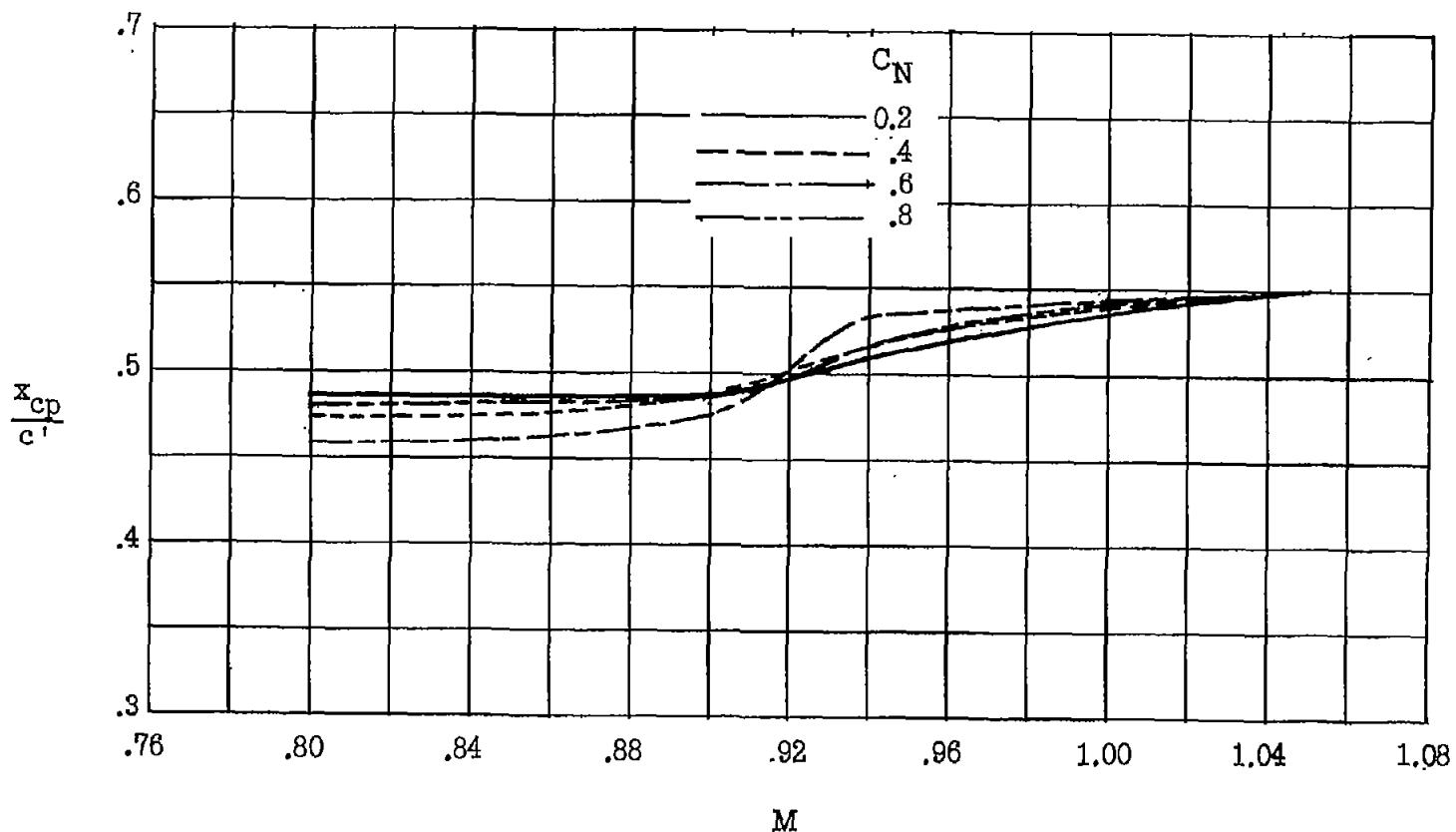


Figure 10.- Effect of Mach number on wing center-of-pressure location.

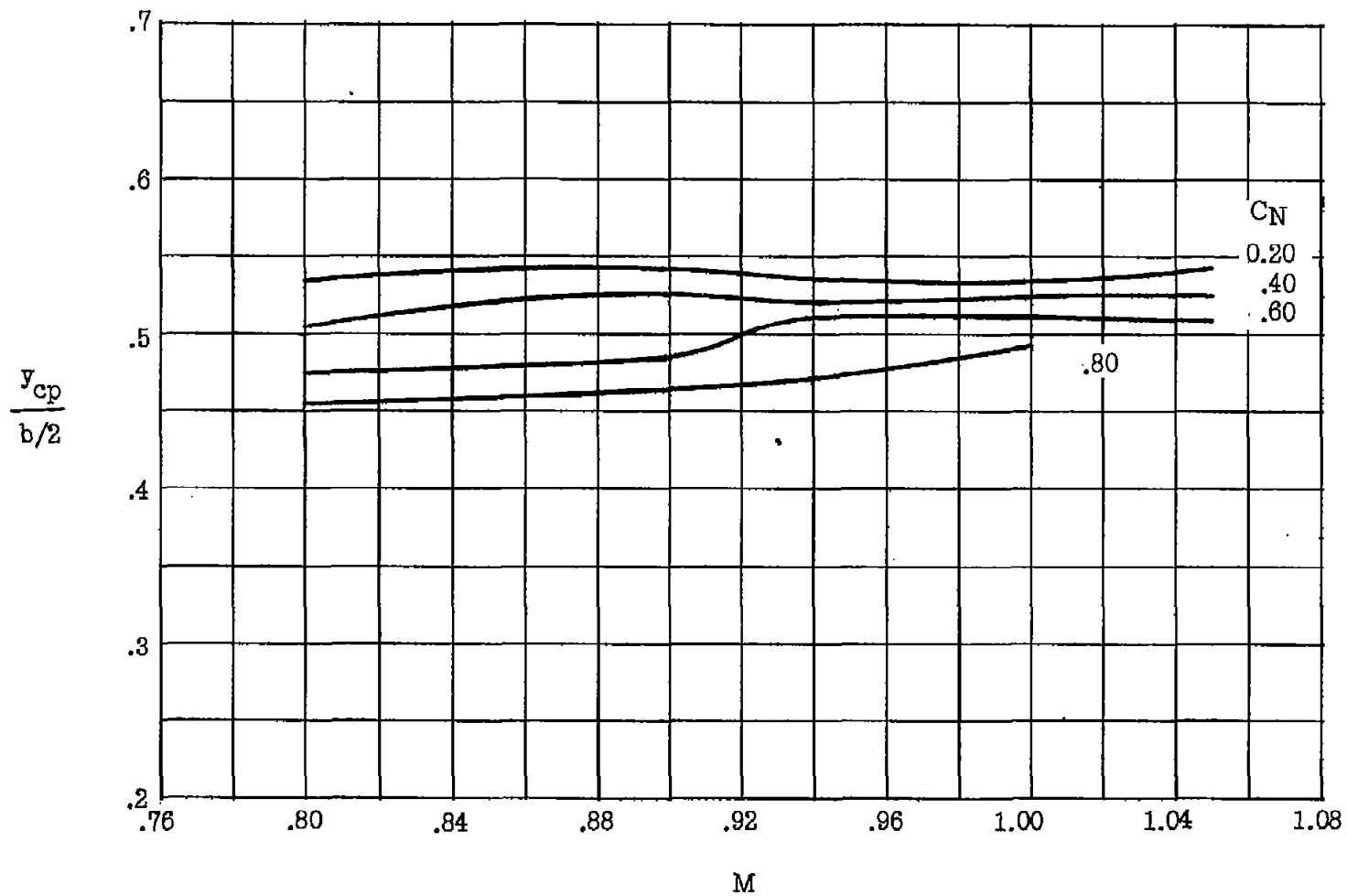
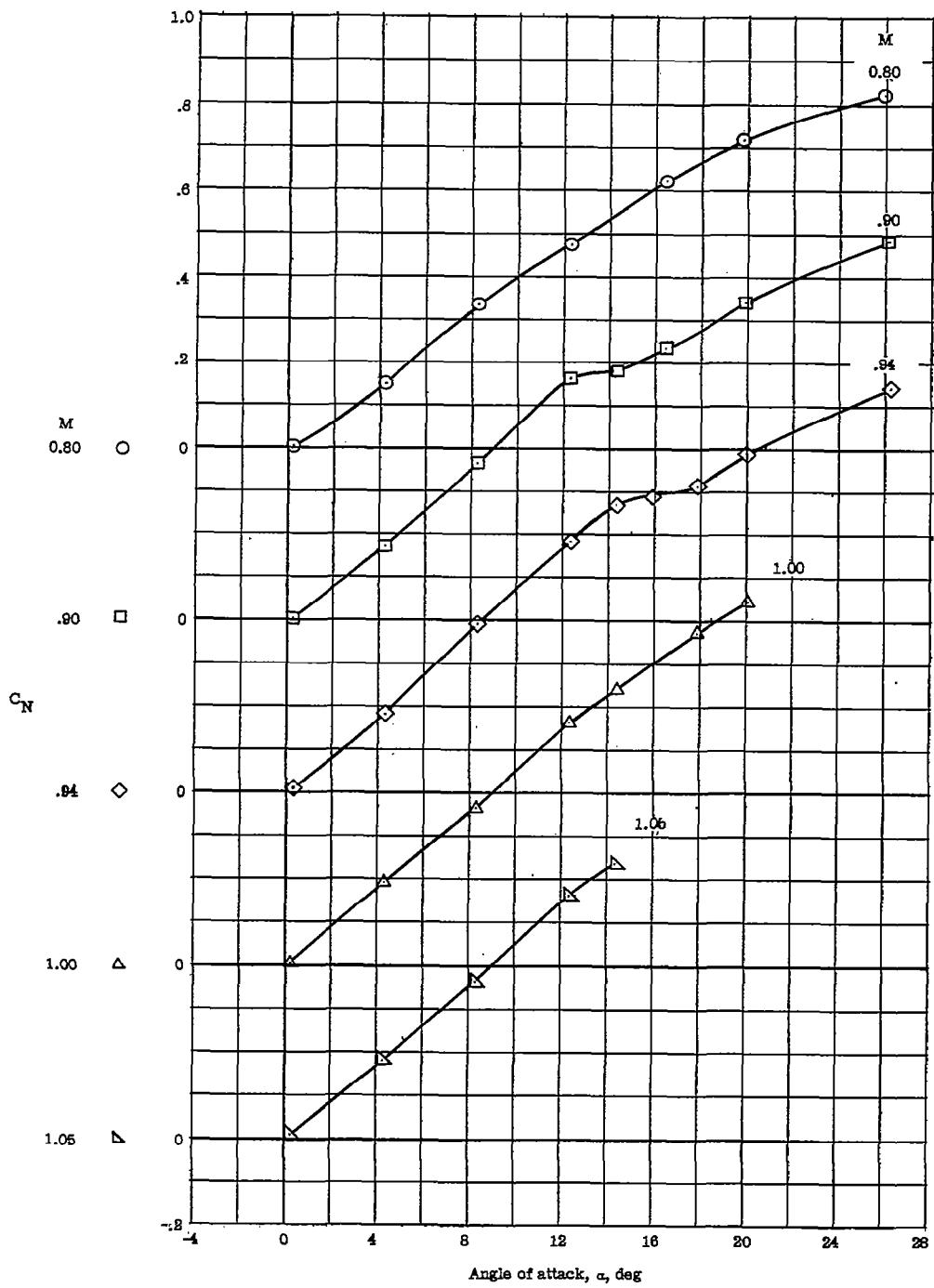


Figure 11.- Variation of spanwise center-of-pressure location with Mach number for constant values of wing normal-force coefficient.



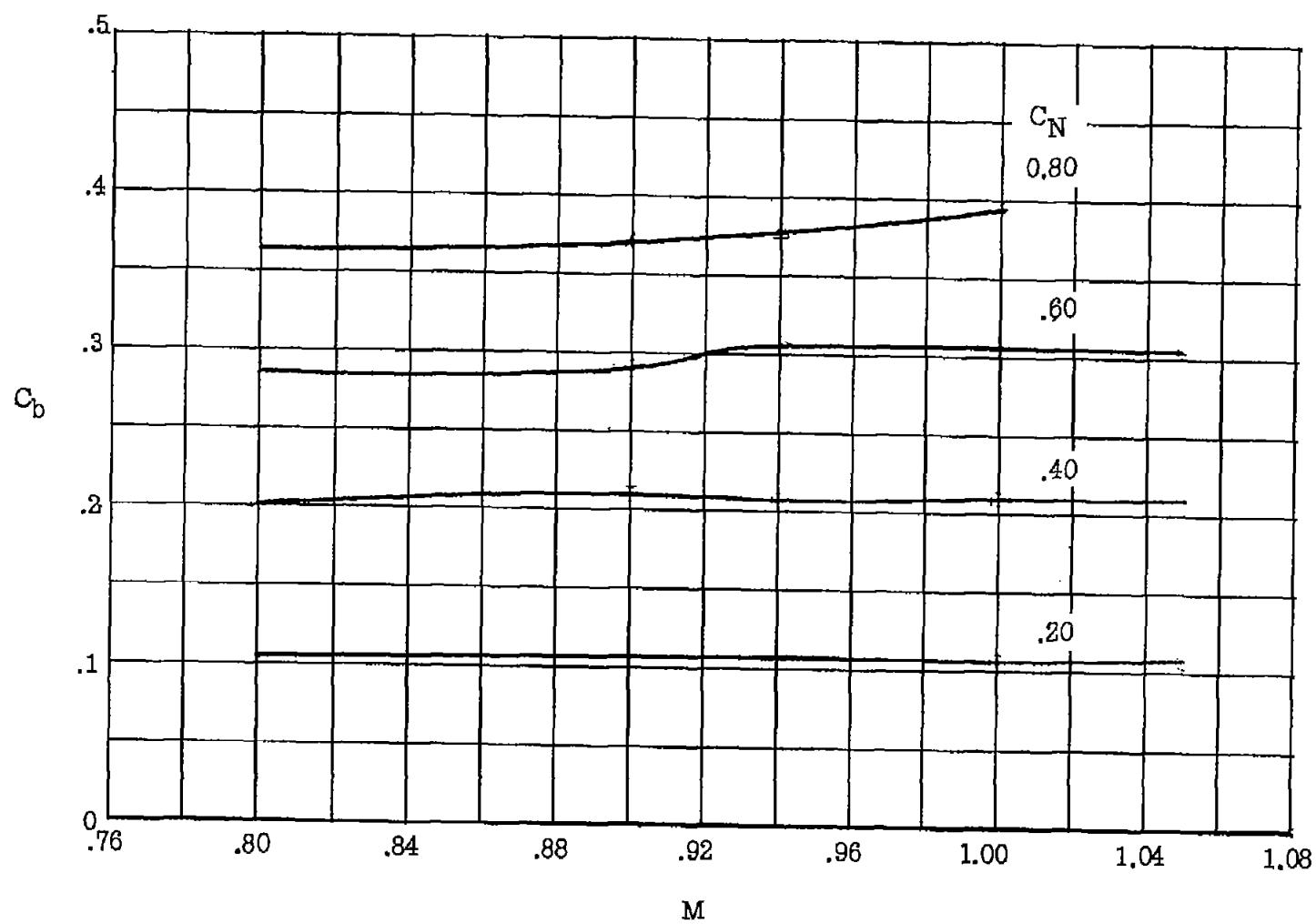


Figure 13.- Effect of Mach number on wing bending-moment coefficient at constant values of wing normal-force coefficient.

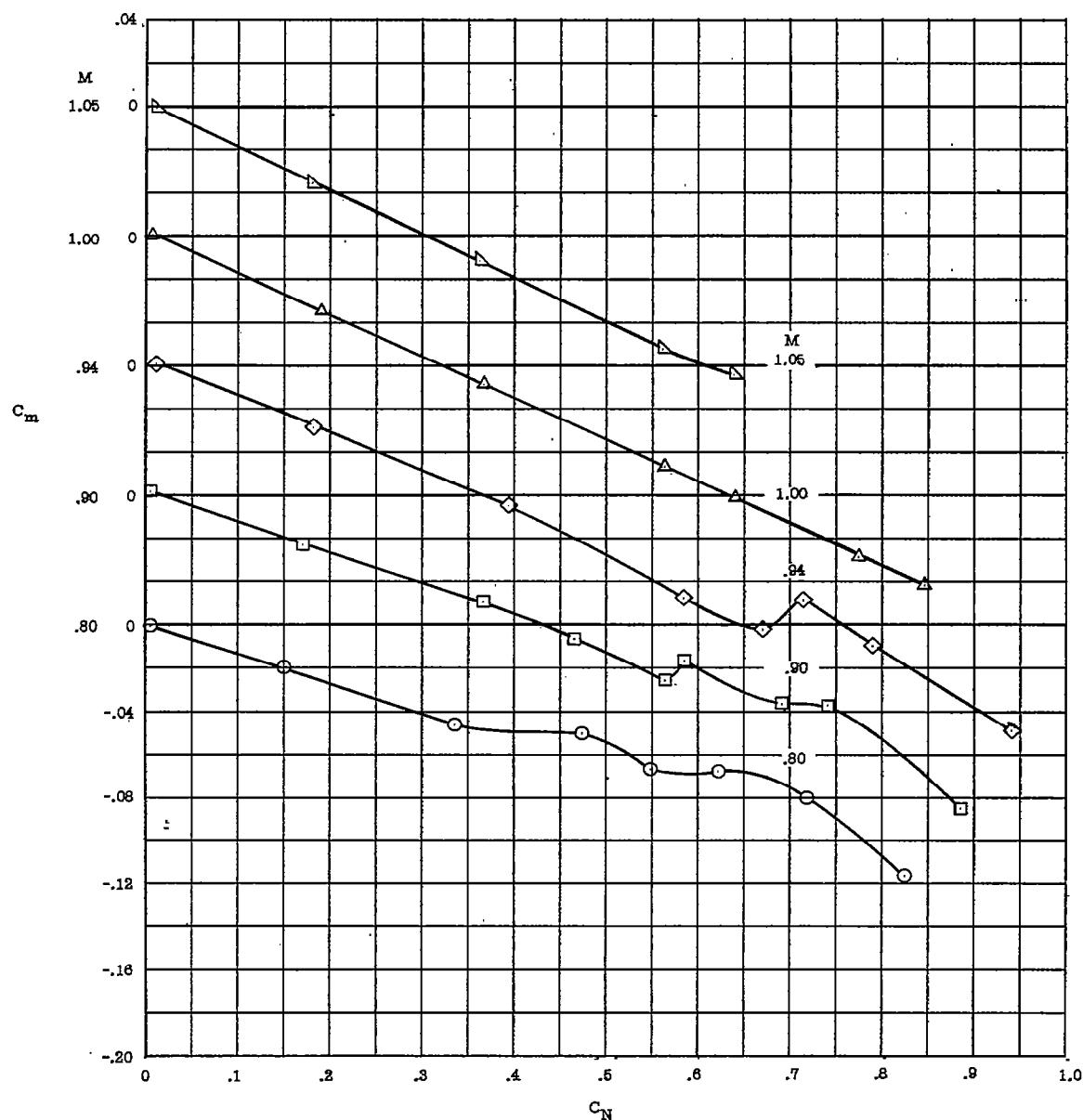


Figure 14.- Variation of pitching-moment coefficient with normal-force coefficient for several Mach numbers.